

DSMC Simulations of Apollo Capsule Aerodynamics for Hypersonic Rarefied Conditions

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Direct simulation Monte Carlo (DSMC) simulations are performed for the Apollo capsule in the hypersonic low-density transitional flow regime. The focus is on flow conditions similar to that experienced by the Apollo 6 Command Module during the high altitude portion of its reentry. Results for aerodynamic forces and moments are presented that demonstrate their sensitivity to rarefaction; that is, for free molecular to continuum conditions. Also, aerodynamic data are presented that shows their sensitivity to a range of reentry velocity, encompassing conditions that include reentry from low Earth orbit, lunar return, and Mars return velocities (7.7 to 15 km/s). The rarefied results are anchored in the continuum regime with data from Navier-Stokes simulations.

Nomenclature

A_{ref}	reference area, $\pi D_b^2/4$, m ²
cg	center of gravity, m
cp	center of pressure, m
C_A	axial force coefficient, Axial Force/ $((0.5\rho_\infty V_\infty^2)(A_{ref}))$
C_D	drag force coefficient, Drag/ $((0.5\rho_\infty V_\infty^2)(A_{ref}))$
C_L	lift force coefficient, Lift/ $((0.5\rho_\infty V_\infty^2)(A_{ref}))$
$C_{m,cg}$	pitching-moment coefficient, Moment about center of gravity/ $((0.5\rho_\infty V_\infty^2)(A_{ref})(D_b))$
$C_{m,0}$	pitching-moment coefficient, Moment about z/ $((0.5\rho_\infty V_\infty^2)(A_{ref})(D_b))$
C_N	normal force coefficient, Normal Force/ $((0.5\rho_\infty V_\infty^2)(A_{ref}))$
D_b	maximum body diameter, m
$Kn_{\infty,D,HS}$	free-stream hard sphere Knudsen number, λ_∞/D_b
L	lift force, N
L/D	lift to drag ratio
mcs	mean collision separation distance, m
mfp	mean free path, m
n	number density, m ⁻³
R_a	afterbody spherical nose radius, m
R_n	blunt forebody spherical nose radius, m
R_s	shoulder radius, m
T	temperature, K
V_∞	free-stream velocity, m/s
x,y,z	model coordinates, m
X	mole fractions
α	angle of incidence, deg

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γ	inertial entry flight-path angle, deg
λ_∞	mean free path in free stream, m
ρ	density, kg/m ³

A. Subscripts

W	wall
∞	free stream

I. Introduction

The space capsule,¹ eclipsed for decades by the more complex and costly shuttle, now appears likely to emerge as its successor. Realization of such a change is essentially assured with China's recent manned spaceflight successes and NASA's announced vision of a new space craft for human space exploration, the Crew Exploration Vehicle (CEV). China conducted a second successful manned spaceflight in October 2005 with the Shenzhou 6, a configuration that is an adaption of Russia's veteran Soyuz design. On September 19, 2005, NASA announced the findings of the Exploration Systems Architecture Study that recommended the use an Apollo-like capsule for the CEV design. The CEV is much larger than Apollo, almost twice the mass of the Apollo Command Module along with a much larger volume, and like Apollo, would be attached to a service module for life support and propulsion. Mission applications of the CEV include that of a low-Earth-orbit (LEO) version with a crew of six to the International Space Station, a lunar version that would carry a crew of four, and a Mars version that would carry a crew of six.

With commitments to evolve the CEV design(s) for LEO, lunar, and Mars missions, aerothermodynamic data bases will be generated utilizing computational and experimental (both ground-based and flight) resources. These new data bases along with an extensive capsule heritage, particularly that from Apollo (Refs. 2 to 6, for example), will provide the basis for optimizing the CEV's design, with particular emphasis on safety, flexibility, and affordability. The current study focuses on the aerodynamics of the Apollo Command Module during the transitional portion of its reentry, from free molecular to near continuum continuum conditions. The primary focus is on flow conditions similar to those experienced by the Apollo 6 flight test, with a reentry velocity of 9.6 km/s. Numerical simulations for the transitional flow regime are made with the 3D DSMC code of Bird,⁷⁻⁹ called DS3V, and for the continuum regime with the 3D Navier-Stokes (NS) code of Gnoffo,¹⁰⁻¹² called LAURA (Langley Aerothermodynamic Upwind Relaxation Algorithm). Results are presented that show the sensitivity of the capsule aerodynamics to rarefaction, velocity variations at an altitude of 105 km, sensitivity to grid resolution, and chemistry model assumptions (number of species) at 85 km with the NS simulations. The DSMC results presented herein along with data from some recent studies^{14,15} demonstrate available capability to address the transitional flow aerodynamics of capsules such as the CEV, a capability that did not exist when the Apollo Command Module design was evolved. The current results show that the lift and lift-to-drag coefficients increase substantially with decreasing rarefaction. Also, the location of the longitudinal center of pressure is very sensitive to the degree of rarefaction and the simulations show that the stable trim point for the Apollo capsule at 105 km altitude occurs at an incidence angle of -164 degrees rather than the nominal -25 degrees flown by Apollo 6; that is, the capsule is statically unstable for much, if not all, of the transitional flow regime, a result not that uncommon for capsules as discussed by Wilmoth et al.¹⁶ for the Stardust sample return capsule and Moss et al.¹⁷ for the Mars Pathfinder capsule. Results of the simulations for variations in free-stream velocity show that the changes in the aerodynamic coefficients with increasing velocity are similar to those incurred with increasing rarefaction; consistent with the correlations demonstrated by Wilhite et al.¹⁸ (Fig. 7, p 172) for the Shuttle Orbiter axial-force coefficients as a function of a viscous correlation parameter.

II. Numerical Programs and Model Parameters

A. DSMC Analyses

The DSMC program used in the current study is the DS3V program of Bird,⁷⁻⁹ a general 3D code that provides both time accurate unsteady flow and time-averaged steady flow simulations. A scalar version

of this program was used in this study where all the simulations were made by using a 3.2 GHz personal computer with a memory of 2.0 GB. Molecular collisions are simulated with the variable hard sphere (VHS) molecular model. The Larsen-Borgnakke statistical model¹³ controls the energy exchange between kinetic and internal modes. For the present study, the simulations are performed by using a five-species reacting air gas model while considering energy exchange between translational, rotational, and vibrational modes. The molecular gas constants used in the current study are those given in Ref. 9. Also, a rotational relaxation collision number of 5 and a temperature dependent vibrational collision number (Eq. 6.53 of Ref. 9) were used. More details regarding the DS3V code can be found in Ref. 7, and examples of recent validation studies are presented in Ref. 8.

For all simulations, the surface is assumed to be noncatalytic and at a specified wall temperature. As for gas-surface interactions, they are assumed to be diffuse, with full energy accommodation.

The geometric size of the computational domain was varied with the degree of rarefaction of the free-stream flow, since the influence of the body on the external flow at high Knudsen numbers extends outward a greater distance than is the case for a denser flow. The total number of cells in the computational domain was also a variable. The grid adaptation used in the current study nominally used 20 simulated molecules per cell. The total number of molecules used in the simulations ranged from approximately 1 to 16 million.

An indicator of the resolution achieved in a given simulation is given by the ratio of the mean collision separation between collision partners to the local mean free path (mcs/mfp). For blunt body flows, as considered herein, the average value for this parameter over the computational domain should be much less than 1 to ensure that the values of mcs/mfp are less than 1 adjacent to the surface. If these guidelines are not met, the calculated results will be inaccurate. Results are presented where the failure to meet this criterion and the resulting impact on the calculated forces and moments are demonstrated.

B. Navier-Stokes Analyses

Navier-Stokes analyses are performed by using the LAURA computational fluid dynamics code.¹⁰⁻¹² LAURA is an upwind-bias, point-implicit/line-implicit relaxation algorithm for obtaining the numerical solution to the Navier-Stokes equations for three-dimensional, viscous, hypersonic flows in thermochemical nonequilibrium. LAURA has both the thin layer and full NS options, and both options were exercised in the current study. All of the LAURA simulations assumed the flow to be a reacting gas mixture with the surface boundary conditions consisting of a constant wall temperature, a noncatalytic surface, and no slip or temperature jump. The volume grid consists of 24 blocks with a total of 1 966 000 cells, and in the direction normal to the wall, there are 80 cells, which cover the region from the wall to the outer boundary. Grid adaption assured cell Reynolds number adjacent to the wall at a nominal value no greater than 0.5 for the highest altitude (95 km) and 5.0 for the lowest altitude (65 km). The structured surface grid consisted of 24 576 cells. To balance the computational load, calculations were performed on 12 dual processor 2.8 GHz Opteron workstations with one block assigned to each of the 24 processors. Solutions were considered converged when the surface properties became steady and changed little after additional integration cycles.

C. Free Molecular and Newtonian Analyses

The free molecular (FM) and modified Newtonian (MN) results were obtained with the DACFREE code of R. G. Wilmoth (private communication, July 2005). DACFREE computes aerodynamic forces and moments on arbitrary bodies using standard free molecular and modified Newtonian methods. This code can handle arbitrary geometries specified as an unstructured collection of triangles, and for the present study, the surface grid was the same as that used in the DSMC simulations.

III. Conditions and Results

A. Conditions

Considerable resources were devoted to quantifying the impact of the aerothermodynamic environment on the Apollo Command Module during reentry, particularly the thermal protection system. Table 1, based on the data presented in Ref. 2, list some of the reentry parameters for the 4 unmanned Apollo heat-shield-qualification flight tests, two at orbital entry velocities and two at superorbital entry velocities (Apollo 6 entered at 1.128 km/s less than was planned due to a re-ignition failure in the upper stage³). The current

study focuses on an altitude range of 200 to 65 km at 9.6 km/s (corresponding to the Apollo 6 reentry condition) at an angle of incidence of -25 degrees, for a range of incidence angles at an altitude of 105 km, and for a range of reentry velocities (7.68 to 15 km/s) at -25 degrees incidence and 105 km altitude.

The axisymmetric geometry for the Apollo Command Module used in the present study is shown in Fig. 1, which does not account for thermal protection thickness variations between the leeward and windward sides. The Apollo capsule was flown at an angle of incidence while using an offset center of gravity (location used in the current study is listed in Fig. 1). The Apollo capsule has a truncated spherical section, followed by a toroidal section, and then a conical section. As discussed by Bertin¹⁹ (page 291), the sonic point, which occurs near the tangency point of the spherical heat shield and the toroidal surface, are inboard of the locations that they would occupy for a full spherical cap. As a result, the entire flowfield in the subsonic portion of the shock layer is modified with respect to those for a full spherical cap.

When the pressures and shear stresses are integrated over the surface, the resultant force acts at the center-of-pressure (cp) of the capsule. The total force vector is usually resolved into components, as shown in Fig. 2. Nomenclature used for the body (axial and normal) and velocity (drag and lift) oriented coordinates are as shown in Fig. 2.

For the DSMC simulations, an unstructured surface grid (Figs. 3 and 4) is used to define the body surface, where the number of surface points and triangles were 1912 and 3718, respectively. This surface resolution was deemed adequate after a calculation was made for a case (105 km altitude and -25 degrees incidence) with a much finer surface grid resolution (5946 and 11,178 points and triangles, respectively) and with a negligible change in results. Note that the numerical simulations take advantage of the problem symmetry in that the flow is computed about only half of the capsule.

Free-stream atmospheric conditions are listed in Tables 2 and 3 and are based on the data of Jacchia²⁰ (an exospheric temperature of 1200 K) for altitudes of 90 km and above and on that of Ref. 21 for altitudes less than 90 km. The surface temperatures are assumed to be uniformly distributed at the values listed in Tables 4, 6, and 8, and are calculated based on the minimum value resulting from either the free-molecular radiative equilibrium heat transfer to the stagnation point or the radiative equilibrium temperature based on the stagnation point heating from the correlation of Sutton (Eq. 1 of Ref. 22). The free-stream Knudsen numbers listed in Table 3 are based on the free-stream number density, a characteristic length of 3.912 m (maximum capsule diameter), and a constant molecular diameter of 3.78×10^{-10} m.

B. Rarefaction and Grid Resolution Effects

Results of the numerical simulations are presented in Tables 4 through 8 and Figs. 5 through 11. Table 4 and Figs. 5 through 7 present results that show the effects of rarefaction, as expressed by the Knudsen number, on the aerodynamic and moment coefficients and the location of the center of pressure. The lift (C_L) and lift-to-drag (L/D) coefficients are shown to be extremely sensitive to rarefaction, increasing in value with decreasing rarefaction. All results presented in Fig. 5 were obtained with the DS3V code, and as noted in this figure these results have been generated with a simulation merit parameter that is too large for the lower altitude cases: that is, the value of mcs/mfp is of order one or larger. Table 7 provides data that is useful in assessing the goodness of the simulation for 5 different altitudes between 105 and 85 km. These grid sensitivity studies were made by sequentially increasing the number of simulated molecules and adapting the grid to 20 molecules per cell. For the 105 and 100 km conditions, we see negligible to small changes in the coefficients as the resolution is refined; that is, by increasing the number of simulated molecules and computational cells with a corresponding reduction in the magnitude of the solution merit parameter mcs/mfp . However, for the 95 to 85 km conditions, it is not possible with a single-processor personal computer to achieve an adequate grid resolution. To determine the impact of the lack of grid resolved DS3V simulations for the lower altitude conditions, Navier-Stokes solutions were generated for these conditions and lower altitudes, as presented in Table 8. When a comparison of the DS3V and LAURA results at 85 km (Tables 4 and 8) is made, one is able to see the quantitative impact of a poorly resolved simulation and its impact of the predicted aerodynamics. For example, the ratio of DS3V to LAURA results at 85 km for drag, lift, and L/D coefficients are 1.06, 0.89, and 0.84, respectively, in which the DS3V simulation was made with a global mean mcs/mfp of 2.45. Recall that a grid resolved DS3V simulation requires a mcs/mfp value that is of order 0.1, clearly demonstrated in the results shown in Table 7.

Aerodynamic data presented in Figs. 6 and 7 includes both the NS and the grid resolved DSMC results, and provides coverage in terms of hard sphere free-stream Knudsen numbers of approximately six orders of magnitude. Even though an overlap with the two simulation methods has not been demonstrated, the results

clearly show that two very different numerical methods are producing similar and reasonably consistent results (joined by dash lines) in the 95 to 100 km altitude range (Kn number of approximately 0.024). As detailed in Table 8, the NS results includes two modeling assumptions accounting for different gas models (a 5-species model that does not account for ionization and 7- and 11-species models that account for ionized species) and the actual equations solved (full NS [FNS] or thin layer NS [TLNS]). Results for the 85 km flow conditions show little sensitivity of the aerodynamics to the effects of either TLNS versus FNS or whether the effects of ionization are included. Results presented in Ref. 23 indicated that as the flow becomes more rarefied, the full NS provides better agreement with the DSMC results and the agreement persists to more rarefied conditions.

Figure 7 details the movement of the center of pressure and the corresponding moment coefficient about the center of gravity as a function of Knudsen number. As the capsule descends from 200 to 65 km, the center of pressure experiences a substantial translation as it moves from a position forward of the center of gravity to one well aft. The corresponding change in the moment coefficient (Fig. 7) is from a negative value to a small positive value.

C. Effects of Angle of Incidence

Figures 8 through 10 and Table 5 present data that show the dependence of the Apollo capsule aerodynamics to variations in angle of incidence for the 105 km altitude conditions and 9.6 km/s. Figure 8 highlights the dependence of L/D on incidence angle and also demonstrates its sensitivity to rarefaction by including the free molecular (FM) and modified Newtonian (MN) results. The FM and MN results were generated with the DACFREE code at the 200 km and 85 km conditions, respectively.

Results of the DSMC simulations for the force coefficients are presented in Fig. 9 as a function of incidence. Results for the center of pressure location and the moment coefficient about the offset center of gravity are presented in Fig. 10. These simulations show that the stable trim point for the Apollo capsule at 105 km altitude occurs at an incidence angle of -164 degrees rather than the nominal -25 degrees flown by Apollo 6; that is, the capsule is statically unstable for much, if not all, of the transitional flow regime, a result not that uncommon^{16,17} for capsules in the transitional rarefied regime.

D. Effects of Free-Stream Velocity

To examine the effects of free-stream velocity variations, simulations were made for the Apollo capsule at an altitude of 105 km and -25 degrees incidence for 5 free-stream velocities ranging from 7.7 to 15 km/s (Table 6). Four of the velocities correspond to the nominal re-entry conditions of the 4 unmanned Apollo qualification flight tests (Table 1). The 15 km/s velocity is representative of the upper bounds for a Mars return mission. Consequently, this range of entry velocities is inclusive of that for reentry from LEO, lunar return, and Mars return missions. Results of the simulations for variations in free-stream velocity show (Table 6 and Fig. 11) that the changes in the aerodynamic coefficients with increasing velocity are similar to those incurred with increasing rarefaction; that is, the magnitude of the drag, axial, and normal force coefficients increases with increasing free-stream velocity while the magnitude of the lift and lift-to-drag ratio coefficients decrease with increasing velocity. These findings are consistent with the correlations demonstrated by Wilhite et al.¹⁸ (Fig. 7, p 172) for the Shuttle Orbiter axial-force coefficients as a function of a viscous correlation parameter.

IV. Concluding Remarks

A computational study of hypersonic flow over the Apollo Command Module is made by using the direct simulation Monte Carlo (DSMC) method. The computations are made for Earth entry conditions, similar to that experienced by Apollo 6, by using a 5-species reacting air model. Simulations are made for altitudes of 200 to 85 km, free-stream velocities of 7.7 to 15 km/s, and various angles of incidence. Results of the simulations show the effect of both rarefaction and entry velocities on the aerodynamic forces and moments. Also, results are presented that show the sensitivity of solutions to grid resolution and the approximate bounds of reliable results when using the DSMC code called DS3V.

The rarefied results are anchored in the continuum regime with simulations made with a Navier-Stokes code for altitudes of 95 to 65 km and a free-stream velocity of 9.6 km/s. Included in the Navier-Stokes simulations were sensitivity studies regarding the use of full Navier Stokes or thin layer Navier Stokes and the impact of including or not including the effects of ionization on the calculated aerodynamics.

Significant findings of the present investigation are as follows: (1) the lift and lift-to-drag coefficients increase substantially with decreasing rarefaction, (2) the location of the longitudinal center of pressure is very sensitive to the degree of rarefaction, (3) the Apollo Command Module is statically unstable for much of the rarefied flow regime, (3) changes in the aerodynamic coefficients with increasing velocity have the same trend as that for increasing rarefaction, (4) the present DSMC simulations are shown to be reliable based on grid resolution studies for altitudes from free molecular to approximately 100 km altitude, (5) even though an overlap of grid converged DSMC and NS simulations were not realized in the current study, the two simulation techniques were sufficiently close in altitude space to indicate that the two simulation methods provide consistent results as they approach each other in the 95 to 100 km altitude interval, and (6) that the NS results for aerodynamics demonstrate a very small sensitivity to the 3 gas model used (5, 7, and 11 species models) for the 85 km altitude conditions.

Acknowledgments

The author acknowledges the assistance of the following individuals: Victor Lessard of Genex Systems for providing the unstructured surface grid and Richard Wheless of NCI Information Systems for assisting with the graphics.

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Table 1. Reentry conditions for the Apollo Command Module flight tests.

Flight Designation	V_∞ , m/s	α , deg	γ , deg	Max Decel, g's	Theoretical Max Heating, W/cm ²
AS-201	7.67	-20	-8.6	14.3	186
AS-202	8.29	-18	-3.5	2.4	91
Apollo 4	10.73	-25	-6.9	7.3	488
Apollo 6	9.6	-25	-5.9	4.6	237

Table 2. Free-stream conditions.

Altitude, km	n_∞ , m ⁻³	ρ_∞ , kg/m ³	T_∞ , K	Molecular weight
200	8.9996×10^{15}	3.2829×10^{-10}	1026	21.970
170	2.2702×10^{16}	8.7777×10^{-10}	892	23.290
150	5.3055×10^{16}	2.1383×10^{-9}	733	24.273
140	9.3528×10^{16}	3.8548×10^{-9}	625	24.823
135	1.3149×10^{17}	5.4862×10^{-9}	564	25.127
130	1.9429×10^{17}	8.2075×10^{-9}	500	25.441
125	3.0598×10^{17}	1.3100×10^{-8}	433	25.783
120	5.2128×10^{17}	2.2642×10^{-8}	368	26.159
115	9.8562×10^{17}	4.3575×10^{-8}	304	26.626
110	2.1246×10^{18}	9.6068×10^{-8}	247	27.232
105	5.0947×10^{18}	2.3640×10^{-7}	208	27.943
100	1.1898×10^{19}	5.5824×10^{-7}	194	28.258
95	3.1167×10^{19}	1.4835×10^{-6}	189	28.613
90	7.0755×10^{19}	3.3848×10^{-6}	188	28.810
85	1.6540×10^{20}	7.9550×10^{-6}	181	28.960
75	9.0130×10^{20}	4.3350×10^{-5}	200	28.960
65	3.4651×10^{21}	1.6665×10^{-4}	293	28.960

Table 3. Atmospheric composition and Knudsen numbers for reentry conditions.

Altitude, km	X_{O_2}	X_{N_2}	X_O	$Kn_{\infty,D,HS}$
200	0.03146	0.45476	0.51378	44.74
170	0.04354	0.54820	0.40826	17.74
150	0.05461	0.61557	0.32982	7.59
140	0.06181	0.65173	0.28646	4.31
135	0.06593	0.67158	0.26248	3.06
130	0.07089	0.69113	0.23799	2.07
125	0.07679	0.71171	0.21150	1.32
120	0.08451	0.73271	0.18278	0.773
115	0.09779	0.75386	0.14835	0.408
110	0.12323	0.77042	0.10635	0.190
105	0.15808	0.78319	0.05873	0.081
100	0.17683	0.78440	0.03877	0.0338
95	0.20040	0.78687	0.01273	0.0139
90	0.20905	0.78748	0.00347	0.0057
85	0.23720	0.76280	0.00000	0.0024
75	0.23720	0.76280	0.00000	0.00045
65	0.23720	0.76280	0.00000	0.00012

Table 4. Effect of rarefaction on aerodynamics for -25° incidence and a free-stream velocity of 9.6 km/s.

Alt., km	T_W , K	C_A	C_N	$C_{m,0}$	C_D	C_L	L/D
200	234	1.731	-0.777	0.113	1.898	0.027	0.014
170	300	1.734	-0.775	0.112	1.899	0.030	0.016
150	373	1.723	-0.757	0.113	1.881	0.042	0.022
140	434	1.702	-0.728	0.112	1.851	0.060	0.032
135	474	1.698	-0.706	0.112	1.838	0.078	0.042
130	524	1.679	-0.685	0.111	1.812	0.089	0.049
125	589	1.664	-0.655	0.110	1.785	0.110	0.062
120	675	1.658	-0.620	0.110	1.764	0.139	0.079
115	795	1.644	-0.566	0.110	1.729	0.181	0.105
110	920	1.604	-0.477	0.112	1.655	0.245	0.148
105	1029	1.529	-0.374	0.113	1.544	0.307	0.199
100	1146	1.448	-0.279	0.109	1.431	0.359	0.251
95	1295	1.380	-0.226	0.104	1.346	0.379	0.281
90	1436	1.354	-0.211	0.100	1.316	0.381	0.290
85	1598	1.351	-0.205	0.099	1.311	0.385	0.293

Table 5. Effect of incidence angle on aerodynamics for a free-stream velocity of 9.6 km/s, an altitude of 105 km, and a wall temperature of 1029 K.

α , deg	C_A	C_N	$C_{m,0}$	C_D	C_L	L/D	$C_{m,cg}$	$C_{m,cg}^*$
0	1.775	0.000	0.000	1.775	0.000	0.000	-0.0726	-0.0726
2	1.773	0.031	-0.010	1.773	0.031	0.017	-0.0732	-0.0719
5	1.764	0.078	-0.024	1.764	-0.077	-0.043	-0.0735	-0.0708
10	1.734	0.154	-0.048	1.734	-0.149	-0.086	-0.0734	-0.0684
15	1.683	0.229	-0.071	1.685	-0.214	-0.127	-0.0723	-0.0654
20	1.615	0.304	-0.093	1.621	-0.267	-0.165	-0.0697	-0.0624
25	1.529	0.374	-0.113	1.544	-0.307	-0.199	-0.0659	-0.0592
30	1.428	0.445	-0.132	1.460	-0.328	-0.225	-0.0601	-0.0568
35	1.317	0.506	-0.152	1.370	-0.341	-0.249	-0.0571	-0.0507
40	1.199	0.571	-0.171	1.285	-0.334	-0.260	-0.0534	-0.0447
45	1.077	0.639	-0.194	1.213	-0.309	-0.255	-0.0534	-0.0374
60	0.678	0.842	-0.258	1.069	-0.166	-0.156	-0.0390	-0.0165
75	0.253	0.986	-0.295	1.018	0.011	0.011	-0.0166	-0.0041
90	-0.143	1.047	-0.298	1.047	0.143	0.136	0.0147	-0.0030
95	-0.258	1.052	-0.292	1.071	0.166	0.155	0.0266	-0.0054
105	-0.460	1.051	-0.275	1.134	0.172	0.152	0.0514	-0.0137
120	-0.708	1.018	-0.241	1.235	0.104	0.084	0.0861	-0.0282
135	-0.915	0.918	-0.192	1.296	-0.002	-0.001	0.1143	-0.0394
150	-1.129	0.736	-0.133	1.346	-0.073	-0.054	0.1285	-0.0362
155	-1.207	0.649	-0.114	1.368	-0.079	-0.057	0.1253	-0.0266
160	-1.279	0.545	-0.094	1.388	-0.074	-0.054	0.1180	-0.0134
165	-1.341	0.424	-0.072	1.406	-0.063	-0.045	0.1074	0.0023
170	-1.389	0.290	-0.049	1.419	-0.045	-0.032	0.0930	0.0206
175	-1.418	0.147	-0.025	1.425	-0.023	-0.016	0.0766	0.0394
178	-1.427	0.059	-0.010	1.428	-0.010	-0.007	0.0661	0.0506
180	-1.429	0.000	-0.000	1.429	-0.000	-0.000	0.0584	0.0584

* The moment coefficient for the corresponding negative angle of incidence.

Table 6. Effect of velocity on aerodynamics for a -25° incidence angle and an altitude of 105 km.

V_∞ , km/s	T_W , K	C_A	C_N	$C_{m,0}$	C_D	C_L	L/D
7680	871	1.515	-0.361	0.113	1.526	0.313	0.205
8290	922	1.520	-0.365	0.113	1.532	0.312	0.204
9600	1029	1.529	-0.374	0.113	1.544	0.307	0.199
10759	1121	1.535	-0.383	0.113	1.553	0.302	0.194
15000	1439	1.552	-0.410	0.113	1.579	0.284	0.180

Table 7. Sensitivity of aerodynamic forces and moments to simulation merit parameter (mcs/mfp) for a -25° incidence angle and $V_\infty = 9.6$ km/s.

Alt., km	Mean mcs/mfp	Simulated Molecules	C_A	C_N	$C_{m,0}$	C_D	C_L	L/D
105	0.130	0.951×10^6	1.529	-0.374	0.113	1.544	0.307	0.199
	0.072	3.792×10^6	1.525	-0.366	0.112	1.537	0.312	0.203
	0.048	9.475×10^6	1.522	-0.363	0.112	1.533	0.315	0.205
100	0.278	1.232×10^6	1.453	-0.297	0.110	1.442	0.345	0.239
	0.149	4.877×10^6	1.448	-0.284	0.110	1.432	0.354	0.247
	0.103	14.536×10^6	1.448	-0.279	0.109	1.431	0.359	0.251
95	0.835	1.372×10^6	1.404	-0.263	0.107	1.384	0.355	0.256
	0.462	5.429×10^6	1.388	-0.240	0.105	1.359	0.369	0.272
	0.338	10.804×10^6	1.383	-0.230	0.104	1.351	0.376	0.278
	0.299	16.114×10^6	1.380	-0.226	0.104	1.346	0.379	0.281
90	1.610	2.468×10^6	1.381	-0.239	0.103	1.353	0.367	0.272
	0.930	9.833×10^6	1.354	-0.211	0.100	1.316	0.381	0.290
85	3.980	2.914×10^6	1.386	-0.254	0.103	1.360	0.364	0.268
	2.450	11.704×10^6	1.351	-0.205	0.099	1.311	0.385	0.293

Table 8. Aerodynamics obtained with the LAURA Navier-Stokes code for -25° incidence angle and 9.6 km/s (results for both full Navier Stokes (FNS) and thin layer Navier Stokes (TLNS)).

Alt., km	Models	T_W	C_A	C_N	$C_{m,0}$	C_D	C_L	L/D
95	FNS, 11 species	1285	1.369	-0.197	0.105	1.324	0.400	0.302
90	TLNS, 11 species	1436	1.337	-0.157	0.098	1.277	0.423	0.331
85	FNS, 11 species	1598	1.302	-0.130	0.092	1.235	0.432	0.350
85	TLNS, 11 species	1598	1.302	-0.129	0.092	1.237	0.434	0.351
85	TLNS, 7 species	1598	1.306	-0.129	0.091	1.233	0.433	0.351
85	TLNS, 5 species	1598	1.298	-0.128	0.091	1.232	0.433	0.351
75	TLNS, 5 species	1975	1.292	-0.110	0.090	1.218	0.446	0.366
65	TLNS, 5 species	2337	1.295	-0.104	0.088	1.217	0.453	0.372

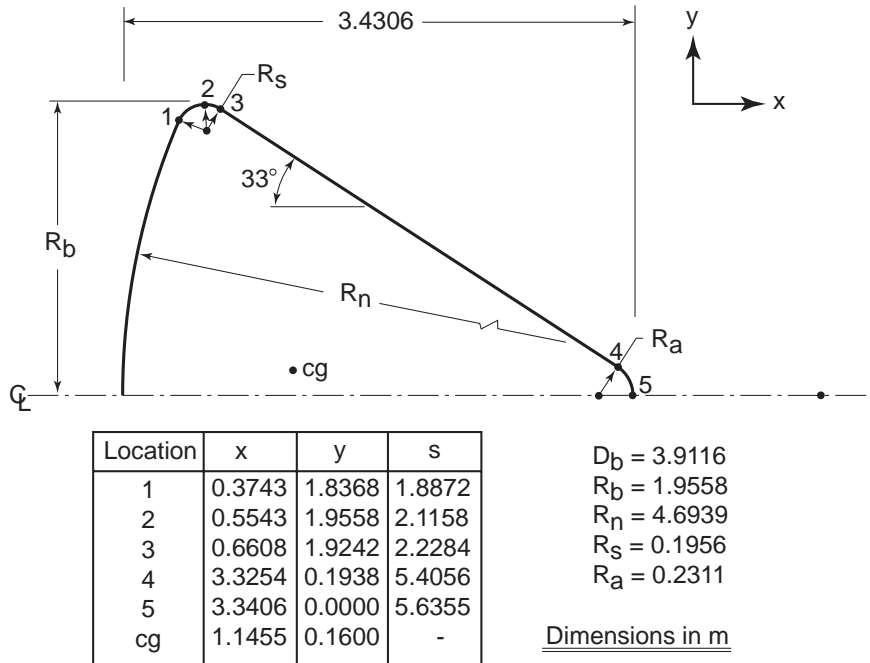


Figure 1. Outer mold line of the Apollo Command Module used in the present work.

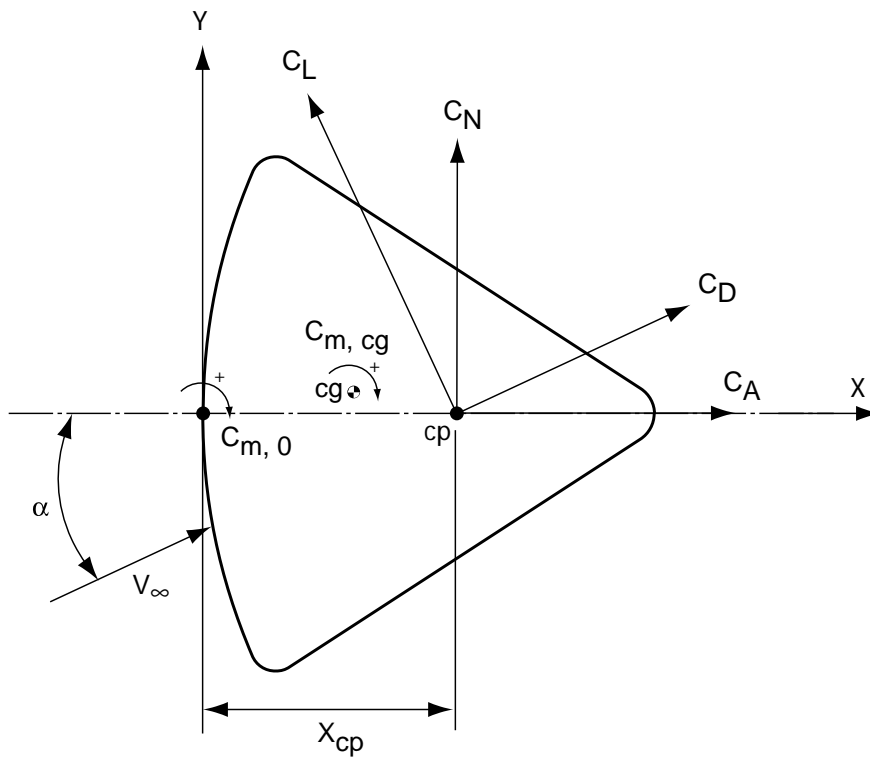


Figure 2. Nomenclature for aerodynamic forces in the pitch plane.

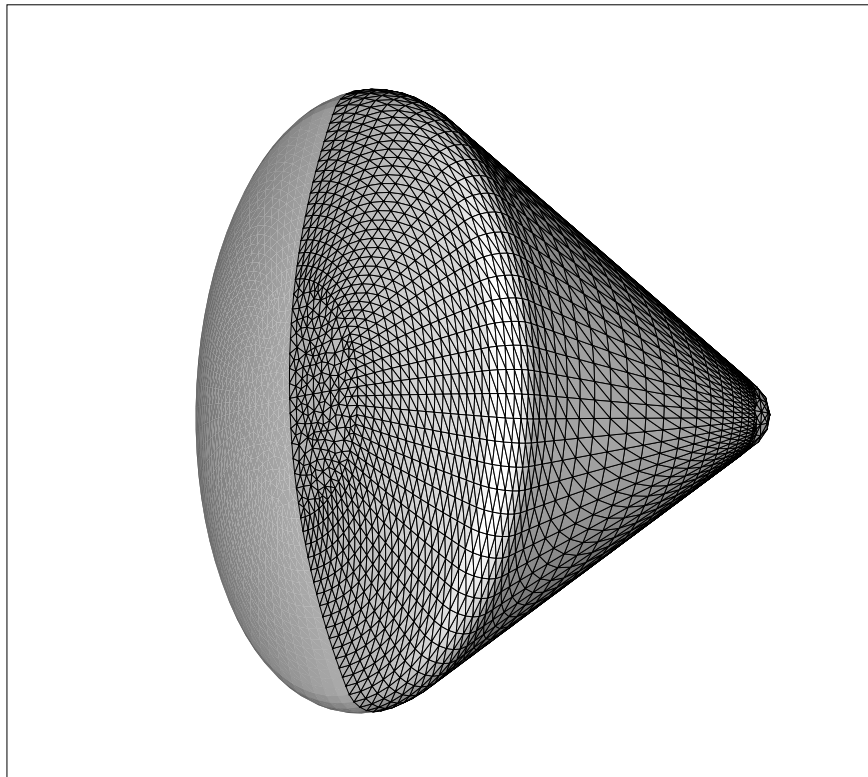


Figure 3. Frontal view of Apollo Command Module unstructured body grid used in present DSMC simulations.

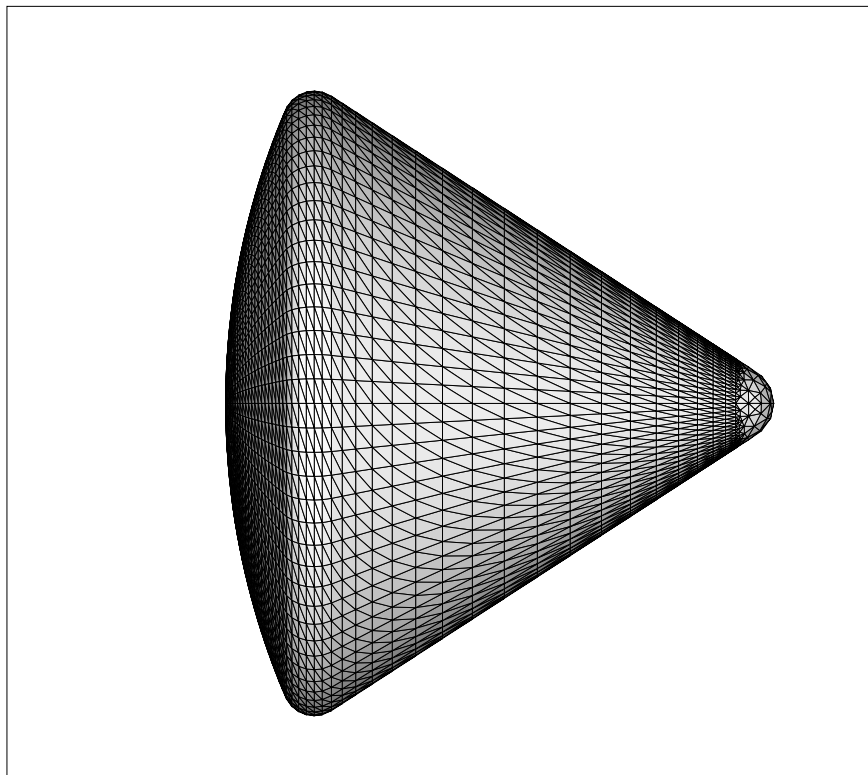


Figure 4. Side view of Apollo Command Module unstructured body grid used in present DSMC simulations.

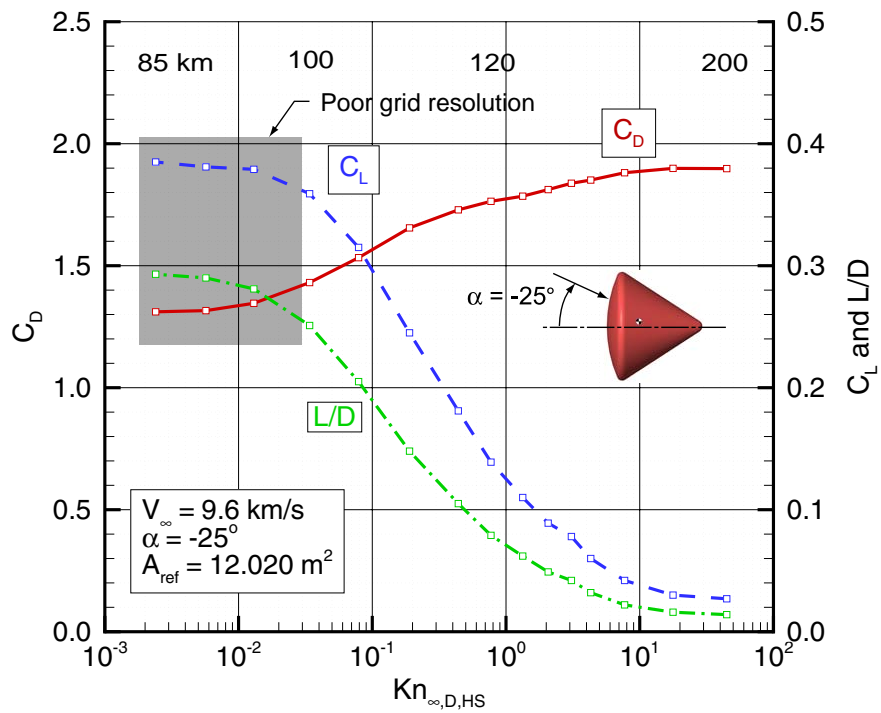
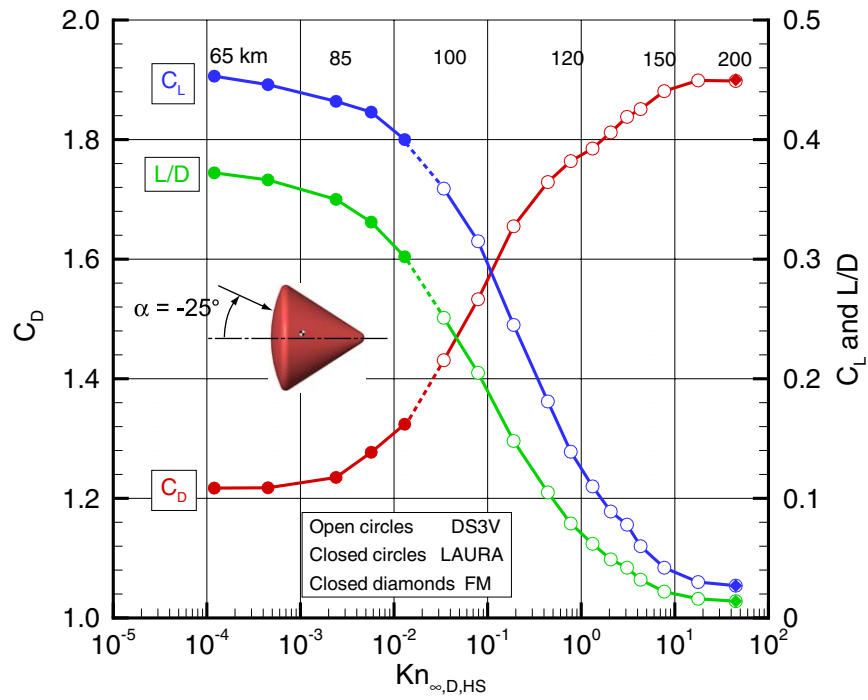
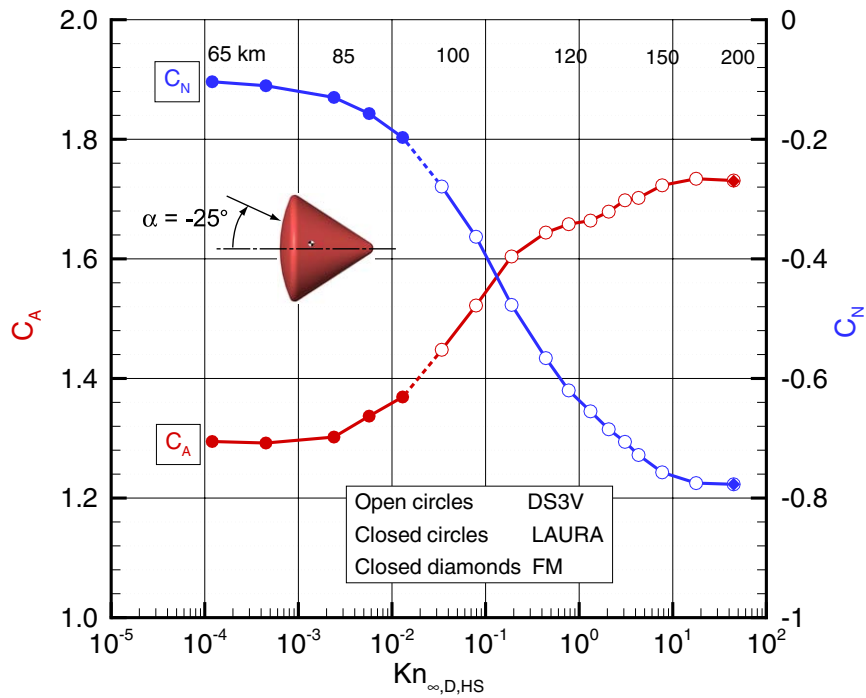


Figure 5. DS3V results for Apollo aerodynamics as a function of rarefaction, including poorly grid resolved results for lower altitudes.



(a) Drag and lift coefficients.



(b) Axial and normal coefficients.

Figure 6. Apollo aerodynamics as a function of rarefaction for an incidence angle of -25 degrees.

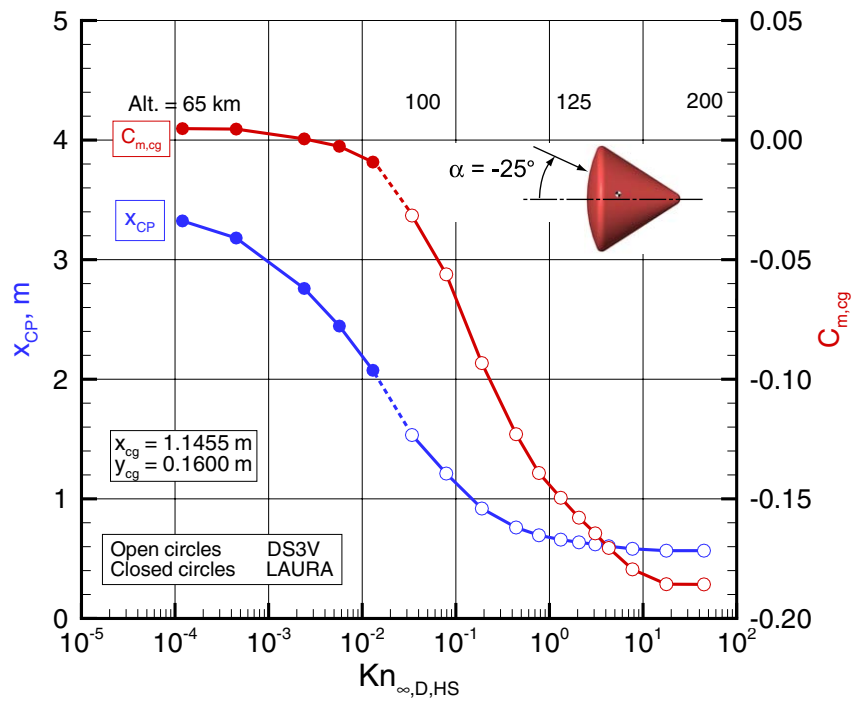


Figure 7. Sensitivity of center of pressure and moment coefficient to rarefaction for -25 degrees angle of incidence.

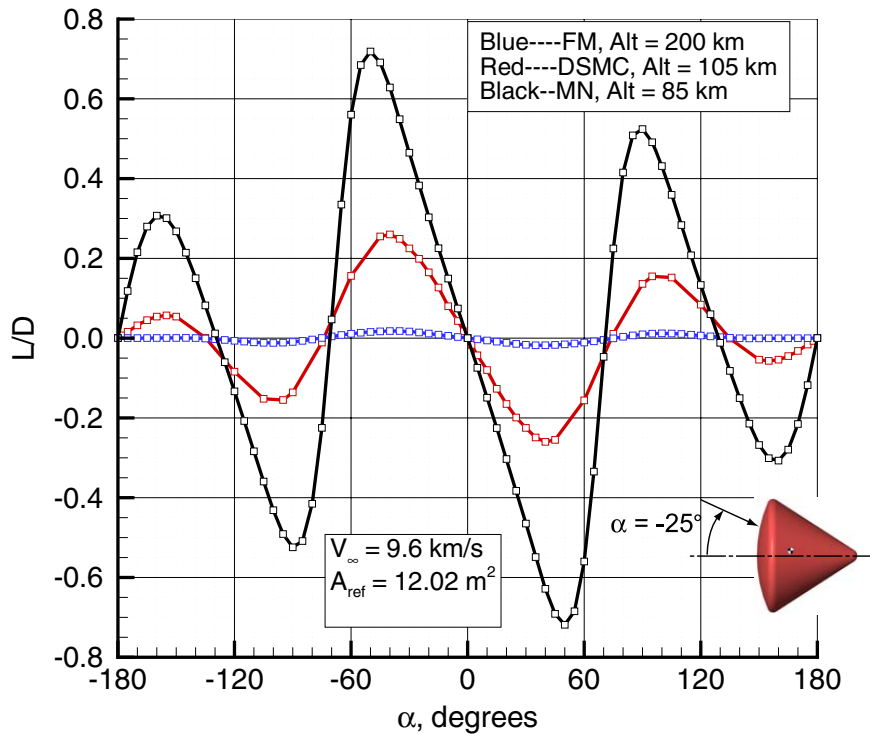
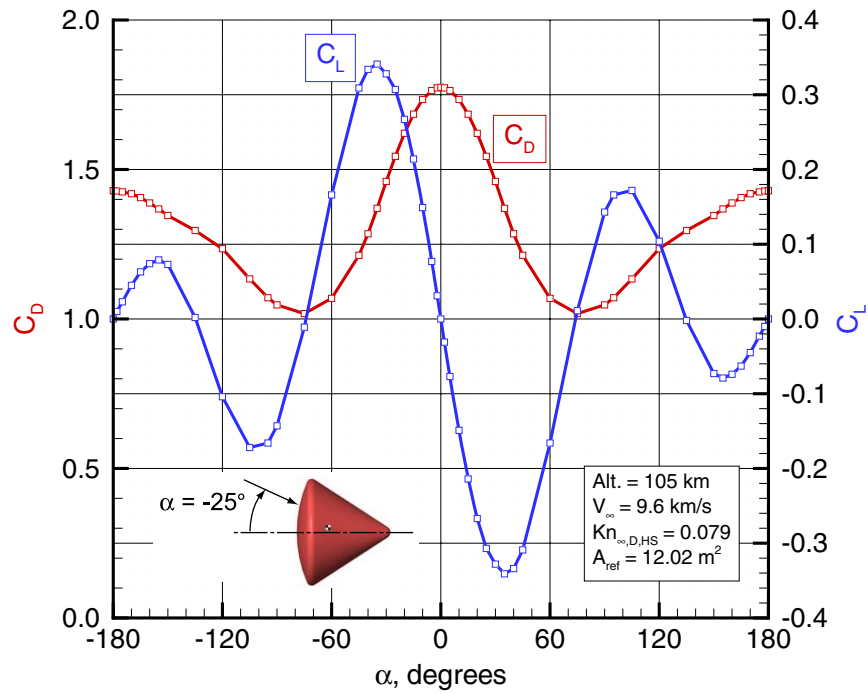
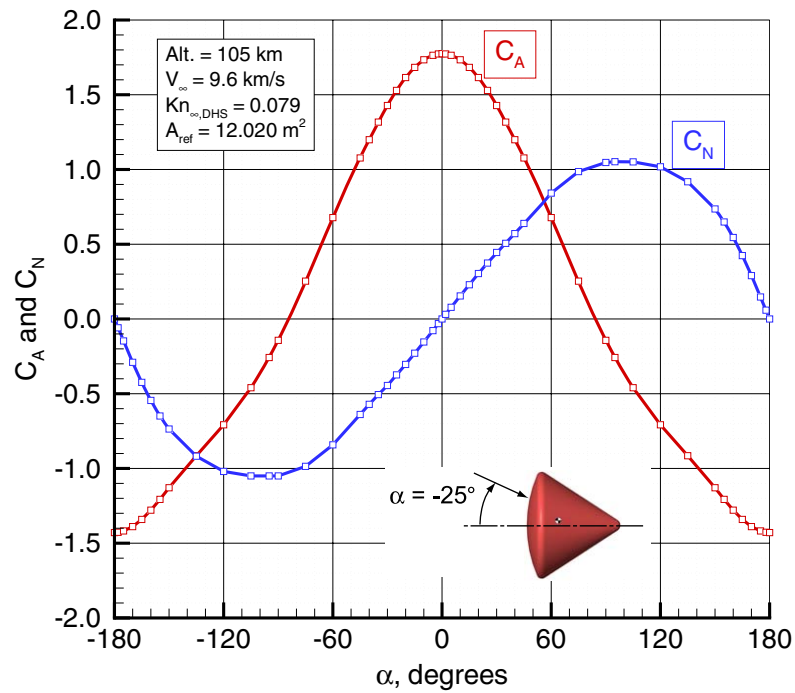


Figure 8. Lift-to-drag (L/D) as a function of incidence for selected altitudes (rarefaction).



(a) Drag and lift coefficients.



(b) Axial and normal coefficients.

Figure 9. Calculated aerodynamics for Apollo capsule at 105 km and 9.6 km/s.

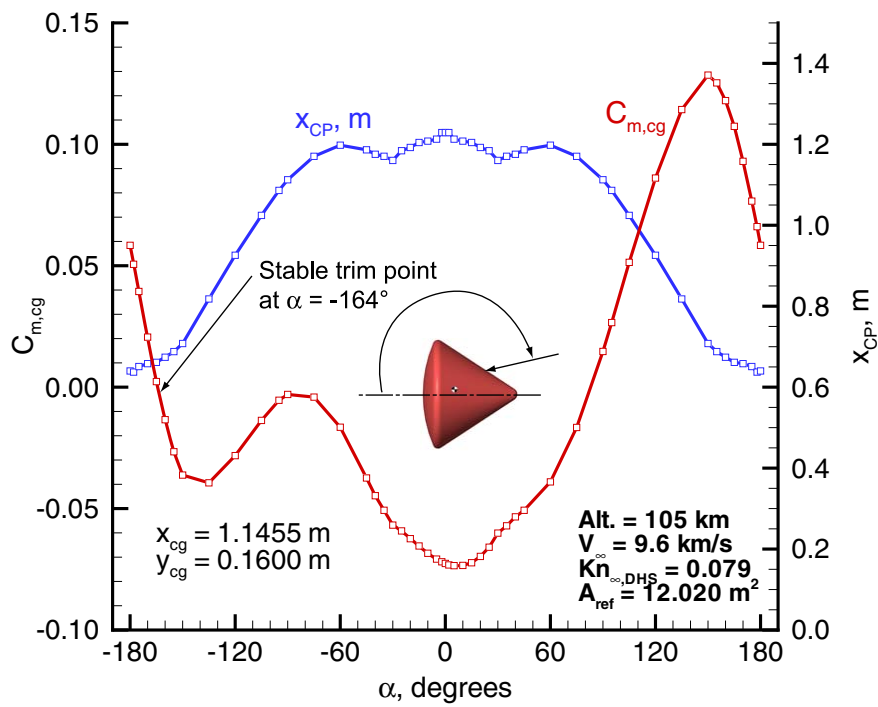
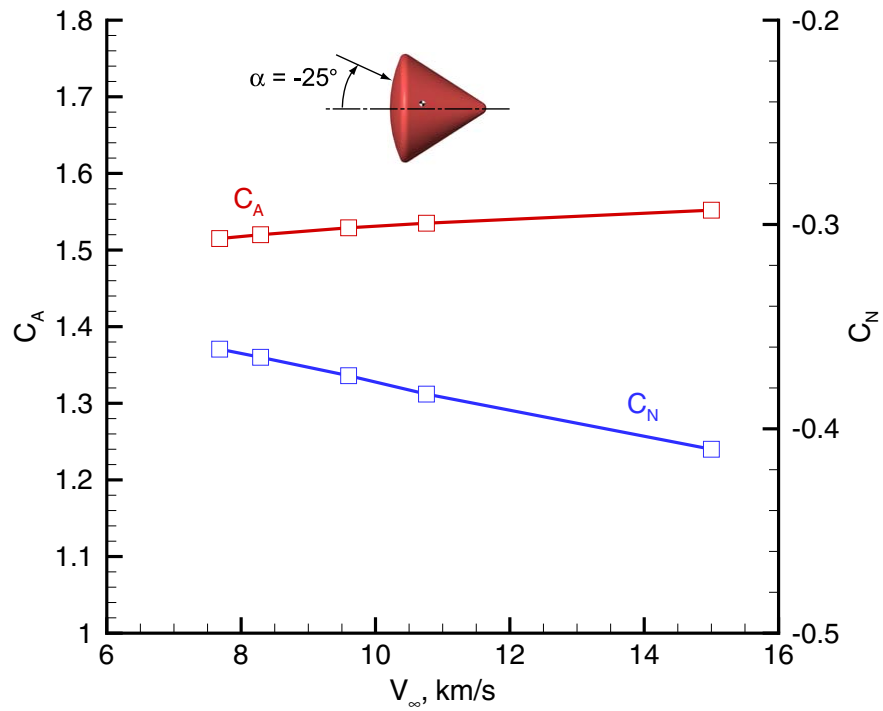
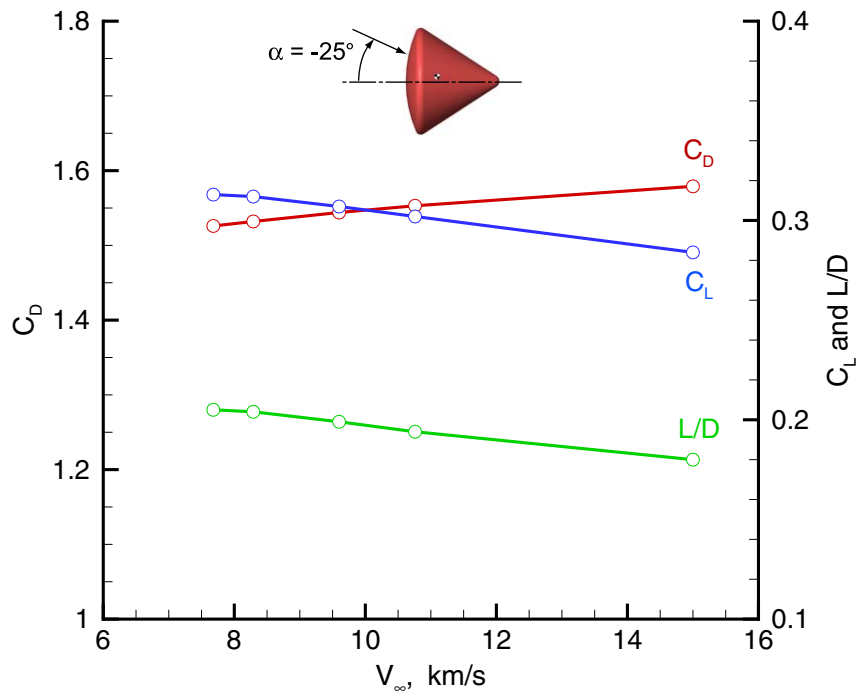


Figure 10. Longitudinal center of pressure location and moment coefficient as a function of incidence.



(a) Axial and normal coefficients.



(b) Lift and drag coefficients.

Figure 11. Sensitivity of aerodynamic coefficients to reentry velocity at 105 km.