



HAL
open science

A new non-linear higher-order shear deformation theory for large-amplitude vibrations of laminated doubly curved shells

Marco Amabili, Junuthula Reddy

► **To cite this version:**

Marco Amabili, Junuthula Reddy. A new non-linear higher-order shear deformation theory for large-amplitude vibrations of laminated doubly curved shells. *International Journal of Non-Linear Mechanics*, Elsevier, 2010, 10.1016/j.ijnonlinmec.2009.12.013 . hal-01302309

HAL Id: hal-01302309

<https://hal.archives-ouvertes.fr/hal-01302309>

Submitted on 14 Apr 2016

HAL is a multi-disciplinary open access archive for the deposit and dissemination of scientific research documents, whether they are published or not. The documents may come from teaching and research institutions in France or abroad, or from public or private research centers.

L'archive ouverte pluridisciplinaire **HAL**, est destinée au dépôt et à la diffusion de documents scientifiques de niveau recherche, publiés ou non, émanant des établissements d'enseignement et de recherche français ou étrangers, des laboratoires publics ou privés.



Distributed under a Creative Commons Attribution - ShareAlike | 4.0 International License

A new non-linear higher-order shear deformation theory for large-amplitude vibrations of laminated doubly curved shells

M. Amabili, Department of Mechanical Engineering, McGill University, Montreal, Canada

J.N. Reddy, Department of Mechanical Engineering, Texas A&M University, College Station, USA

A consistent higher-order shear deformation non-linear theory is developed for shells of generic shape, taking geometric imperfections into account. The geometrically non-linear strain–displacement relationships are derived retaining full non-linear terms in the in-plane displacements; they are presented in curvilinear coordinates in a formulation ready to be implemented. Then, large-amplitude forced vibrations of a simply supported, laminated circular cylindrical shell are studied (i) by using the developed theory, and (ii) keeping only non-linear terms of the von Karman type. Results show that inaccurate results are obtained by keeping only non-linear terms of the von Kármán type for vibration amplitudes of about two times the shell thickness for the studied case.

Shells, Nonlinear theory, Shear deformation, Vibrations, Nonlinear vibrations, Rotary inertia,

1. Introduction

Classical shell theories, which neglect shear deformation and rotary inertia, give inaccurate natural frequencies for moderately thick or laminated anisotropic shells and plates (e.g. [1]). In order to overcome this limitation, shear deformation theories have been introduced. These theories can be classified as first- and higher-order shear deformation theories [2]; in the first category, a shear correction factor is required for the equilibrium since a uniform shear strain is assumed through the shell thickness. Higher-order shear deformation theories overcome this limitation since a realistic shear stress distribution through the shell thickness is assumed, which also satisfies the condition of zero shear stresses at both top and bottom shell surfaces.

Several higher-order shear deformation shell theories have been proposed. Librescu [3] developed a non-linear shell theory by expanding the shell displacements with cubic terms in the transverse coordinate. A linear higher-order shear deformation theory of shells has been introduced by Reddy [4] and Reddy and Liu [5]. Arciniega and Reddy [6] have improved the theory developed in [5]. A review of shell theories has been presented by Reddy and Arciniega [7].

Reddy [8] developed the non-linear higher-order shear deformation theory of plates, taking into account von Kármán type non-linear terms. Dennis and Palazotto [9] and Palazotto and Dennis [10] have extended Reddy shell theory [4] to non-linear deformation by introducing the von Kármán type non-linear terms. These theories have been also discussed in the books of Amabili [2] and Reddy [11].

In the existing higher-order shear deformation geometrically non-linear shell theories, the von Kármán type non-linear terms (i.e. those involving the normal displacement only) have been added to the linear equations, so that consistent derivation has not been performed. Moreover, the von Kármán type non-linear terms are known for being accurate for classical shell theories only for small displacements and moderately small rotations, as shown by Amabili [12] in case of vibrations. Therefore, it is important to derive a consistent higher-order shear deformation theory that keeps all the non-linear terms in the normal and in-plane displacements.

In the present study, a consistent higher-order shear deformation non-linear theory is developed for shells of generic shape; geometric imperfections are also taken into account. The geometrically non-linear strain–displacement relationships are derived retaining full non-linear terms in the in-plane displacements. They are presented in curvilinear coordinates in a formulation that can be readily implemented. Then, large-amplitude forced vibrations of a simply supported, laminated circular cylindrical shells are studied (i) by using the developed

theory, and (ii) keeping only non-linear terms of the von Kármán type. Results show that inaccurate results are obtained by keeping only non-linear terms of the von Kármán type for vibration amplitudes of about two times the shell thickness for the studied case.

2. Non-linear higher-order shear deformation theory

A laminated shell of arbitrary shape, made of a finite number of orthotropic layers, oriented arbitrarily with respect to the shell principal curvilinear coordinates (α_1, α_2) , is considered, as shown in Fig. 1. The development of the theory remains the same for shells made of isotropic, orthotropic or functionally graded materials. The displacements of an arbitrary point (α_1, α_2) on the middle surface of the shell are denoted by u, v and w , in the α_1, α_2 and z directions, respectively; w is taken positive outward from the center of the smallest radius of curvature. Initial geometric imperfections of the shell associated with zero initial tension are denoted by displacement w_0 in normal direction, also taken positive outward. The thickness h of the shell is assumed to be small compared to the principal radii of curvature of the shell, so that only moderately thick shells can be considered. The displacements (u_1, u_2, u_3) of a generic point are related to the middle surface displacements by

$$u_1 = (1+z/R_1)u + z\phi_1 + z^2\psi_1 + z^3\gamma_1 + z^4\theta_1, \quad (1a)$$

$$u_2 = (1+z/R_2)v + z\phi_2 + z^2\psi_2 + z^3\gamma_2 + z^4\theta_2, \quad (1b)$$

$$u_3 = w + z\chi + w_0, \quad (1c)$$

where ϕ_1 and ϕ_2 are the rotations of the transverse normals at $z=0$ about the α_2 and α_1 axes, respectively, and $\psi_1, \psi_2, \gamma_1, \gamma_2, \theta_1, \theta_2$ and χ are functions to be determined in terms of u, v, w, ϕ_1 and ϕ_2 . Thus, the five variables describing the shell deformation are u, v, w, ϕ_1 and ϕ_2 . In Eqs. (1) the in-plane displacements have been expanded up to the 4th order in z while the normal displacement has been assumed to be linear in z ; Eqs. (1a,b) give a high-order distribution of shear effects through the thickness.

The shear and normal Green's strains for three-dimensional elasticity are [2]

$$\begin{aligned} \gamma_{13} = & \frac{\partial u_1}{\partial z} + \frac{1}{1+z/R_1} \left[\left(\frac{1}{A_1} \frac{\partial u_3}{\partial \alpha_1} - \frac{u_1}{R_1} \right) + \left(\frac{1}{A_1} \frac{\partial u_1}{\partial \alpha_1} + \frac{1}{A_1 A_2} \frac{\partial A_1}{\partial \alpha_2} u_2 + \frac{u_3}{R_1} \right) \frac{\partial u_1}{\partial z} \right. \\ & \left. + \left(\frac{1}{A_1} \frac{\partial u_2}{\partial \alpha_1} - \frac{1}{A_1 A_2} \frac{\partial A_1}{\partial \alpha_2} u_1 \right) \frac{\partial u_2}{\partial z} + \left(\frac{1}{A_1} \frac{\partial u_3}{\partial \alpha_1} - \frac{u_1}{R_1} \right) \frac{\partial u_3}{\partial z} \right], \end{aligned} \quad (2a)$$

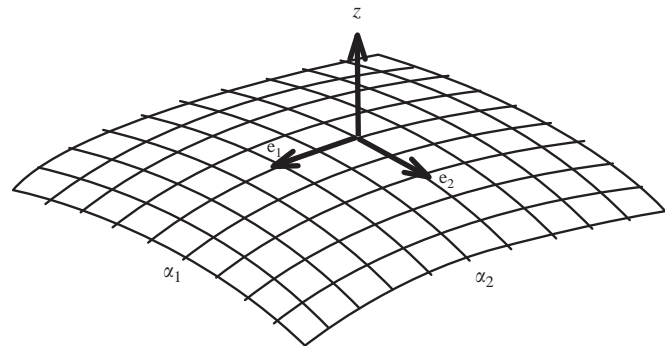


Fig. 1. Doubly curved shell and curvilinear coordinate system.

$$\begin{aligned} \gamma_{23} = & \frac{\partial u_2}{\partial z} + \frac{1}{1+z/R_2} \left[\left(\frac{1}{A_2} \frac{\partial u_3}{\partial \alpha_2} - \frac{u_2}{R_2} \right) + \left(\frac{1}{A_2} \frac{\partial u_2}{\partial \alpha_2} + \frac{1}{A_1 A_2} \frac{\partial A_2}{\partial \alpha_1} u_1 + \frac{u_3}{R_2} \right) \right. \\ & \left. \times \frac{\partial u_2}{\partial z} + \left(\frac{1}{A_2} \frac{\partial u_1}{\partial \alpha_2} - \frac{1}{A_1 A_2} \frac{\partial A_2}{\partial \alpha_1} u_2 \right) \frac{\partial u_1}{\partial z} + \left(\frac{1}{A_2} \frac{\partial u_3}{\partial \alpha_2} - \frac{u_2}{R_2} \right) \frac{\partial u_3}{\partial z} \right], \end{aligned} \quad (2b)$$

$$\varepsilon_{33} = \frac{\partial u_3}{\partial z} + \frac{1}{2} \left[\left(\frac{\partial u_1}{\partial z} \right)^2 + \left(\frac{\partial u_2}{\partial z} \right)^2 + \left(\frac{\partial u_3}{\partial z} \right)^2 \right]. \quad (2c)$$

Eqs. (2a–c) are non-linear; in Eqs. (2a,b), R_1 and R_2 (functions of the coordinates α_1 and α_2) are the principal radii of curvature in α_1 and α_2 directions, respectively, and A_1 and A_2 are the Lamé parameters. The shear deformation, see Eqs. (2a,b), is neglected in classical shell theories, which is a very good approximation for thin and moderately thick isotropic shells and for very thin laminated shells [2,12]. For laminated shells that cannot be considered very thin, shear deformation should be retained in order to obtain accurate results. However, shear deformation plays a smaller role than in-plane strains and bending (since it is even neglected in classical theories), so that it is reasonable to keep only linear displacement terms in Eqs. (2a,b). This assumption leads to the possibility of expressing the displacement field in Eqs. (1a–c) by using only linear expressions in u, v, w, ϕ_1 and ϕ_2 . Therefore, the following relationships are obtained for the transverse shear strains:

$$\gamma_{13} = \frac{\partial u_1}{\partial z} + \frac{1}{1+z/R_1} \left(\frac{\partial u_3}{A_1 \partial \alpha_1} - \frac{u_1}{R_1} \right), \quad (3a)$$

$$\gamma_{23} = \frac{\partial u_2}{\partial z} + \frac{1}{1+z/R_2} \left(\frac{\partial u_3}{A_2 \partial \alpha_2} - \frac{u_2}{R_2} \right). \quad (3b)$$

It is also assumed that normals to the middle surface of the shell before deformation are not elongated after deformation. This is mathematically expressed by

$$\varepsilon_{33}|_{z=0} = \frac{\partial u_3}{\partial z} + \frac{1}{2} \left[\left(\frac{\partial u_1}{\partial z} \right)^2 + \left(\frac{\partial u_2}{\partial z} \right)^2 + \left(\frac{\partial u_3}{\partial z} \right)^2 \right] = 0. \quad (4)$$

The constraint given by Eq. (4) could be removed and the shell theory will become a six-variable theory, instead of having five variables, χ being the additional variable.

The expressions for the transverse shear strains are obtained by substituting Eqs. (1a–c) into (3a,b); the vanishing of the shear strains at the top and the bottom surfaces of the shell requires

$$\gamma_{13}|_{z=\pm h/2} = 0, \quad \gamma_{23}|_{z=\pm h/2} = 0. \quad (5a,b)$$

The following approximation for moderately thick shell is introduced

$$\frac{h}{2R \pm h} \simeq \frac{h}{2R} \left(1 \mp \frac{h}{2R} \right). \quad (6)$$

Collecting first- and third-order terms in h from Eq. (5a) into two separate equations, the following expressions are obtained since these expressions change sign if evaluated at $z = \pm h/2$ and must be set equal to zero

$$\frac{1}{4R_1^2} \psi_1 - \frac{1}{4R_1} \gamma_1 + \theta_1 = 0, \quad (7a)$$

$$-\frac{1}{2R_1} \phi_1 + \psi_1 + \frac{1}{2A_1} \frac{\partial \chi}{\partial \alpha_1} - \frac{1}{2R_1 A_1} \frac{\partial w}{\partial \alpha_1} = 0. \quad (7b)$$

Eqs. (7a,b) give

$$\psi_1 = \frac{1}{2R_1} \phi_1 + \frac{1}{2R_1} \frac{\partial w}{\partial \alpha_1} - \frac{1}{2A_1} \frac{\partial \chi}{\partial \alpha_1}, \quad (8a)$$

$$\theta_1 = \frac{1}{4R_1}\gamma_1 - \frac{1}{4R_1^2}\psi_1. \quad (8b)$$

Then, Eq. (5a) gives

$$\gamma_1 = \frac{-4}{3h^2} \left[\phi_1 + \frac{\partial w}{A_1 \partial \alpha_1} + \frac{h^2}{8R_1^2} \left(\phi_1 - \frac{\partial w}{A_1 \partial \alpha_1} - R_1 \frac{\partial \chi}{A_1 \partial \alpha_1} \right) \right]. \quad (8c)$$

Similarly, from Eq. (5b), the following expressions are obtained:

$$\psi_2 = \frac{1}{2R_2}\phi_2 + \frac{1}{2R_2}\frac{\partial w}{\partial \alpha_2} - \frac{1}{2A_2}\frac{\partial \chi}{\partial \alpha_2}, \quad (9a)$$

$$\theta_2 = \frac{1}{4R_2}\gamma_2 - \frac{1}{4R_2^2}\psi_2, \quad (9b)$$

$$\gamma_2 = \frac{-4}{3h^2} \left[\phi_2 + \frac{\partial w}{A_2 \partial \alpha_2} + \frac{h^2}{8R_2^2} \left(\phi_2 - \frac{\partial w}{A_2 \partial \alpha_2} - R_2 \frac{\partial \chi}{A_2 \partial \alpha_2} \right) \right]. \quad (9c)$$

By using Eq. (4), the function χ can be determined

$$\begin{aligned} \chi &= -1 + \sqrt{1 - \left(\frac{u}{R_1^2} + \phi_1 \right)^2 - \left(\frac{v}{R_2^2} + \phi_2 \right)^2} \\ &\simeq -\frac{1}{2} \left(\phi_1^2 + \phi_2^2 + 2\phi_1 \frac{u}{R_1} + 2\phi_2 \frac{v}{R_2} + \frac{u^2}{R_1^2} + \frac{v^2}{R_2^2} \right), \end{aligned} \quad (10a)$$

where the approximation in Eq. (10a) is accurate for $\phi_1, \phi_2 \ll 1$. Eq. (10a) shows that χ is a second-order term with respect to the other five variables, i.e. displacements and rotations, when the constraint given by Eq. (4) is introduced. Therefore, it is possible to approximate

$$\chi \simeq 0. \quad (10b)$$

Under this approximation, Eqs. (8a–c) and (9a–c) are linear; moreover, Eqs. (1a–c) are also linear in the five variables describing the shell deformation u, v, w, ϕ_1 and ϕ_2 . The linear expression of Eqs. (1a–c) is a consequence of the assumption (10b) and of retaining only linear terms in Eqs. (2a,b).

By substituting Eqs. (1a–c) into (3a,b) and using the approximations (6) and Eqs. (8), (9), (10b), the following strain–displacement relationships are obtained for the shear strains keeping terms up to z^3 :

$$\gamma_{13} = \gamma_{13,0} + z(k_{13}^{(0)} + zk_{13}^{(1)} + z^2k_{13}^{(2)}), \quad (11a)$$

$$\gamma_{23} = \gamma_{23,0} + z(k_{23}^{(0)} + zk_{23}^{(1)} + z^2k_{23}^{(2)}), \quad (11b)$$

where

$$\gamma_{13,0} = \phi_1 + \frac{\partial w}{A_1 \partial \alpha_1}, \quad (12a)$$

$$\gamma_{23,0} = \phi_2 + \frac{\partial w}{A_2 \partial \alpha_2}, \quad (12b)$$

$$k_{13}^{(0)} = 0, \quad k_{13}^{(1)} = -\frac{4}{h^2} \left(\phi_1 + \frac{\partial w}{A_1 \partial \alpha_1} \right) + \frac{u}{R_1^3}, \quad k_{13}^{(2)} = 0, \quad (13a-c)$$

$$k_{23}^{(0)} = 0, \quad k_{23}^{(1)} = -\frac{4}{h^2} \left(\phi_2 + \frac{\partial w}{A_2 \partial \alpha_2} \right) + \frac{v}{R_2^3}, \quad k_{23}^{(2)} = 0. \quad (13d-f)$$

Eqs. (11)–(13) show that the strain and stress distribution through the thickness is parabolic and therefore the shear correction factor is no longer required.

The expressions of the other Green's strain tensor ε_{ij} in the curvilinear coordinate system, are [2]

$$\varepsilon_{11} = \bar{\varepsilon}_1 + \frac{1}{2}(\bar{\varepsilon}_1^2 + \omega_1^2 + \bar{\Theta}^2), \quad (14a)$$

$$\varepsilon_{22} = \bar{\varepsilon}_2 + \frac{1}{2}(\omega_2^2 + \bar{\varepsilon}_2^2 + \bar{\Psi}^2), \quad (14b)$$

$$\gamma_{12} = \omega_1 + \omega_2 + (\bar{\varepsilon}_1 \omega_2 + \bar{\varepsilon}_2 \omega_1 + \bar{\Theta} \bar{\Psi}), \quad (14c)$$

where

$$\bar{\varepsilon}_1 = \frac{1}{1+z/R_1} \left(\frac{1}{A_1} \frac{\partial u_1}{\partial \alpha_1} + \frac{1}{A_1 A_2} \frac{\partial A_1}{\partial \alpha_2} u_2 + \frac{u_3}{R_1} \right), \quad (15a)$$

$$\bar{\varepsilon}_2 = \frac{1}{1+z/R_2} \left(\frac{1}{A_2} \frac{\partial u_2}{\partial \alpha_2} + \frac{1}{A_1 A_2} \frac{\partial A_2}{\partial \alpha_1} u_1 + \frac{u_3}{R_2} \right), \quad (15b)$$

$$\omega_1 = \frac{1}{1+z/R_1} \left(\frac{1}{A_1} \frac{\partial u_2}{\partial \alpha_1} - \frac{1}{A_1 A_2} \frac{\partial A_1}{\partial \alpha_2} u_1 \right), \quad (15c)$$

$$\omega_2 = \frac{1}{1+z/R_2} \left(\frac{1}{A_2} \frac{\partial u_1}{\partial \alpha_2} - \frac{1}{A_1 A_2} \frac{\partial A_2}{\partial \alpha_1} u_2 \right), \quad (15d)$$

$$\bar{\Theta} = \frac{1}{1+z/R_1} \left(-\frac{1}{A_1} \frac{\partial u_3}{\partial \alpha_1} + \frac{u_1}{R_1} \right), \quad (15e)$$

$$\bar{\Psi} = \frac{1}{1+z/R_2} \left(-\frac{1}{A_2} \frac{\partial u_3}{\partial \alpha_2} + \frac{u_2}{R_2} \right). \quad (15f)$$

The strain–displacement equations for the higher-order shear deformation theory to be added to Eqs. (11a,b), keeping terms up to z^3 and using approximations (6) and (10), are obtained by substituting Eqs. (1), (8), (9), (10b) into Eqs. (14) and (15)

$$\varepsilon_{11} = \varepsilon_{11,0} + z(k_{11}^{(0)} + zk_{11}^{(1)} + z^2k_{11}^{(2)}), \quad (16a)$$

$$\varepsilon_{22} = \varepsilon_{22,0} + z(k_{22}^{(0)} + zk_{22}^{(1)} + z^2k_{22}^{(2)}), \quad (16b)$$

$$\gamma_{12} = \gamma_{12,0} + z(k_{12}^{(0)} + zk_{12}^{(1)} + z^2k_{12}^{(2)}), \quad (16c)$$

where

$$\begin{aligned} \varepsilon_{11,0} &= \frac{1}{A_1} \frac{\partial u}{\partial \alpha_1} + \frac{1}{A_1 A_2} \frac{\partial A_1}{\partial \alpha_2} v + \frac{w}{R_1} + \frac{1}{2} \left[\left(\frac{\partial u}{A_1 \partial \alpha_1} + \frac{1}{A_1 A_2} \frac{\partial A_1}{\partial \alpha_2} v + \frac{w}{R_1} \right)^2 \right. \\ &\quad \left. + \left(\frac{\partial v}{A_1 \partial \alpha_1} - \frac{1}{A_1 A_2} \frac{\partial A_1}{\partial \alpha_2} u \right)^2 + \left(\frac{\partial w}{A_1 \partial \alpha_1} - \frac{u}{R_1} \right)^2 \right] \\ &\quad + \frac{w_0}{R_1} \left(\frac{\partial u}{A_1 \partial \alpha_1} + \frac{1}{A_1 A_2} \frac{\partial A_1}{\partial \alpha_2} v + \frac{w}{R_1} \right) + \frac{\partial w_0}{A_1 \partial \alpha_1} \left(\frac{\partial w}{A_1 \partial \alpha_1} - \frac{u}{R_1} \right), \end{aligned} \quad (17a)$$

$$\begin{aligned} \varepsilon_{22,0} &= \frac{1}{A_2} \frac{\partial v}{\partial \alpha_2} + \frac{1}{A_1 A_2} \frac{\partial A_2}{\partial \alpha_1} u + \frac{w}{R_2} + \frac{1}{2} \left[\left(\frac{1}{A_2} \frac{\partial u}{\partial \alpha_2} - \frac{1}{A_1 A_2} \frac{\partial A_2}{\partial \alpha_1} v \right)^2 \right. \\ &\quad \left. + \left(\frac{1}{A_2} \frac{\partial v}{\partial \alpha_2} + \frac{1}{A_1 A_2} \frac{\partial A_2}{\partial \alpha_1} u + \frac{w}{R_2} \right)^2 + \left(\frac{1}{A_2} \frac{\partial w}{\partial \alpha_2} - \frac{v}{R_2} \right)^2 \right] \\ &\quad + \frac{w_0}{R_2} \left(\frac{1}{A_2} \frac{\partial v}{\partial \alpha_2} + \frac{1}{A_1 A_2} \frac{\partial A_2}{\partial \alpha_1} u + \frac{w}{R_2} \right) + \frac{\partial w_0}{A_2 \partial \alpha_2} \left(\frac{1}{A_2} \frac{\partial w}{\partial \alpha_2} - \frac{v}{R_2} \right), \end{aligned} \quad (17b)$$

$$\begin{aligned} \gamma_{12,0} &= \frac{1}{A_1} \frac{\partial v}{\partial \alpha_1} + \frac{1}{A_2} \frac{\partial u}{\partial \alpha_2} - \frac{1}{A_1 A_2} \frac{\partial A_1}{\partial \alpha_2} u - \frac{1}{A_1 A_2} \frac{\partial A_2}{\partial \alpha_1} v \\ &\quad + \left(\frac{1}{A_1} \frac{\partial u}{\partial \alpha_1} + \frac{1}{A_1 A_2} \frac{\partial A_1}{\partial \alpha_2} v + \frac{w}{R_1} \right) \left(\frac{1}{A_2} \frac{\partial u}{\partial \alpha_2} - \frac{1}{A_1 A_2} \frac{\partial A_2}{\partial \alpha_1} v \right) \\ &\quad + \left(\frac{1}{A_2} \frac{\partial v}{\partial \alpha_2} + \frac{1}{A_1 A_2} \frac{\partial A_2}{\partial \alpha_1} u + \frac{w}{R_2} \right) \left(\frac{1}{A_1} \frac{\partial v}{\partial \alpha_1} - \frac{1}{A_1 A_2} \frac{\partial A_1}{\partial \alpha_2} u \right) \\ &\quad + \left(\frac{1}{A_1} \frac{\partial w}{\partial \alpha_1} - \frac{u}{R_1} \right) \left(\frac{1}{A_2} \frac{\partial w}{\partial \alpha_2} - \frac{v}{R_2} \right) + \frac{w_0}{R_1} \left(\frac{1}{A_2} \frac{\partial u}{\partial \alpha_2} - \frac{1}{A_1 A_2} \frac{\partial A_2}{\partial \alpha_1} v \right) \\ &\quad + \frac{w_0}{R_2} \left(\frac{1}{A_1} \frac{\partial v}{\partial \alpha_1} - \frac{1}{A_1 A_2} \frac{\partial A_1}{\partial \alpha_2} u \right) + \frac{1}{A_1} \frac{\partial w_0}{\partial \alpha_1} \left(\frac{1}{A_2} \frac{\partial w}{\partial \alpha_2} - \frac{v}{R_2} \right) \\ &\quad + \frac{1}{A_2} \frac{\partial w_0}{\partial \alpha_2} \left(\frac{1}{A_1} \frac{\partial w}{\partial \alpha_1} - \frac{u}{R_1} \right), \end{aligned} \quad (17c)$$

$$k_1^{(0)} = \frac{\partial \phi_1}{A_1 \partial \alpha_1} - \frac{w}{R_1^2} + \frac{v}{A_1 A_2} \frac{\partial A_1}{\partial \alpha_2} \left(-\frac{1}{R_1} + \frac{1}{R_2} \right) + \frac{\phi_2}{A_1 A_2} \frac{\partial A_1}{\partial \alpha_2}, \quad (18a)$$

$$k_1^{(1)} = -\frac{1}{R_1 A_1} \frac{\partial \phi_1}{2 \partial \alpha_1} + \frac{\partial u}{R_1 \partial \alpha_1} + \frac{\partial A_1}{2 A_1^2 \partial \alpha_1} \frac{\partial w}{\partial \alpha_1} - \frac{\partial^2 w}{2 A_1 \partial \alpha_1^2} \left(-\frac{\partial A_1}{A_1 A_2 \partial \alpha_2} \left[\phi_2 \left(\frac{1}{R_1} - \frac{1}{2 R_2} \right) + \frac{v}{R_1 R_2} - \frac{\partial w}{2 R_2 A_2 \partial \alpha_2} \right] \right), \quad (18b)$$

$$k_1^{(2)} = -\frac{4}{3 h^2} \left(\frac{\partial \phi_1}{A_1 \partial \alpha_1} + \frac{\partial^2 w}{A_1^2 \partial \alpha_1^2} + \frac{\phi_2}{A_1 A_2} \frac{\partial A_1}{\partial \alpha_2} - \frac{\partial w}{A_1^3 \partial \alpha_1} \frac{\partial A_1}{\partial \alpha_1} + \frac{\partial w}{A_1 A_2^2 \partial \alpha_2} \frac{\partial A_1}{\partial \alpha_2} \right) - \frac{\phi_2}{A_1 A_2} \frac{\partial A_1}{\partial \alpha_2} \left(\frac{1}{6 R_2^2} + \frac{1}{2 R_1 R_2} \right) + \frac{\partial w}{A_1 A_2^2 \partial \alpha_2} \frac{\partial A_1}{\partial \alpha_2} \left(\frac{1}{6 R_2^2} - \frac{1}{2 R_1 R_2} \right) - \frac{1}{3 R_1^2} \frac{2 \partial \phi_1}{A_1 \partial \alpha_1} + \frac{\partial^2 w}{A_1^2 \partial \alpha_1^2} - \frac{\partial w}{A_1^3 \partial \alpha_1} \frac{\partial A_1}{\partial \alpha_1}, \quad (18c)$$

$$k_2^{(0)} = \frac{\partial \phi_2}{A_2 \partial \alpha_2} - \frac{w}{R_2^2} + \frac{u}{A_1 A_2} \frac{\partial A_2}{\partial \alpha_1} \left(\frac{1}{R_1} - \frac{1}{R_2} \right) + \frac{\phi_1}{A_1 A_2} \frac{\partial A_2}{\partial \alpha_1}, \quad (18d)$$

$$k_2^{(1)} = -\frac{1}{R_2 A_2} \frac{\partial \phi_2}{2 \partial \alpha_2} + \frac{\partial v}{R_2 \partial \alpha_2} + \frac{\partial A_2}{2 A_2^2 \partial \alpha_2} \frac{\partial w}{\partial \alpha_2} - \frac{\partial^2 w}{2 A_2 \partial \alpha_2^2} \left(-\frac{\partial A_2}{A_1 A_2 \partial \alpha_1} \left[\phi_1 \left(\frac{1}{R_2} - \frac{1}{2 R_1} \right) + \frac{u}{R_1 R_2} - \frac{\partial w}{2 R_1 A_1 \partial \alpha_1} \right] \right), \quad (18e)$$

$$k_2^{(2)} = -\frac{4}{3 h^2} \left(\frac{\partial \phi_2}{A_2 \partial \alpha_2} + \frac{\partial^2 w}{A_2^2 \partial \alpha_2^2} + \frac{\phi_1}{A_1 A_2} \frac{\partial A_2}{\partial \alpha_1} - \frac{\partial w}{A_2^3 \partial \alpha_2} \frac{\partial A_2}{\partial \alpha_2} + \frac{\partial w}{A_1^2 A_2 \partial \alpha_1} \frac{\partial A_2}{\partial \alpha_1} \right) - \frac{\phi_1}{A_1 A_2} \frac{\partial A_2}{\partial \alpha_1} \left(\frac{1}{6 R_1^2} + \frac{1}{2 R_1 R_2} \right) + \frac{\partial w}{A_1^2 A_2 \partial \alpha_1} \frac{\partial A_2}{\partial \alpha_1} \left(\frac{1}{6 R_1^2} - \frac{1}{2 R_1 R_2} \right) - \frac{1}{3 R_2^2} \frac{2 \partial \phi_2}{A_2 \partial \alpha_2} + \frac{\partial^2 w}{A_2^2 \partial \alpha_2^2} - \frac{\partial w}{A_2^3 \partial \alpha_2} \frac{\partial A_2}{\partial \alpha_2}, \quad (18f)$$

$$k_{12}^{(0)} = \frac{\partial \phi_1}{A_2 \partial \alpha_2} + \frac{\partial \phi_2}{A_1 \partial \alpha_1} - \frac{\phi_1}{A_1 A_2} \frac{\partial A_1}{\partial \alpha_2} - \frac{\phi_2}{A_1 A_2} \frac{\partial A_2}{\partial \alpha_1} + \frac{\partial u}{A_2 \partial \alpha_2} \left(\frac{1}{R_1} - \frac{1}{R_2} \right) + \frac{\partial v}{A_1 \partial \alpha_1} \left(-\frac{1}{R_1} + \frac{1}{R_2} \right), \quad (18g)$$

$$k_{12}^{(1)} = -\frac{\partial \phi_1}{A_2 \partial \alpha_2} \left(\frac{1}{R_2} - \frac{1}{2 R_1} \right) - \frac{\partial \phi_2}{A_1 \partial \alpha_1} \left(\frac{1}{R_1} - \frac{1}{2 R_2} \right) - \frac{1}{R_1 R_2} \left(\frac{\partial u}{A_2 \partial \alpha_2} + \frac{\partial v}{A_1 \partial \alpha_1} \right) + \frac{\partial^2 w}{2 A_1 A_2 \partial \alpha_1 \partial \alpha_2} \left(\frac{1}{R_1} + \frac{1}{R_2} \right) + \frac{1}{A_1 A_2} \left(\frac{\phi_1}{2 R_1} \frac{\partial A_1}{\partial \alpha_2} + \frac{\phi_2}{2 R_2} \frac{\partial A_2}{\partial \alpha_1} + \frac{u}{R_1^2} \frac{\partial A_1}{\partial \alpha_2} + \frac{v}{R_2^2} \frac{\partial A_2}{\partial \alpha_1} - \frac{\partial w}{R_1 A_1 \partial \alpha_1} \frac{\partial A_1}{\partial \alpha_2} - \frac{\partial w}{R_2 A_2 \partial \alpha_2} \frac{\partial A_2}{\partial \alpha_1} \right), \quad (18h)$$

$$k_{12}^{(2)} = -\frac{4}{3 h^2} \left(\frac{\partial \phi_1}{A_2 \partial \alpha_2} + \frac{\partial \phi_2}{A_1 \partial \alpha_1} + 2 \frac{\partial^2 w}{A_1 A_2 \partial \alpha_1 \partial \alpha_2} - 2 \frac{\partial w}{A_1^2 A_2 \partial \alpha_1} \frac{\partial A_1}{\partial \alpha_2} - 2 \frac{\partial w}{A_1 A_2^2 \partial \alpha_2} \frac{\partial A_2}{\partial \alpha_1} - \frac{\phi_1}{A_1 A_2} \frac{\partial A_1}{\partial \alpha_2} - \frac{\phi_2}{A_1 A_2} \frac{\partial A_2}{\partial \alpha_1} \right) + \frac{\phi_1}{A_1 A_2} \frac{\partial A_1}{\partial \alpha_2} \frac{2}{3 R_1^2} + \frac{\phi_2}{A_1 A_2} \frac{\partial A_2}{\partial \alpha_1} \frac{2}{3 R_2^2} - \frac{\partial \phi_1}{A_2 \partial \alpha_2} \left(\frac{1}{6 R_1^2} + \frac{1}{2 R_1 R_2} \right) - \frac{\partial \phi_2}{A_1 \partial \alpha_1} \left(\frac{1}{6 R_2^2} + \frac{1}{2 R_1 R_2} \right)$$

$$+ \frac{\partial w}{A_1^2 A_2 \partial \alpha_1 \partial \alpha_2} \left(\frac{1}{6 R_1^2} + \frac{1}{2 R_1 R_2} \right) + \frac{\partial w}{A_1 A_2^2 \partial \alpha_2 \partial \alpha_1} \left(\frac{1}{6 R_2^2} + \frac{1}{2 R_1 R_2} \right) + \frac{\partial^2 w}{A_1 A_2 \partial \alpha_1 \partial \alpha_2} \left(\frac{1}{6 R_1^2} + \frac{1}{6 R_2^2} - \frac{1}{R_1 R_2} \right). \quad (18i)$$

Eqs. (18a–i) give the changes in curvature and torsion of the middle surface, and they have been obtained retaining only linear terms; in fact, non-linear terms in the changes in curvature and torsion play a very small role, at least for moderate vibration amplitudes of the order of the shell thickness [12]. Eqs. (17a–c), giving the middle surface strains, are coincident with those obtained by using Novozhilov non-linear shell theory [2,13], which neglects shear deformation and rotary inertia. Moreover, it can be observed that $k_1^{(1)}$, $k_2^{(1)}$, $k_{12}^{(1)}$ are negligible for shells that are not very thick.

3. Elastic strain and kinetic energies, including rotary inertia, for laminated shells

It is assumed that the transverse normal stress $\sigma_3 = 0$ is negligible (plane stress assumption); in general, it is verified that σ_3 is small compared to τ_{13} and τ_{23} , except near the shell edges, so that the hypothesis is a good approximation of the actual behavior of moderately thick shells. The stress–strain relations for the k -th orthotropic lamina of the shell, in the material principal coordinates (x, y, z) , are obtained under the hypothesis $\sigma_3 = 0$ [2],

$$\begin{Bmatrix} \sigma_x \\ \sigma_y \\ \tau_{yz} \\ \tau_{xz} \\ \tau_{xy} \end{Bmatrix}^{(k)} = \begin{bmatrix} c_{11} & c_{12} & 0 & 0 & 0 \\ c_{21} & c_{22} & 0 & 0 & 0 \\ 0 & 0 & G_{yz} & 0 & 0 \\ 0 & 0 & 0 & G_{xz} & 0 \\ 0 & 0 & 0 & 0 & G_{xy} \end{bmatrix}^{(k)} \begin{Bmatrix} \varepsilon_{xx} \\ \varepsilon_{yy} \\ \gamma_{yz} \\ \gamma_{xz} \\ \gamma_{xy} \end{Bmatrix}, \quad (19)$$

where G_{xy} , G_{xz} and G_{yz} are the shear moduli in x – y , x – z and y – z directions, respectively, and the coefficients c_{ij} are given in Appendix A and in Ref. [11]; τ_{xz} and τ_{yz} are the shear stresses and the superscript (k) refers to the k -th layer within a laminate. Eq. (19) is obtained (i) under the transverse isotropy assumption with respect to planes orthogonal to the x axis, i.e. assuming fibers in the direction parallel to axis x , so that $E_y = E_z$, $G_{xy} = G_{xz}$ and $\nu_{xy} = \nu_{xz}$, and (ii) solving the constitutive equations for ε_{zz} as function of ε_{xx} and ε_{yy} and then eliminating it.

Eq. (19) can be transformed to the shell coordinates (α_1, α_2, z) by the following equation [2]:

$$\begin{Bmatrix} \sigma_1 \\ \sigma_2 \\ \tau_{23} \\ \tau_{13} \\ \tau_{12} \end{Bmatrix}^{(k)} = [Q]^{(k)} \begin{Bmatrix} \varepsilon_{11} \\ \varepsilon_{22} \\ \gamma_{23} \\ \gamma_{13} \\ \gamma_{12} \end{Bmatrix}, \quad (20)$$

where $[Q]^{(k)}$ is the 5×5 matrix of the material properties of the k -th layer expressed in the shell principal coordinates and it is given in Appendix A. Eq. (20) can be rewritten as

$$\{\sigma^{(k)}\} = [Q^{(k)}] \left[\begin{Bmatrix} \varepsilon_{1,0} \\ \varepsilon_{2,0} \\ \gamma_{23,0} \\ \gamma_{13,0} \\ \gamma_{12,0} \end{Bmatrix} + z \begin{Bmatrix} k_1^{(0)} \\ k_2^{(0)} \\ 0 \\ 0 \\ k_{12}^{(0)} \end{Bmatrix} + z^2 \begin{Bmatrix} 0 \\ 0 \\ k_{23}^{(1)} \\ k_{13}^{(1)} \\ 0 \end{Bmatrix} + z^3 \begin{Bmatrix} k_1^{(2)} \\ k_2^{(2)} \\ 0 \\ 0 \\ k_{12}^{(2)} \end{Bmatrix} \right]. \quad (21)$$

In Eq. (21), $k_1^{(1)}$, $k_2^{(1)}$, $k_{12}^{(1)}$ have been neglected, supposing a moderately thick shell.

The elastic strain energy U_S of the shell is given by

$$U_S = \frac{1}{2} \sum_{k=1}^K \int_0^a \int_0^b \int_{h^{(k-1)}}^{h^{(k)}} (\sigma_1^{(k)} \varepsilon_{11} + \sigma_2^{(k)} \varepsilon_{22} + \tau_{12}^{(k)} \gamma_{12} + \tau_{13}^{(k)} \gamma_{13} + \tau_{23}^{(k)} \gamma_{23}) \times (1+z/R_1)(1+z/R_2) d\alpha_1 d\alpha_2 dz, \quad (22)$$

where K is the total number of layers in the laminated shell and $(h^{(k-1)}, h^{(k)})$ are the z coordinates of the k -th layer. For simplicity a shell of rectangular base of dimension a and b in α_1 and α_2 directions, respectively, has been considered.

The kinetic energy T_S of the shell, including rotary inertia, is given by

$$T_S = \frac{1}{2} \sum_{k=1}^K \rho_S^{(k)} \int_0^a \int_0^b \int_{h^{(k-1)}}^{h^{(k)}} (\dot{u}_1^2 + \dot{u}_2^2 + \dot{u}_3^2)(1+z/R_1)(1+z/R_2) d\alpha_1 d\alpha_2 dz. \quad (23)$$

The z and z^3 terms vanish after integration on z in the case of a laminate with symmetric density with respect to the z axis. In particular, for a laminate with the same density for all the layers and uniform thickness, the following simplified expression is obtained:

$$T_S = \frac{1}{2} \rho_S h \int_0^a \int_0^b \left\{ \dot{u}^2 + \dot{v}^2 + \dot{w}^2 + h^2 \left[\frac{17}{315} (\dot{\phi}_1^2 + \dot{\phi}_2^2) + \dot{\phi}_1 \dot{u} \left(\frac{41}{120R_1} + \frac{2}{15R_2} \right) + \dot{\phi}_2 \dot{v} \left(\frac{2}{15R_1} + \frac{41}{120R_2} \right) + \frac{1}{4} \left(\frac{\dot{u}^2}{R_1} + \frac{\dot{v}^2}{R_2} \right) \left(\frac{1}{R_1} + \frac{1}{R_2} \right) + \frac{\dot{w}^2}{12R_1R_2} + \frac{\partial \dot{w}}{A_1 \partial \alpha_1} \frac{\partial \dot{w}}{252A_1 \partial \alpha_1} + \frac{\dot{u}}{120R_1} - \frac{\dot{u}}{30R_2} - 8 \frac{\dot{\phi}_1}{315} \right) + \frac{\partial \dot{w}}{A_2 \partial \alpha_2} \frac{\partial \dot{w}}{252A_2 \partial \alpha_2} + \frac{\dot{v}}{120R_2} - \frac{\dot{v}}{30R_1} - 8 \frac{\dot{\phi}_2}{315} \right] \right\} d\alpha_1 d\alpha_2 + O(h^5). \quad (24)$$

4. Boundary conditions and discretization of a circular cylindrical shell

In order to reduce the system to finite dimensions, the middle surface displacements u , v and w are expanded by using approximate functions. Circular cylindrical shells with simply supported boundary conditions are analyzed in the following part of the study, as shown in Fig. 2. In particular, $R_1 = \infty$, $R_2 = R$, $\alpha_1 = x$, $\alpha_2 = \theta$, $A_1 = 1$, $A_2 = R$, $a = L$ and $b = 2\pi$. The following boundary conditions are imposed at the shell ends, $x=0, L$:

$$w = 0, \quad v = 0, \quad \phi_2 = 0, \quad (25a-c)$$

$$N_x = 0, \quad M_x = 0, \quad (25d, e)$$

where N_x is the axial stress resultant per unit length and M_x is the axial stress moment resultant per unit length, i.e.,

$$\begin{Bmatrix} N_x \\ M_x \end{Bmatrix} = \sum_{k=1}^K \int_{h^{(k-1)}}^{h^{(k)}} \sigma_x^{(k)} \begin{Bmatrix} 1 \\ z \end{Bmatrix} (1+z/R) dz. \quad (26)$$

Moreover, the three displacements and the two rotations must be continuous in θ .

The following base of shell displacements, which satisfy identically the geometric boundary conditions (25a-c), is used

to discretize the system:

$$u(x, \theta, t) = \sum_{m=1}^{M_1} \sum_{j=1}^N [u_{m,j,c}(t) \cos(j\theta) + u_{m,j,s}(t) \sin(j\theta)] \cos(\lambda_m x) + \sum_{m=1}^{M_2} u_{m,0}(t) \cos(\lambda_m x), \quad (27a)$$

$$v(x, \theta, t) = \sum_{m=1}^{3M_1} \sum_{j=1}^{2N} [v_{m,j,c}(t) \sin(j\theta) + v_{m,j,s}(t) \cos(j\theta)] \sin(\lambda_m x), \quad (27b)$$

$$w(x, \theta, t) = \sum_{m=1}^{M_1} \sum_{j=1}^N [w_{m,j,c}(t) \cos(j\theta) + w_{m,j,s}(t) \sin(j\theta)] \sin(\lambda_m x) + \sum_{m=1}^{M_2} w_{m,0}(t) \sin(\lambda_m x), \quad (27c)$$

$$\phi_1(x, \theta, t) = \sum_{m=1}^{M_1} \sum_{j=1}^N [\phi_{1,m,n,c}(t) \cos(j\theta) + \phi_{1,m,n,s}(t) \sin(j\theta)] \cos(\lambda_m x) + \sum_{m=1}^{M_2} \phi_{1,m,0}(t) \cos(\lambda_m x), \quad (27d)$$

$$\phi_2(x, y, t) = \sum_{m=1}^{M_1} \sum_{j=1}^N [\phi_{2,m,n,c}(t) \sin(j\theta) + \phi_{2,m,n,s}(t) \cos(j\theta)] \sin(\lambda_m x), \quad (27e)$$

where j is the number of circumferential waves, m is the number of longitudinal half-waves, $\lambda_m = m\pi/L$ and t is the time; $u_{m,j}(t)$, $v_{m,j}(t)$, $w_{m,j}(t)$, $\phi_{1,m,j}$ and $\phi_{2,m,j}$ are the generalized coordinates that are unknown functions of t ; the additional subscript c or s indicates if the generalized coordinate is associated to cosine or sine function in θ except for v , for which the notation is reversed (no additional subscript is used for axisymmetric terms). The integers N , M_1 and M_2 must be selected with care in order to obtain the required accuracy and acceptable dimension of the non-linear problem. By using a different number of terms in the expansions, it is possible to study the convergence and the accuracy of the solution.

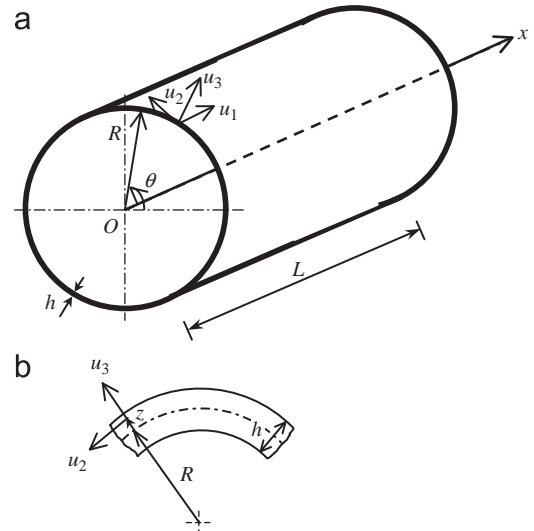


Fig. 2. Circular cylindrical shells: (a) symbols used for dimensions and displacement of a generic point; (b) cross-section of the shell with close-up view.

The minimal expansion giving accurate results for excitation in the neighborhood of resonance of mode ($m=1, n$) is

$$u(x, \theta, t) = [u_{1,n,c}(t)\cos(n\theta) + u_{1,n,s}(t)\sin(n\theta)]\cos(\lambda_1 x) + \sum_{m=1}^3 u_{2m-1,0}(t)\cos(\lambda_{2m-1} x), \quad (28a)$$

$$v(x, \theta, t) = \sum_{k=1}^2 [v_{1,kn,c}(t)\sin(kn\theta) + v_{1,kn,s}(t)\cos(kn\theta)]\sin(\lambda_1 x) + [v_{3,2n,c}(t)\sin(2n\theta) + v_{3,2n,s}(t)\cos(2n\theta)]\sin(\lambda_3 x), \quad (28b)$$

$$w(x, \theta, t) = [w_{1,n,c}(t)\cos(n\theta) + w_{1,n,s}(t)\sin(n\theta)]\sin(\lambda_1 x) + \sum_{m=1}^3 w_{2m-1,0}(t)\sin(\lambda_{2m-1} x), \quad (28c)$$

$$\phi_1(x, \theta, t) = [\phi_{1,n,c}(t)\cos(n\theta) + \phi_{1,n,s}(t)\sin(n\theta)]\cos(\lambda_1 x) + \sum_{m=1}^3 \phi_{1_{2m-1,0}}(t)\cos(\lambda_{2m-1} x), \quad (28d)$$

$$\phi_2(x, \theta, t) = [\phi_{2,n,c}(t)\sin(n\theta) + \phi_{2,n,s}(t)\cos(n\theta)]\sin(\lambda_1 x). \quad (28e)$$

This expansion has 23 generalized coordinates (degrees of freedom) and guarantees good accuracy for the calculation performed in the present work.

Initial geometric imperfections of the shell are considered only in radial direction. They are assumed to be associated with zero initial stress. The imperfection w_0 is expanded in the same form of w

$$w_0(x, \theta) = \sum_{m=1}^{\tilde{M}_1} \sum_{n=1}^{\tilde{N}} [A_{m,n} \cos(n\theta) + B_{m,n} \sin(n\theta)]\sin(\lambda_m x) + \sum_{m=1}^{\tilde{M}_2} A_{m,0} \sin(\lambda_m x), \quad (29)$$

where $A_{m,n}$, $B_{m,n}$ and $A_{m,0}$ are the modal amplitudes of imperfections; \tilde{N} , \tilde{M}_1 and \tilde{M}_2 are integers indicating the number of terms in the expansion.

Eq. (25d) is not identically satisfied by the shell displacements given in Eqs. (27) and (28). In particular, after integration with respect to z as shown in Eq. (26), it becomes

$$N_x = \sum_{k=1}^N \{Q_{11}^{(k)}, Q_{12}^{(k)}, Q_{15}^{(k)}\} \left[\begin{array}{l} \varepsilon_{x,0} \\ \varepsilon_{\theta,0} \\ \gamma_{x\theta,0} \end{array} \right] \left(h^{(k)} - h^{(k-1)} + \frac{h^{(k)2} - h^{(k-1)2}}{2R} \right) + \left\{ \begin{array}{l} k_x^{(0)} \\ k_\theta^{(0)} \\ k_{x\theta}^{(0)} \end{array} \right\} \left(\frac{h^{(k)2} - h^{(k-1)2}}{2} + \frac{h^{(k)3} - h^{(k-1)3}}{3R} \right) + \left\{ \begin{array}{l} k_x^{(2)} \\ k_\theta^{(2)} \\ k_{x\theta}^{(2)} \end{array} \right\} \left(\frac{h^{(k)4} - h^{(k-1)4}}{4} + \frac{h^{(k)5} - h^{(k-1)5}}{5R} \right) = 0 \quad \text{at } x=0, L, \quad (30)$$

where $Q_{11}^{(k)}$, $Q_{12}^{(k)}$, $Q_{15}^{(k)}$ are given by Eqs. (A.6)–(A.8) in Appendix A and the strain–displacement relations are given in Appendix B. In order to satisfy Eqs. (30) up to second-order non-linear terms, a second-order term $\hat{u}(x, \theta, t)$ must be added to the expansion of u in Eqs. (27a) and (28a). In particular, eliminating in Eq. (30) null

terms at $x=0, L$, the following expression is obtained:

$$N_x = \sum_{k=1}^N \{Q_{11}^{(k)}, Q_{12}^{(k)}, Q_{15}^{(k)}\} \times \left[\begin{array}{l} \frac{\partial \hat{u}}{\partial x} + \frac{1}{2} \left[\left(\frac{\partial v}{\partial x} \right)^2 + \left(\frac{\partial w}{\partial x} \right)^2 \right] + \frac{\partial w}{\partial x} \frac{\partial w_0}{\partial x} \\ \frac{1}{2} \left(\frac{1}{R} \frac{\partial u}{\partial \theta} \right)^2 \\ \frac{\partial v}{\partial x} + \frac{1}{R} \frac{\partial u}{\partial \theta} \end{array} \right] \left(h^{(k)} - h^{(k-1)} + \frac{h^{(k)2} - h^{(k-1)2}}{2R} \right) + \left\{ \begin{array}{l} 0 \\ 0 \\ k_{x\theta}^{(0)} \end{array} \right\} \left(\frac{h^{(k)2} - h^{(k-1)2}}{2} + \frac{h^{(k)3} - h^{(k-1)3}}{3R} \right) + \left\{ \begin{array}{l} 0 \\ 0 \\ k_{x\theta}^{(2)} \end{array} \right\} \left(\frac{h^{(k)4} - h^{(k-1)4}}{4} + \frac{h^{(k)5} - h^{(k-1)5}}{5R} \right) = 0 \quad \text{at } x=0, L, \quad (31a)$$

where the term $\hat{u}(x, \theta, t)$ has been neglected in second-order terms. In Eq. (31), all the linear terms $(\partial v / \partial x) + (1/R)(\partial u / \partial \theta)$, $k_{x\theta}^{(0)}$ and $k_{x\theta}^{(2)}$ can be eliminated since they establish a linear relationship which is satisfied by using the minimization of energy in the process of building the Lagrange equations of motion; in fact, this is equivalent to the Rayleigh–Ritz method and therefore it is necessary only to satisfy only geometrical boundary conditions. Therefore Eq. (31a) can be simplified into

$$N_x = \sum_{k=1}^N \{Q_{11}^{(k)}, Q_{12}^{(k)}\} \left[\begin{array}{l} \frac{\partial \hat{u}}{\partial x} + \frac{1}{2} \left[\left(\frac{\partial v}{\partial x} \right)^2 + \left(\frac{\partial w}{\partial x} \right)^2 \right] + \frac{\partial w}{\partial x} \frac{\partial w_0}{\partial x} \\ \frac{1}{2} \left(\frac{1}{R} \frac{\partial u}{\partial \theta} \right)^2 \end{array} \right] \times \left(h^{(k)} - h^{(k-1)} + \frac{h^{(k)2} - h^{(k-1)2}}{2R} \right) = 0 \quad \text{at } x=0, L, \quad (31b)$$

which immediately gives

$$N_x = \sum_{k=1}^N \left(h^{(k)} - h^{(k-1)} + \frac{h^{(k)2} - h^{(k-1)2}}{2R} \right) \left[Q_{11}^{(k)} \frac{\partial \hat{u}}{\partial x} + \frac{1}{2} \left[\left(\frac{\partial v}{\partial x} \right)^2 + \left(\frac{\partial w}{\partial x} \right)^2 \right] + \frac{\partial w}{\partial x} \frac{\partial w_0}{\partial x} \right] + Q_{12}^{(k)} \frac{1}{2} \left(\frac{1}{R} \frac{\partial u}{\partial \theta} \right)^2 = 0 \quad \text{at } x=0, L. \quad (31c)$$

By introducing the notation

$$D_{11} = \sum_{k=1}^N Q_{11}^{(k)} \left(h^{(k)} - h^{(k-1)} + \frac{h^{(k)2} - h^{(k-1)2}}{2R} \right),$$

$$D_{12} = \sum_{k=1}^N Q_{12}^{(k)} \left(h^{(k)} - h^{(k-1)} + \frac{h^{(k)2} - h^{(k-1)2}}{2R} \right),$$

Eq. (31c) can be expressed in the following form

$$N_x = D_{11} \left(\frac{\partial \hat{u}}{\partial x} + \frac{1}{2} \left[\left(\frac{\partial v}{\partial x} \right)^2 + \left(\frac{\partial w}{\partial x} \right)^2 \right] + \frac{\partial w}{\partial x} \frac{\partial w_0}{\partial x} \right) + D_{12} \frac{1}{2} \left(\frac{1}{R} \frac{\partial u}{\partial \theta} \right)^2 = 0 \quad \text{at } x=0, L. \quad (31d)$$

In Eq. (31d) it is necessary to retain only the resonant mode (m,n) . The expression of \hat{u} can be obtained from Eq. (31d) as

$$\begin{aligned} \hat{u}(t) = & -\frac{1}{8}[a(t)+b(t)\cos(2n\theta)+c(t)\sin(2n\theta)]\sin(2m\pi x/L)-(m\pi/L) \\ & \times [w_{m,n,c}(t)\cos(n\theta)+w_{m,n,s}(t)\sin(n\theta)] \sum_{j=0}^{\tilde{N}} \sum_{i=1}^{\tilde{M}_1} \frac{i}{m+i} [A_{ij} \cos(j\theta) \\ & + B_{ij} \sin(j\theta)] \sin[(m+i)\pi x/L], \end{aligned} \quad (32)$$

where

$$\begin{aligned} a(t) = & (m\pi/L)(w_{m,n,c}^2 + w_{m,n,s}^2 + v_{m,n,c}^2 + v_{m,n,s}^2) \\ & + (D_{12}/D_{11})[Ln^2/(m\pi R^2)](u_{m,n,c}^2 + u_{m,n,s}^2), \end{aligned}$$

$$\begin{aligned} b(t) = & (m\pi/L)(w_{m,n,c}^2 - w_{m,n,s}^2 + v_{m,n,c}^2 - v_{m,n,s}^2) \\ & + (D_{12}/D_{11})[Ln^2/(m\pi R^2)](u_{m,n,s}^2 - u_{m,n,c}^2), \end{aligned}$$

$$\begin{aligned} c(t) = & (2m\pi/L)(w_{m,n,c}w_{m,n,s} + v_{m,n,c}v_{m,n,s}) \\ & - 2(D_{12}/D_{11})[Ln^2/(m\pi R^2)]u_{m,n,c}u_{m,n,s}. \end{aligned}$$

The boundary condition (25e) is identically satisfied for symmetric laminates if the term z/R is neglected in Eq. (26), i.e. for thin shells. This is due to the expressions of $k_x^{(0)}$ and $k_x^{(2)}$ given in Appendix B, which are zero at $x=0, L$ for the expansions assumed in Eqs. (27a–e). Additional terms must be added to the expansion of the in-plane displacement u for asymmetric laminates and moderately thick shells. In fact, bending and stretching are coupled for asymmetric laminates.

5. Lagrange equations of motion

The virtual work W done by the external forces is written as

$$W = \int_0^L \int_0^{2\pi} (q_x u + q_y v + q_z w) dx R d\theta, \quad (33)$$

where q_x, q_θ and q_z are the distributed forces per unit area acting in x, θ and z directions, respectively, applied at the middle surface. Only a single harmonic force orthogonal to the shell is considered; therefore, $q_x = q_\theta = 0$. The external distributed load q_z applied to the shell, due to the normal concentrated force \tilde{f} , is given by

$$q_z = \tilde{f} \delta(x - \tilde{x}) \delta(\theta - \tilde{\theta}) \cos(\omega t), \quad (34)$$

where ω is the excitation frequency, t is the time, δ is the Dirac delta function, \tilde{f} gives the force magnitude positive in z direction and \tilde{x} and $\tilde{\theta}$ give the position of the point of application of the force. Here, the point excitation is located at the center of shell, that is, $\tilde{x} = L/2, \tilde{\theta} = 0$. Eq. (33) can be rewritten in the following form:

$$W = \tilde{f} \cos(\omega t) (w)_{x=L/2, \theta=0}. \quad (35)$$

The non-conservative damping forces are assumed to be of viscous type and are taken into account by using the Rayleigh dissipation function

$$F = \frac{1}{2} c \int_0^L \int_0^{2\pi} (\dot{u}^2 + \dot{v}^2 + \dot{w}^2 + h^2 \dot{\phi}_1^2 + h^2 \dot{\phi}_2^2) dx R d\theta, \quad (36)$$

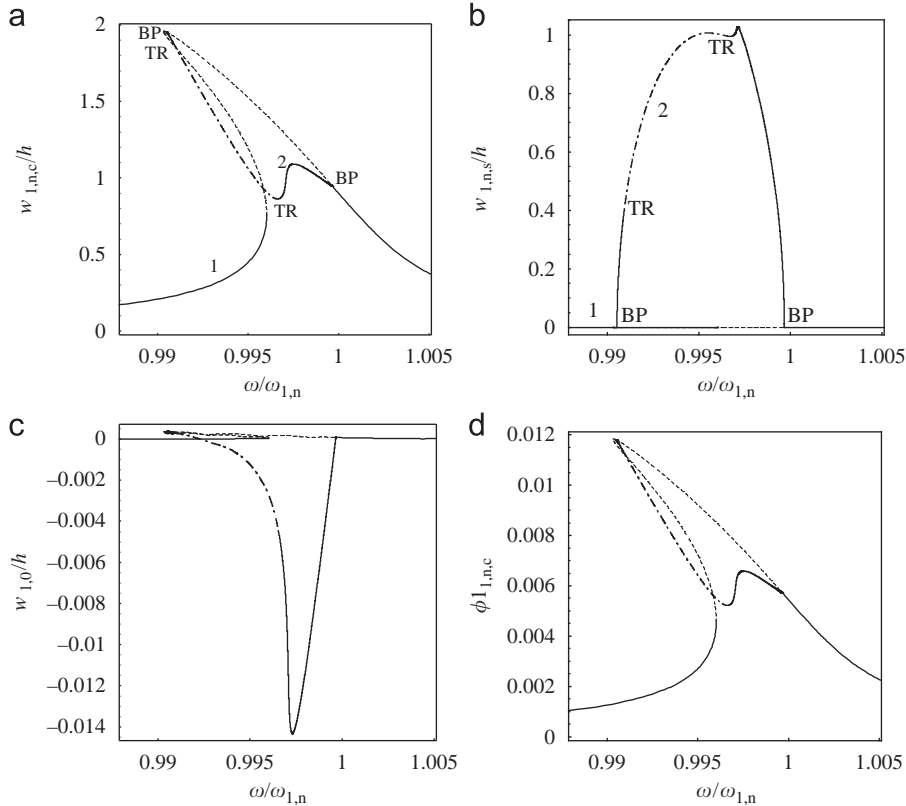


Fig. 3. Frequency–response curve of the circular cylindrical shell obtained by using the proposed non-linear theory; $n=4$, 23 degrees of freedom; —, stable periodic solution; - - -, stable quasi-periodic solution; ···, unstable solutions; BP, pitchfork bifurcation; TR, Neimark–Sacker bifurcation. (a) Maximum of the generalized coordinate $w_{1,n,c}$ (driven mode) normalized with respect to the shell thickness h ; (b) maximum of the generalized coordinate $w_{1,n,s}$ (companion mode) normalized with respect to the shell thickness h ; (c) maximum of the generalized coordinate $w_{1,0}$ (first axisymmetric mode) normalized with respect to the shell thickness h ; (d) maximum of the generalized coordinate $\phi_{1,n,c}$ (rotation about the curvilinear coordinate y).

where c has a different value for each term of the mode expansion; in particular

$$F = \frac{1}{2} \frac{L\pi}{2} \left[\sum_{n=1}^N \sum_{m=1}^M c_{m,n} \frac{\dot{u}_{m,n}^2 + \dot{v}_{m,n}^2}{h^2} + \sum_{n=1}^{\hat{N}} \sum_{m=1}^{\hat{M}} c_{m,n} \left(\frac{\dot{w}_{m,n}^2}{h^2} + \dot{\phi}_{1,m,n}^2 + \dot{\phi}_{2,m,n}^2 \right) \right]. \quad (37)$$

In Eq. (37) displacements are non-dimensionalized dividing by h , while rotations are already non-dimensional. The damping coefficient $c_{m,n}$ is related to the modal damping ratio that can be evaluated from experiments by $\zeta_{m,n} = c_{m,n}/(2\mu_{m,n}\omega_{m,n})$, where $\omega_{m,n}$ is the natural circular frequency of mode (m, n) and $\mu_{m,n}$ is the modal mass of this mode.

The following notation is introduced for brevity:

$$\mathbf{q} = \{u_{m,n}, v_{m,n}, w_{m,n}, \phi_{1,m,n}, \phi_{2,m,n}\}^T, \quad m = 1, \dots, M \text{ or } \hat{M} \text{ and } n = 1, \dots, N \text{ or } \hat{N}. \quad (38)$$

The generic element of the time-dependent vector \mathbf{q} is referred to as q_j ; the dimension of \mathbf{q} is \bar{N} , which is the number of degrees of freedom (*dofs*) used in the mode expansion.

The generalized forces Q_j are obtained by differentiation of the Rayleigh dissipation function and of the virtual work done by external forces:

$$Q_j = -\frac{\partial F}{\partial \dot{q}_j} + \frac{\partial W}{\partial q_j}. \quad (39)$$

The Lagrange equations of motion are

$$\frac{d}{dt} \left(\frac{\partial T_P}{\partial \dot{q}_j} \right) - \frac{\partial T_P}{\partial q_j} + \frac{\partial U_P}{\partial q_j} = Q_j, \quad j = 1, \dots, \bar{N}, \quad (40)$$

where $\partial T_P / \partial q_j = 0$. The complicated term, derived from the maximum potential energy of the plate, giving quadratic and cubic non-linearities, can be written in the form

$$\frac{\partial U}{\partial q_j} = \sum_{i=1}^{\bar{N}} f_{j,i} q_i + \sum_{i,k=1}^{\bar{N}} f_{j,i,k} q_i q_k + \sum_{i,k,l=1}^{\bar{N}} f_{j,i,k,l} q_i q_k q_l, \quad j = 1, \dots, \bar{N}, \quad (41)$$

where the coefficients f have long expressions that include also geometric imperfections. It is interesting to observe that in Eq. (41) there are quadratic and cubic terms.

5.1. Inertial coupling in the equations of motion

For shells with rotary inertia, inertial coupling arises in the equations of motion (see Eqs. (23) and (24)) so that they cannot be immediately transformed in the form required for numerical integration. In particular, the equations of motion take the following form:

$$\mathbf{M}\ddot{\mathbf{q}} + \mathbf{C}\dot{\mathbf{q}} + [\mathbf{K} + \mathbf{N}_2(\mathbf{q}) + \mathbf{N}_3(\mathbf{q}, \mathbf{q})]\mathbf{q} = \mathbf{f}_0 \cos(\omega t), \quad (42)$$

where \mathbf{M} is the non-diagonal mass matrix of dimension $\bar{N} \times \bar{N}$ (\bar{N} being the number of degrees of freedom), \mathbf{C} is the damping matrix, \mathbf{K} is the linear stiffness matrix, which does not present terms involving \mathbf{q} , \mathbf{N}_2 is a matrix that involves linear terms in \mathbf{q} , therefore giving the quadratic non-linear stiffness terms, \mathbf{N}_3 is a matrix that involves quadratic terms in \mathbf{q} , therefore giving the cubic non-linear stiffness terms, \mathbf{f}_0 is the vector of excitation amplitudes and \mathbf{q} is the vector of the \bar{N} generalized coordinates, defined in Eq. (38). In particular, by using Eq. (41), the generic elements $k_{j,i}$, $n_{2,ji}$ and $n_{3,ji}$, of the matrices \mathbf{K} , \mathbf{N}_2 and \mathbf{N}_3 , respectively, are given by

$$k_{j,i} = f_{j,i}, \quad n_{2,ji}(\mathbf{q}) = \sum_{k=1}^{\bar{N}} f_{j,i,k} q_k, \quad n_{3,ji}(\mathbf{q}, \mathbf{q}) = \sum_{k,l=1}^{\bar{N}} f_{j,i,k,l} q_k q_l. \quad (43a-c)$$

Eq. (42) is pre-multiplied by \mathbf{M}^{-1} in order to diagonalize the mass matrix, as a consequence that the matrix \mathbf{M} is always invertible; the result is

$$\ddot{\mathbf{q}} + \mathbf{M}^{-1}\mathbf{C}\dot{\mathbf{q}} + [\mathbf{M}^{-1}\mathbf{K} + \mathbf{M}^{-1}\mathbf{N}_2(\mathbf{q}) + \mathbf{M}^{-1}\mathbf{N}_3(\mathbf{q}, \mathbf{q})]\mathbf{q} = \mathbf{M}^{-1}\mathbf{f}_0 \cos(\omega t), \quad (44)$$

which can be rewritten in the following form

$$\ddot{\mathbf{q}} + \tilde{\mathbf{C}}\dot{\mathbf{q}} + [\tilde{\mathbf{K}} + \mathbf{M}^{-1}\mathbf{N}_2(\mathbf{q}) + \mathbf{M}^{-1}\mathbf{N}_3(\mathbf{q}, \mathbf{q})]\mathbf{q} = \tilde{\mathbf{f}}_0 \cos(\omega t), \quad (45)$$

where

$$\tilde{\mathbf{C}} = \mathbf{M}^{-1}\mathbf{C}, \quad \tilde{\mathbf{K}} = \mathbf{M}^{-1}\mathbf{K} \quad \text{and} \quad \tilde{\mathbf{f}}_0 = \mathbf{M}^{-1}\mathbf{f}_0. \quad (46a-c)$$

Eq. (45) is in the form suitable for numerical integration.

6. Numerical results

The dimensions and material properties of a simply supported, imperfection-free, laminated circular cylindrical shell made of graphite/epoxy plain weave fabric are: $L=520$ mm, $R=150$ mm, $h=1$ mm, $E_1=E_2=55 \times 10^9$ Pa, $G_{12}=4.48 \times 10^9$ Pa, $G_{13}=G_{23}=4.14 \times 10^9$ Pa, $\nu_{12}=\nu_{21}=0.04$ and $\rho=1500$ kg/m³; the fundamental mode ($m=1, n=4$) is considered, with natural frequency of 309.7 Hz; it has $n=4$ circumferential waves.

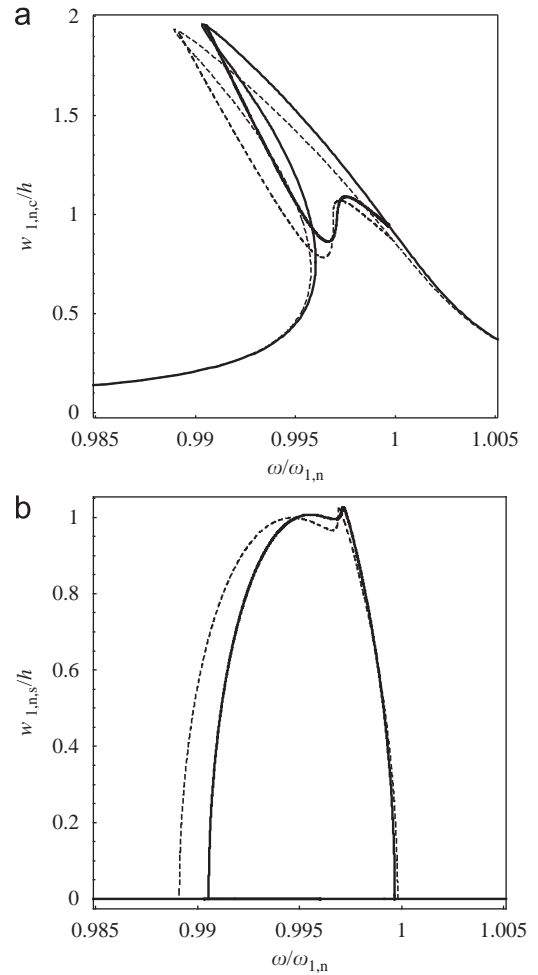


Fig. 4. Frequency-response curve of the circular cylindrical shell; $n=4$; —, solution keeping all the non-linear terms; - -, solution keeping only von Kármán non-linear terms. (a) Maximum of the generalized coordinate $w_{1,n,c}$ (driven mode) normalized with respect to the shell thickness h ; (b) maximum of the generalized coordinate $w_{1,n,s}$ (companion mode) normalized with respect to the shell thickness h .

Numerical results have been obtained by using the software AUTO [14] for continuation and bifurcation analysis by using the pseudo-arc length continuation method.

Fig. 3 presents the frequency–response curve for forced vibrations of the shell around the resonance of the fundamental mode computed by using the 3rd order shear deformation non-linear theory introduced in Section 2. A modal damping coefficient $\zeta_{1,n} = 0.001$ and harmonic point force excitation of magnitude 3.04 N and frequency ω applied at shell mid-length are assumed. The main branch 1 in Fig. 3 corresponds to vibration with zero amplitude of the companion mode $w_{1,n,s}(t)$. This branch has pitchfork bifurcations (BP) at $\omega/\omega_{1,n} = 0.9905$ and at 0.9997, where branch 2 appears. This new branch corresponds to participation of both $w_{1,n,c}(t)$ and $w_{1,n,s}(t)$, giving a traveling-wave response. The companion mode presents a node at the location of the excitation force and therefore it is not directly excited; its amplitude is different than zero only for large-amplitude vibrations, due to non-linear coupling. In the narrow frequency region where both $w_{1,n,c}(t)$ and $w_{1,n,s}(t)$ are different from zero, they give rise to a traveling wave around the shell; phase shift between the two coordinates is almost equal to $\pi/2$. The appearance of branch 2 is related to the 1:1 internal resonance of $w_{1,n,c}(t)$ and $w_{1,n,s}(t)$, which is due to the axial symmetry of the circular cylindrical shell. This branch appears for sufficiently large excitation, and it can be observed for vibration amplitude of the order of 1/10 of the shell thickness. Branch 2 undergoes two Neimark–Sacker (torus) bifurcations (TR), at $\omega/\omega_{1,n} = 0.9909$ and 0.9965. Amplitude-modulated (quasi-periodic) response is indicated in Fig. 3 on branch 2 for $0.9909 < \omega/\omega_{1,n} < 0.9965$, that is, bracketed by the two Neimark–Sacker bifurcations.

A comparison between the results obtained by using the present theory and those obtained keeping only von Kármán non-linear terms in Eqs. (B.1)–(B.3), or in Eqs. (17a–c), is shown in Fig. 4. Results indicate that, for the studied case, it is important to retain non-linearities in the in-plane displacements in order to obtain accurate results. In fact, the difference between the two calculations is significant from the quantitative point of view, even if the qualitative behavior is the same.

7. Conclusion

The proposed theory is the first 3rd order shear deformation theory retaining non-linearities in the in-plane displacements. Moreover, it has been derived with consistency without introducing non-linearities of the von Kármán at the end of the derivation of the theory. Numerical results show that, for the studied case, these non-linear terms are important to predict with accuracy the non-linear forced response of a laminated graphite/epoxy plain weave fabric circular cylindrical shell.

Acknowledgments

The first author acknowledges the financial support of the NSERC Discovery Grant, Canada Research Chair and Canada Foundation for Innovation programs of Canada and the McGill University start-up grant. The second author gratefully acknowledge the support of this research by the Oscar S. Wyatt Endowed Chair.

Appendix A. Stress–strain relations for a layer within a laminate

The coefficients in Eq. (19) for a lamina are given by

$$c_{11} = \frac{E_x}{1 - \nu_{xy}\nu_{yx}}, \quad c_{12} = c_{21} = \frac{E_y\nu_{xy}}{1 - \nu_{xy}\nu_{yx}}, \quad c_{22} = \frac{E_y}{1 - \nu_{xy}\nu_{yx}},$$

$$\nu_{ij}E_j = \nu_{ji}E_i. \quad (\text{A.1})$$

Usually, the lamina material axes (x, y) do not coincide with the plate reference axes (α_1, α_2), while the 3 axis is coincident with z . Then, the strains and stresses on material axes can be related to the reference axes by using the following invertible expressions [11,15]:

$$\begin{Bmatrix} \sigma_x \\ \sigma_y \\ \tau_{yz} \\ \tau_{xz} \\ \tau_{xy} \end{Bmatrix} = \mathbf{T}_1 \begin{Bmatrix} \sigma_1 \\ \sigma_2 \\ \tau_{23} \\ \tau_{13} \\ \tau_{12} \end{Bmatrix}, \quad \begin{Bmatrix} \varepsilon_{xx} \\ \varepsilon_{yy} \\ \gamma_{yz} \\ \gamma_{xz} \\ \gamma_{xy} \end{Bmatrix} = \mathbf{T}_2 \begin{Bmatrix} \varepsilon_{11} \\ \varepsilon_{22} \\ \gamma_{23} \\ \gamma_{13} \\ \gamma_{12} \end{Bmatrix}, \quad (\text{A.2a, b})$$

where

$$\mathbf{T}_1 = \begin{bmatrix} \cos^2 \theta & \sin^2 \theta & 0 & 0 & 2 \sin \theta \cos \theta \\ \sin^2 \theta & \cos^2 \theta & 0 & 0 & -2 \sin \theta \cos \theta \\ 0 & 0 & \cos \theta & -\sin \theta & 0 \\ 0 & 0 & \sin \theta & \cos \theta & 0 \\ -\sin \theta \cos \theta & \sin \theta \cos \theta & 0 & 0 & \cos^2 \theta - \sin^2 \theta \end{bmatrix}, \quad (\text{A.3})$$

$$\mathbf{T}_2 = \begin{bmatrix} \cos^2 \theta & \sin^2 \theta & 0 & 0 & \sin \theta \cos \theta \\ \sin^2 \theta & \cos^2 \theta & 0 & 0 & -\sin \theta \cos \theta \\ 0 & 0 & \cos \theta & -\sin \theta & 0 \\ 0 & 0 & \sin \theta & \cos \theta & 0 \\ -2 \sin \theta \cos \theta & 2 \sin \theta \cos \theta & 0 & 0 & \cos^2 \theta - \sin^2 \theta \end{bmatrix}, \quad (\text{A.4})$$

θ being the angle between the shell principal coordinate α_1 and the material axis x .

It can be shown that [11,16]

$$(\mathbf{T}_1^{-1}) = \mathbf{T}_2. \quad (\text{A.5})$$

Therefore, the matrix $[\mathbf{Q}]^{(k)}$ in Eqs. (20) and (21) is given by

$$[\mathbf{Q}]^{(k)} = [\mathbf{T}_1^{-1} \mathbf{C} (\mathbf{T}_1^{-1})^{(k)}], \quad (\text{A.6})$$

where \mathbf{C} is the 5×5 matrix of c_{ij} and G_{ij} coefficients in Eq. (19). As a consequence of the discontinuous variation of the stiffness matrix $[\mathbf{Q}]^{(k)}$ from layer to layer, the stresses may be discontinuous layer to layer. In particular,

$$Q_{11}^{(k)} = [c_{11} \cos^4 \theta + \sin^2 \theta (2c_{12} \cos^2 \theta + 4c_{66} \cos^2 \theta + c_{22} \sin^2 \theta)]^{(k)}, \quad (\text{A.7})$$

$$Q_{12}^{(k)} = [c_{12} (\sin^4 \theta + \cos^4 \theta) + \sin^2 \theta (c_{11} \cos^2 \theta + c_{22} \cos^2 \theta - 4c_{66} \cos^2 \theta)]^{(k)}. \quad (\text{A.8})$$

Appendix B. Strain–displacement relations for a circular cylindrical shell

$$\varepsilon_{x,0} = \frac{\partial u}{\partial x} + \frac{1}{2} \left[\left(\frac{\partial u}{\partial x} \right)^2 + \left(\frac{\partial v}{\partial x} \right)^2 + \left(\frac{\partial w}{\partial x} \right)^2 \right] + \frac{\partial w_0}{\partial x} \frac{\partial w}{\partial x}, \quad (\text{B.1})$$

$$\varepsilon_{\theta,0} = \frac{1}{R} \frac{\partial v}{\partial \theta} + \frac{w}{R} + \frac{1}{2} \left[\left(\frac{1}{R} \frac{\partial u}{\partial \theta} \right)^2 + \left(\frac{1}{R} \frac{\partial v}{\partial \theta} + \frac{w}{R} \right)^2 + \left(\frac{1}{R} \frac{\partial w}{\partial \theta} - \frac{v}{R} \right)^2 \right]$$

$$+ \frac{w_0}{R} \left(\frac{1}{R} \frac{\partial v}{\partial \theta} + \frac{w}{R} \right) + \frac{1}{R} \frac{\partial w_0}{\partial \theta} \left(\frac{1}{R} \frac{\partial w}{\partial \theta} - \frac{v}{R} \right), \quad (\text{B.2})$$

$$\gamma_{x\theta,0} = \frac{\partial v}{\partial x} + \frac{1}{R} \frac{\partial u}{\partial \theta} + \frac{1}{R} \frac{\partial u}{\partial x} \frac{\partial u}{\partial \theta} + \left(\frac{1}{R} \frac{\partial v}{\partial \theta} + \frac{w}{R} \right) \frac{\partial v}{\partial x} + \frac{\partial w}{\partial x} \left(\frac{1}{R} \frac{\partial w}{\partial \theta} - \frac{v}{R} \right)$$

$$+ \frac{w_0}{R} \frac{\partial v}{\partial x} + \frac{\partial w_0}{\partial x} \left(\frac{1}{R} \frac{\partial w}{\partial \theta} - \frac{v}{R} \right) + \frac{1}{R} \frac{\partial w_0}{\partial \theta} \frac{\partial w}{\partial x}, \quad (\text{B.3})$$

$$\gamma_{xz,0} = \phi_1 + \frac{\partial w}{\partial x}, \quad (\text{B.4})$$

$$\gamma_{0z,0} = \phi_2 + \frac{1}{R} \frac{\partial w}{\partial \theta}, \quad (\text{B.5})$$

$$k_x^{(0)} = \frac{\partial \phi_1}{\partial x}, \quad (\text{B.6})$$

$$k_x^{(1)} = 0, \quad (\text{B.7})$$

$$k_x^{(2)} = -\frac{4}{3h^2} \frac{\partial \phi_1}{\partial x} + \frac{\partial^2 w}{\partial x^2}, \quad (\text{B.8})$$

$$k_\theta^{(0)} = \frac{1}{R} \frac{\partial \phi_2}{\partial \theta} - \frac{w}{R^2}, \quad (\text{B.9})$$

$$k_\theta^{(1)} = -\frac{1}{R^2} \frac{1}{2} \frac{\partial \phi_2}{\partial \theta} + \frac{1}{R} \frac{\partial v}{\partial \theta} - \frac{1}{2R} \frac{\partial^2 w}{\partial \theta^2}, \quad (\text{B.10})$$

$$k_\theta^{(2)} = -\frac{4}{3h^2} \frac{1}{R} \frac{\partial \phi_2}{\partial \theta} + \frac{1}{R^2} \frac{\partial^2 w}{\partial \theta^2} - \frac{1}{3R^2} \frac{2}{R} \frac{\partial \phi_2}{\partial \theta} + \frac{1}{R^2} \frac{\partial^2 w}{\partial \theta^2}, \quad (\text{B.11})$$

$$k_{x\theta}^{(0)} = \frac{1}{R} \frac{\partial \phi_1}{\partial \theta} + \frac{\partial \phi_2}{\partial x} + \frac{1}{R} \left(\frac{\partial v}{\partial x} - \frac{1}{R} \frac{\partial u}{\partial \theta} \right), \quad (\text{B.12})$$

$$k_{x\theta}^{(1)} = \frac{1}{R} - \frac{1}{R} \frac{\partial \phi_1}{\partial \theta} + \frac{1}{2} \frac{\partial \phi_2}{\partial x} + \frac{1}{2R} \frac{\partial^2 w}{\partial x \partial \theta}, \quad (\text{B.13})$$

$$k_{x\theta}^{(2)} = -\frac{4}{3h^2} \frac{1}{R} \frac{\partial \phi_1}{\partial \theta} + \frac{\partial \phi_2}{\partial x} + \frac{2}{R} \frac{\partial^2 w}{\partial x \partial \theta} + \frac{1}{6R^2} - \frac{\partial \phi_2}{\partial x} + \frac{1}{R} \frac{\partial^2 w}{\partial x \partial \theta}, \quad (\text{B.14})$$

$$k_{xz}^{(0)} = 0, \quad (\text{B.15})$$

$$k_{xz}^{(1)} = -\frac{4}{h^2} \left(\phi_1 + \frac{\partial w}{\partial x} \right), \quad (\text{B.16})$$

$$k_{xz}^{(2)} = 0, \quad (\text{B.17})$$

$$k_{\theta z}^{(0)} = 0, \quad (\text{B.18})$$

$$k_{\theta z}^{(1)} = -\frac{4}{h^2} \left(\phi_2 + \frac{1}{R} \frac{\partial w}{\partial \theta} \right) + \frac{v}{R^3}, \quad (\text{B.19})$$

$$k_{\theta z}^{(2)} = 0. \quad (\text{B.20})$$

Appendix C. Kinetic energy for a circular cylindrical shell

In particular, for a laminate with the same density for all the layers and uniform thickness, the following simplified expression is obtained:

$$T_S = \frac{1}{2} \rho_s h \int_0^{2\pi} \int_0^L \left\{ \dot{u}^2 + \dot{v}^2 + \dot{w}^2 + h^2 \left[\frac{17}{315} (\dot{\phi}_1^2 + \dot{\phi}_2^2) + \left(\frac{2}{15R} \right) \dot{\phi}_1 \dot{u} + \left(\frac{41}{120R} \right) \dot{\phi}_2 \dot{v} + \frac{1}{4R^2} \dot{v}^2 + \frac{\partial \dot{w}}{\partial x} \left(\frac{1}{252} \frac{\partial \dot{w}}{\partial x} - \frac{1}{30R} \dot{u} - \frac{8}{315} \dot{\phi}_1 \right) + \frac{1}{R} \frac{\partial \dot{w}}{\partial \theta} \left(\frac{1}{252R} \frac{\partial \dot{w}}{\partial \theta} + \frac{\dot{v}}{120R} - \frac{8}{315} \dot{\phi}_2 \right) \right] \right\} R d\theta dx + O(h^5). \quad (\text{C.1})$$

If the ratio h/R is negligible (i.e. for thin shells), Eq. (C.1) can be simplified.

References

- [1] M. Amabili, S. Farhadi, Shear deformable versus classical theories for nonlinear vibrations of rectangular isotropic and laminated composite plates, *Journal of Sound and Vibration* 320 (2009) 649–667.
- [2] M. Amabili, *Nonlinear Vibrations and Stability of Shells and Plates*, Cambridge University Press, New York, USA, 2008.
- [3] L. Librescu, Refined geometrically nonlinear theories of anisotropic laminated shells, *Quarterly of Applied Mathematics* 45 (1987) 1–22.
- [4] J.N. Reddy, Exact solutions of moderately thick laminated shells, *Journal of Engineering Mechanics* 110 (1984) 794–809.
- [5] J.N. Reddy, C.F. Liu, A higher-order shear deformation theory of laminated elastic shells, *International Journal of Engineering Science* 23 (1985) 319–330.
- [6] R.A. Arciniega, J.N. Reddy, A consistent third-order shell theory with application to composite circular cylinders, *AIAA Journal* 43 (2005) 2024–2038.
- [7] J.N. Reddy, R.A. Arciniega, Shear deformation plate and shell theories: from Stavsky to present, *Mechanics of Advanced Materials and Structures* 11 (2004) 535–582.
- [8] J.N. Reddy, A general non-linear third-order theory of plates with moderate thickness, *International Journal of Non-Linear Mechanics* 25 (1990) 677–686.
- [9] S.T. Dennis, A.N. Palazotto, Large displacement and rotation formulation for laminated shells including parabolic transverse shear, *International Journal of Non-Linear Mechanics* 25 (1990) 67–85.
- [10] A.N. Palazotto, S.T. Dennis, *Nonlinear Analysis of Shell Structures*, AIAA Educational Series, Washington, DC, USA, 1992.
- [11] J.N. Reddy, *Mechanics of Laminated Composite Plates and Shells: Theory and Analysis*, second ed, CRC Press, Boca Raton, FL, USA, 2004.
- [12] M. Amabili, Comparison of shell theories for large-amplitude vibrations of circular cylindrical shells: Lagrangian approach, *Journal of Sound and Vibration* 264 (2003) 1091–1125.
- [13] V.V. Novozhilov, *Foundations of the Nonlinear Theory of Elasticity*, Graylock Press, Rochester, NY, USA, 1953 (now available from Dover, NY, USA).
- [14] E.J. Doedel, A.R. Champneys, T.F. Fairgrieve, Y.A. Kuznetsov, B. Sandstede, X. Wang, AUTO 97: Continuation and Bifurcation Software for Ordinary Differential Equations (with HomCont), Concordia University, Montreal, Canada, 1998.
- [15] J.E. Ashton, J.M. Whitney, *Theory of Laminated Plates*, Technomic, Stamford, CT, USA, 1970.
- [16] A.K. Nayak, S.S.J. Moy, R.A. Shenoi, Free vibration analysis of composite sandwich plates based on Reddy's higher-order theory, *Composites: Part B* 33 (2002) 505–519.