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Abstract

The NASA's Evolutionary Xenon Thruster (NEXT) program is developing the next-generation ion propulsion system with significant enhancements beyond the state-of-the-art in ion propulsion to provide future NASA science missions with enhanced mission capabilities at a low total development cost. As part of a comprehensive thruster service life assessment utilizing both testing and analyses, a Long-Duration Test (LDT) was initiated to validate and qualify the NEXT propellant throughput capability to a qualification-level of 450 kg, 1.5 times the mission-derived throughput requirement of 300 kg. This wear test is being conducted with a modified, flight-representative NEXT engineering model ion thruster, designated EM3. As of June 21, 2007, the thruster has accumulated 10,100 hr of operation at the thruster full-input-power of 6.9 kW with 3.52 A beam current and 1800 V beam power supply voltage. The thruster has processed 207 kg of xenon and demonstrated a total impulse of 8.5×10^6 N-s; *the highest total impulse ever demonstrated by an ion thruster in the history of space propulsion*. Thruster performance tests are conducted periodically over the entire NEXT throttle table with input power ranging 0.5 to 6.9 kW. Overall ion thruster performance parameters including thrust, input power, specific impulse, and thruster efficiency have been nominal with little variation to date. Lifetime-limiting component erosion rates have been consistent with the NEXT service life assessment, which predicts the earliest failure sometime after 750 kg of xenon propellant throughput; well beyond the mission-derived lifetime requirement. The NEXT wear test data confirm that the erosion of the discharge keeper orifice, enlarging of nominal-current-density accelerator grid aperture cusps, and the decrease in cold grid-gap observed during the NSTAR Extended Life Test have been mitigated. This paper presents the status of the NEXT LDT to date.

Nomenclature

$A_{D/S}$	accel aperture downstream area, m ²
d	axial distance, m
d_{Dome}	accel grid dome height, m
J_B	beam current, A
J_{NK}	neutralizer keeper current, A
J^+	singly-charged ion current, A
J^{++}	doubly-charged ion current, A
\mathcal{L}	aperture length, m
m_C	discharge cathode flowrate, sccm
m_M	main plenum flowrate, sccm
m_N	neutralizer cathode flowrate, sccm
P_{IN}	thruster input power, kW

P_t	neutral total transmission factor
R	accel grid spherical radius, m
R_1	aperture radius, m
w_{Dome}	accel grid dome width, m
V_A	accelerator grid voltage, V
V_B	beam power supply voltage, V
V_{DC}	discharge voltage, V
x	radial distance, m
β	accel aperture wall half-angle, deg
ϕ	diameter

Introduction

NASA's Evolutionary Xenon Thruster (NEXT), led by the NASA Glenn Research Center (GRC), is being developed to replace existing state-of-the-art ion propulsion technology with a more-advanced, higher-power ion propulsion system (IPS) at low total development cost. The success of the NASA Solar Electric Propulsion Technology Applications Readiness (NSTAR) ion propulsion system on Deep Space 1 secured the future for ion propulsion technology for future NASA missions (refs. 1 to 3). In-space propulsion technology analyses conducted at NASA identified the need for a higher-power, higher total throughput capability ion propulsion system beyond the 2.3 kW NSTAR ion thruster targeted for robotic exploration of the outer planets. The NEXT project initially targeted Flagship-class Deep Space Design Reference Missions (DSDRM) such as a Titan Explorer or Neptune orbiter assuming aerocapture at the destinations as the design driver mission applications (refs. 4 and 5). A refocus study was conducted in 2004 to assess mission benefits of the NEXT IPS for Discovery- and New Frontiers-class missions. Several of the Discovery-class mission studies demonstrated NEXT outperforming the state-of-the-art (SOA) NSTAR, yielding higher net payload mass with fewer thrusters (ref. 6). Several of the New Frontiers and Flagship-class mission studies showed that NEXT was either mission-enhancing or mission-enabling (ref. 7). NEXT technology is applicable to a wide range of NASA solar system exploration missions, as well as earth-space commercial and other missions of national interest. NEXT affords larger delivered payloads and smaller launch vehicle size than chemical propulsion for Discovery, New Frontiers, Mars Exploration, and Flagship outer-planet exploration missions.

The NEXT system consists of a high-performance, 7 kW ion thruster; a high-efficiency, modular, 7 kW power processing unit (PPU)¹ with an efficiency and a specific power greater the NSTAR PPU; a highly-flexible, advanced xenon propellant management system (PMS)² that utilizes proportional valves and thermal throttles to reduce mass and volume; a lightweight engine gimbal³; and key elements of a digital control interface unit (DCIU) including software algorithms (refs. 8 to 14). The NEXT thruster and component technologies demonstrate a significant advancement in technology beyond SOA NSTAR thruster systems. NEXT performance exceeds single or multiple NSTAR thrusters over most of the thruster input power range. The wet propulsion system mass has been reduced by higher-efficiency, higher-specific impulse, and lower specific mass. With a predicted throughput capability more than double that of NSTAR, fewer NEXT thrusters are required compared NSTAR.

Validation of the NEXT thruster service life capability is being addressed via a comprehensive service life validation scheme utilizing a combination of test and analyses. A NEXT service life assessment was conducted at GRC employing several models to evaluate all known failure modes incorporating the results of the NEXT 2,000 hr wear test conducted on an engineering model (EM) NEXT ion thruster at 6.9 kW input power. The assessment predicts the earliest failure occurring sometime after 750 kg of xenon throughput; well beyond the mission-derived lifetime requirement of 300 kg (ref. 15). To

¹Power Processing Unit development led by L3 Comm ETI (Torrance, CA).

²Propellant Management System and DCIU simulator development led by Aerojet (Redmond, WA).

³Gimbal development led by the Jet Propulsion Laboratory and Swales Aerospace.

validate the NEXT thruster service life model and qualify the NEXT thruster, the NEXT Long-Duration Test (LDT) was initiated. The purpose of the NEXT LDT is to 1) characterize thruster performance over the test duration, 2) measure the erosion rates of all thruster components, 3) identify unknown life-limiting mechanisms, and 4) demonstrate 1.5 times the mission-derived propellant throughput requirement resulting in a qualification propellant throughput requirement of 450 kg. The NEXT thruster service life analyses will be updated based upon the LDT data and findings if needed.

Test Article

The NEXT LDT is being conducted with an engineering model ion thruster, designated EM3, shown in figure 1. The EM3 thruster has been modified to a flight-representative configuration so it is more comparative to the NEXT prototype-model (PM) thruster through incorporating PM ion optics and a graphite discharge cathode keeper electrode. To reduce the risk of a facility-induced failure of the thruster, the neutralizer assembly was enclosed to protect insulators from sputter deposition and all critical surfaces were grit-blasting for flake retention. The PM ion optics active area was reduced to 36 cm diameter to reduce outer-radius accelerator aperture erosion caused by beamlet over-focusing in these low current density regions. Reducing the ion optics active area from 40 cm also reduces the maximum thruster beam divergence and neutral loss rate without a significant increase in discharge losses (ref. 8). The PM ion optics geometry retains many of the key features of the EM design, however, improved manufacturing techniques implemented by a new vendor led to better control of aperture variation as a function of grid radius, a reduced and more consistent cusp profile, and elimination of “worm track” surface problems previously encountered (ref. 13). The aperture variation for the PM accelerator grid is +1/–6 percent compared to +11/–16 percent for EM optics (ref. 13). One of the unexpected findings from the NSTAR Extended Life Test (ELT) was the anomalous discharge cathode keeper erosion, which was more severe and qualitatively different than prior 1,000 and 8,200 hr NSTAR wear tests (refs. 16 to 18). Due to the complete NSTAR ELT discharge cathode keeper faceplate erosion and the NEXT EM 2,000 hr wear test results, a graphite discharge cathode keeper is employed on EM3, similar to the NEXT PM thruster design, to mitigate keeper erosion. The erosion rate of carbon due to the low-energy discharge plasma ion impacts is over 20 times lower than molybdenum, thus utilization of a graphite keeper electrode dramatically extends thruster service life (ref. 19).

The NEXT thruster is nominally a 0.5 to 6.9 kW input power xenon ion thruster with 2-grid ion optics, screen (+) upstream and accelerator (–) downstream. The technical approach for the NEXT design is a continuation of the derating philosophy used for the NSTAR ion thruster. A beam extraction area 1.6 times NSTAR allows higher thruster input power while maintaining low voltages and ion current densities, thus maintaining thruster longevity. The semi-conic discharge chamber utilizes a hollow cathode emitter with a ring-cusp magnetic topology created by high-strength, rare earth magnets for electron confinement. A flake retention scheme identical to that employed on the NSTAR thruster enhances the adhesion of thin films to the discharge chamber surfaces (ref. 20). New, compact propellant isolators with higher voltage isolation capability than those used by the NSTAR thruster are utilized. The NEXT neutralizer design is mechanically similar to the International Space Station Plasma Contactor leveraging this extensive database to reduce risk. Additional description of the NEXT EM3 thruster design can be found in references 21 to 25.

Test Support Hardware

The following section briefly describes the NEXT LDT supporting hardware. More detailed descriptions can be found in references 25 to 28.

Vacuum Facility

The NEXT LDT is being conducted in the 2.7 m diameter by 8.5 m long Vacuum Facility 16 (VF-16) at NASA GRC, shown in figure 2. VF-16 has an emergency bell jar on the end cap into which the thruster can be withdrawn and isolated in the event of a facility emergency. VF-16 is equipped with 10 cryogenic pumps for nominal thruster operation and an additional cryo-pump on the isolated bell jar for emergency use. With all 10 cryo-pumps operating, the base pressure is less than 3×10^{-7} Torr. Facility pressure is monitored by two ionization gauges, a glass-tube ion gauge located on the facility wall 0.5 m downstream of the thruster and a dual-filament nude near-thruster ion gauge mounted 0.5 m radially beside EM3. In addition, the isolated bell jar has an ion gauge that is turned off during normal operation. The measured facility pumping speed, corrected for xenon, is 180 kL/s using the wall-mounted ion gauge. With 10 operational cryo-pumps, the near-thruster facility background pressure is 2.5×10^{-6} Torr, corrected for xenon, when the thruster is operating at full-power. A quadrupole residual gas analyzer (RGA) measures and records the quantity of individual gas species inside VF-16 continuously every minute. All interior surfaces downstream of the thruster are lined with 1.2 cm thick graphite paneling to reduce the back-sputtered material flux to the thruster and test support hardware. The back-sputter rate, nominally $3 \mu\text{m}/\text{hr}$, is monitored by a quartz-crystal microbalance (QCM) located next to the ion thruster. Three pinhole cameras are mounted next to the QCM and will be microscopically analyzed at the conclusion of the life test to determine the source of back-sputtered material. In addition to the pinhole cameras, five quartz witness plates are mounted along the length of the vacuum chamber wall.

Power Console and Xenon Feed System

A power console consisting of commercially available power supplies, similar to that described in reference 29, powers the ion engine. A high-purity gas feed system provides xenon to the discharge cathode, discharge chamber main plenum, and neutralizer cathode through separate mass flow controllers. Xenon can be supplied to the discharge chamber main plenum by either a mass flow controller or an engineering model propellant management system kernel provided by Aerojet.

Diagnostics

A computerized data acquisition and control system is used to monitor and record ion engine and facility operations. Data are sampled at a frequency range of 10 to 20 Hz and stored every minute. A set of data consists of individual mass flow rates, ion engine currents measured with current shunts, voltages measured with voltage dividers, facility pressures, and the QCM measurement. As part of the periodic thruster characterization, the thruster is connected to an electrically floating power supply circuit used to determine screen grid ion transparency and discharge keeper ion current. The circuit electrically ties the screen grid or discharge keeper to the discharge cathode during normal operation, but biases the screen grid or discharge keeper negative relative to discharge cathode potential to repel electrons and measure the collected ion current.

Ion beam diagnostics include staggered planar probes mounted onto a translation stage to measure radial ion current density profiles and an ExB probe, or Wien Filter, to measure the doubly-to-singly charged ion signature. Each molybdenum Faraday probe has 1-cm^2 circular current-collection area and is biased -30 V relative to facility ground to repel electrons. The collected currents are measured through separate isolated shunt resistors. The ExB probe is positioned 7.6 m downstream of the thruster on centerline, yielding a doubly-to-singly charged ion signature in the far-field. The ExB probe design is described in reference 30. The Faraday probes and ExB probe are protected from the high-energy ion beam by parking the probes outside the beam and protection by a graphite shutter, respectively. The LDT ion beam diagnostics are described in detail in reference 27.

Erosion of critical ion engine components is monitored by six in-situ CCD cameras, shown in figure 3, which capture erosion patterns and wear rates throughout the life test. The cameras image the

downstream neutralizer keeper and cathode orifice plate, the discharge cathode keeper and cathode orifice plate, accelerator grid apertures at various radial locations from centerline, and the cold grid-gap of the thruster ion optics. The cameras are mounted to a single-axis positioning system that moves the cameras radially in front of the thruster.

Operating Condition and Performance Tests

The NEXT ion thruster is designed for solar electric propulsion (SEP) applications that experience variation in power available as solar flux changes at various distances from the sun throughout the mission. Ion thruster input power is designed to be throttled from 0.5 to 6.9 kW to accommodate this variation in available power. It is necessary to demonstrate the engine's propellant throughput capability at the worst erosion, most thermally stressful, and electrically taxing operating point. The thruster full-input-power operating condition, table 1, corresponds to 1800 V beam power supply voltage and 3.52 A beam current, met these criteria. The input power indicated in table 1 is a nominal operating power requirement from the NEXT throttle table at the thruster beginning-of-life and may differ slightly from thruster to thruster (ref. 22). The NEXT LDT consists of three operating phases 1) operate at the full-power point until the nominal mission requirement of 300 kg propellant throughput has been demonstrated, 2) throttle according to a mission-representative profile after 300 kg, and 3) throttle to full-power until either decision is made to end the test or the thruster fails. The goal of the LDT is to demonstrate the qualification level propellant throughput of 450 kg, 1.5 times the mission-derived requirement.

Throughout the LDT, performance characterization tests are conducted to assess performance of the thruster and thruster components at several power levels that envelope the entire NEXT throttle table, listed in table 3. Periodic component performance assessments of the discharge chamber, ion optics, and neutralizer cathode are performed at the various thruster operating conditions. Ion optics performance includes electron backstreaming, perveance, and screen grid ion transparency measurements. Discharge chamber performance is assessed by measuring discharge losses as a function of discharge propellant utilization efficiency for fixed discharge voltages. Neutralizer performance is evaluated by measuring dc keeper voltage, ac keeper voltage, and ac keeper current as a function of neutralizer flow for a fixed neutralizer current.

Thruster Performance

As of June 21, 2007, the NEXT EM3 thruster has accumulated 10,100 hr of operation, 98.5 percent of which has been at the full-input power operating condition. Nonfull-power operating time has been due to the performance characterization testing to date. The thruster has processed 207 kg of xenon, more than 2.5 times that of the Deep Space 1 NSTAR flight thruster, as illustrated in figure 4. Figure 4 shows the NEXT LDT propellant throughput as a function of elapsed time with reference to previous NSTAR wear tests, the individual thruster requirements of the NSTAR thrusters used on the DAWN mission – 202 kg, the DAWN total throughput requirement – 404 kg, and the NEXT qualification level throughput – 450 kg (refs. 31 to 33). The NEXT thruster has demonstrated a total impulse of 8.5×10^6 N-s to date; *the highest total impulse ever demonstrated by an ion thruster in the history of space propulsion*. This is also the highest total impulse ever demonstrated by an electric propulsion device with an input power less than 10 kW (ref. 34). The NEXT LDT total impulse demonstrated exceeds that of the 30,000 hr NSTAR ELT in less than 1/3rd the thruster operating duration, shown in figure 5.

Performance of the thruster and thruster components has been steady with minimal degradation. Erosion of critical components has been within modeling predictions and consistent with the NEXT service life assessment. Several of the NSTAR ELT-observed anomalies have been significantly reduced and in most cases eliminated. There has been no observed discharge cathode keeper orifice erosion, no measured increase in the nominal-beam-current-density accelerator grid aperture cusps, and no measured

change in the cold grid-gap of the ion optics for the NEXT engine – all of which were observed during the NSTAR ELT of the DS1 flight spare (ref. 33).

The following section describes the observed trends in thruster performance at the full-input-power operating point and during the performance characterization tests conducted periodically at various operating conditions over the range of the NEXT throttle table. For document flow and ease of reading, the thruster runtime, performance characterization data, and erosion data are located at the end of the document.

Engine Performance

Thruster performance parameters that are important for mission planning such as thrust, specific impulse, input power, and efficiency are plotted at the nominal full-power operating point as a function of time in figures 28 and 31. For thrust calculations, the beam divergence thrust correction factor and the total doubly-to-singly-charged ion current ratio are assumed to be in the 0.975 and 0.043, respectively, based upon the methodology developed for NSTAR thrusters (ref. 35). Ingested mass flow due to facility background pressure was included in the total mass flow rate to the engine for determining specific impulse and thrust efficiency (ref. 36). At full-power, thrust and specific impulse have remained constant at values of 237 ± 3 mN and 4170 ± 70 s, respectively. The indicated uncertainty in thruster performance values are discussed in reference 37. Noise in the runtime data is predominantly due to changes in the beam current, which is maintained by manually adjusting the discharge current. Spikes in the data are due to thruster shutdowns and restart events. At full-power, thrust efficiency decreased from the beginning-of-test value of 0.710 to 0.707 due to an increase in input power to the thruster from 6.85 to 6.87 kW. The increase in thruster input power is a result of increasing thruster discharge losses and will be discussed in the additional analyses section. Increases in specific impulse and thrust efficiency are observed at the beginning of the test due to an intentional decrease in neutralizer flow from 5.16 to 4.01 sccm to improve overall propellant utilization efficiency.

Performance parameters for all of the characterized throttle levels calculated thrust, specific impulse, input power, and efficiency are plotted in figures 29, 30, 32, and 33, respectively. For thrust calculations, the beam divergence thrust correction factor and the total doubly-to-singly-charged ion current ratio ranged from 0.962 to 0.975 and 0.028 to 0.060, respectively, based upon the methodology developed for NSTAR thrusters and ingested mass is included in specific impulse and thrust efficiency calculations (ref. 35). Trends at all operating conditions are similar to the full-power operating point: constant thrust, constant specific impulse after the neutralizer flow decrease at beginning-of-test, slightly increasing input power due to increasing discharge losses, and slightly decreasing efficiency (after neutralizer flow decrease) due to increasing input power.

Discharge Chamber

The discharge cathode and main plenum flow rates are set to 4.87 and 49.6 sccm during full-power operation, respectively. Full-power discharge voltage and current are shown as a function of time in figure 34. Discharge current and voltage have increased over the duration of the test from 18.6 A, 23.3 V to 19.3 A, 23.9 V after 10,100 hr of operation, respectively. The increasing discharge current, required to maintain a constant beam current due to increasing neutral transparency as accelerator apertures erode, is expected from past ion thruster wear testing (refs. 16 and 18). The 4 percent increase in discharge current produces an increase in the discharge primary electron number density offsetting the reduction in discharge chamber neutral number density resulting from the chamfering of accelerator apertures to maintain the same discharge chamber ion production. As will be shown in the analysis section, it is possible to use accelerator aperture erosion measurements to predict the change in discharge current required to maintain the beam current. The beginning-of-test decrease and overall 3 percent increase in discharge voltage may be influenced by the following: the buildup of a resistive coating on the discharge chamber, changes in neutral density and electron temperature caused by increasing neutral transparency,

changes in the electron emitter surface conditions or heat transfer, or other changes in the discharge chamber conditions.

Discharge voltage and current data for each of the characterization conditions are shown in figures 35 and 36, respectively. The overall trends for each operating condition with time are similar to that of the full-power test condition, namely increases in voltage and current of a few percent. The beginning of test decrease in discharge voltage is more pronounced for the lower-power operation conditions.

At full-power, discharge propellant utilization efficiency and discharge losses as a function of time are shown in figure 37. Discharge propellant utilization efficiency, which is the beam current divided by discharge propellant flow rate (including ingested flow) in equivalent amperes, has been constant at 0.89. Full-power discharge loss, discharge power divided by the beam current, has risen from 123 to 131 W/A over the 10,100 hr of operation. The gradual 6 percent increase in discharge losses is in contrast to the larger and more rapid beginning-of-test increase in discharge losses exhibited by the NSTAR thruster. NSTAR thruster full-power discharge losses, considerably higher (~50 W/A) than NEXT due primarily to the smaller discharge chamber, increased by 10 to 15 W/A within the first 500 hr of operation in three separate wear test (refs. 16, 17, and 38). The NEXT thruster initial change in discharge losses has been small compared to that of NSTAR thrusters during the NEXT LDT and the NEXT 2,000 hr wear test, illustrated in figure 6, resulting in more constant thruster input power and thrust efficiency at the beginning of life (refs. 16, 17, 28, 38, and 39). The reduced beginning-of-life (BOL) increase in discharge losses in the NEXT design is a result of a flatter NEXT beam profile, thicker accelerator grid, smaller cusp ion optics, and more focused beamlets at the full-power operating condition. Furthermore, the ~2 W/A LDT initial increase in discharge losses are less than the ~5 W/A increase observed in the NEXT 2,000 hr wear test due to the incorporation of PM ion optics on the EM3 thruster. The PM optics have reduced cusp size, increased aperture uniformity compared to EM optics, and 36 cm diameter active area (removing the outer 2 cm apertures where over-focusing occurs), which reduce the BOL increase in discharge losses (ref. 13). After 207 kg, the NEXT LDT full-power discharge losses increased by 8 W/A compared to the NSTAR ELT increase of 22 W/A after 210 kg (refs. 18 and 39).

Discharge propellant utilization efficiency and discharge losses are plotted as a function of time for each of the operating conditions characterized during the LDT in figures 38 and 39. Constant discharge propellant utilization efficiencies are observed after the initial beginning-of-test point with the higher neutralizer flow that led to higher facility pressure and ingested flow, most prominent in the 1 A beam current operating condition. Discharge losses are lowest for the full-power operating condition and increase with decreasing total voltage, i.e., the sum of the absolute values of the beam and accelerator power supply voltages. That the discharge losses increases with decreasing total voltage is largely due to a decreasing screen grid ion transparency (ref. 23). Modest increases in discharge losses, ≤6 percent, are observed for each of the operating conditions consistent with the trend at full-power.

Figure 40 shows discharge losses as a function of discharge propellant utilization efficiency for the full-power and a low-power operating conditions throughout the wear test. The ratio of main to discharge cathode flow rate is adjusted to maintain the nominal discharge voltage. No significant changes in the functional dependence of discharge losses on propellant utilization efficiency are observed. Shifting of the curves up by ~8 W/A at full-power is consistent with the increase in runtime data discharge losses.

Figure 41 shows the discharge keeper voltage relative to discharge cathode potential and discharge keeper ion saturation current as a function of time at full-power throughout the wear test. These parameters directly impact the discharge keeper wear rate. The keeper ion current increased by ~4 percent, consistent with the observed increase in discharge current and indicates an increase in localized ion density near the discharge cathode assembly (DCA). The discharge keeper voltage increased from 4.3 to 5.3 V over the first 10,100 hr of the test following the observed increase in discharge voltage from 23.3 to 23.9 V. There have not been any unexpected changes in discharge keeper voltage and current during the LDT, such as keeper shorting to cathode common or any step-changes, indicating that the discharge keeper wear rate at full-power has been relatively constant. The NEXT LDT full-power discharge keeper ion saturation current and voltage are consistent with the NEXT 2,000 hr wear test indicating similar levels of ion impingement.

Discharge cathode ignition durations are plotted in figure 42 as a function of elapsed time. With the exception of the post-facility-regeneration ignitions, all discharge cathode ignitions have been less than 5 min. After 2,000 hr, the post-regeneration ignition durations began to increase with thruster operating time. The highest ignition duration, ~24 min, occurred following an extended period of thruster downtime for cryo-pump repairs. After 7,000 hr of operation, a xenon purge of 4.5 sccm discharge cathode flow was maintained during the facility regenerations in order to maintain collisional flow inside the cathode tube. The purge flow has effectively reduced the post-facility-regeneration ignition durations to their nominal values. The exact cause of the post-regeneration increased ignition durations is unknown, but is a result of facility effects due to exposure to elevated pressures (up to 50 mTorr) during regenerations. One possibility is absorption of moisture by the graphite keeper, which acts as a getter for moisture during the facility regeneration. This moisture would not necessarily be driven off by the cathode conditioning sequence, utilized prior to ignition, as the procedure was developed for cathodes with metal keeper electrodes. The absorbed moisture may be released during the subsequent cathode ignition exposing the emitter to higher levels of moisture and oxygen. The cathode purge flow may be reducing moisture absorption on the interior surfaces of the keeper and orifice. Regardless, a flight thruster in space would not experience this type of repeated exposure and therefore this issue is limited to ground-based testing.

Neutralizer

Neutralizer flow and neutralizer keeper current are set to 4.01 sccm and 3.00 A, respectively, for the full-power operating condition. Neutralizer keeper voltage, relative to neutralizer cathode common, and the coupling voltage, which is neutralizer cathode common relative to vacuum facility ground, are plotted at the full-power condition as a function of time in figure 43. Following the initial increase in keeper voltage at the beginning of the test due to an intentional decrease in the neutralizer flow rate, the keeper voltage has decreased slightly from 11.2 to 10.7 V over 10,100 hr. As with most of the thruster runtime data, “spikes” are observed in the neutralizer keeper voltage corresponding to engine restarts, which is similar to NSTAR wear test and NEXT 2,000 hr wear test neutralizer behavior (refs. 16, 17, 28, and 38). The mean coupling voltage has been steady at $-10.2 \text{ V} \pm 0.2 \text{ V}$ over the duration of the test. The low coupling voltage magnitude is a result of high keeper current and neutralizer flow rate, selected to maintain neutralizer operation in spot mode throughout ion thruster service life while imposing only modest sacrifices in engine performance. The neutralizer keeper and coupling voltages indicate no neutralizer performance degradation.

Figure 44 shows the neutralizer flow margin, which is the difference between the set point and the transition flow from spot to plume mode operation, as a function of elapsed test duration for various beam currents. Spot mode is characterized by low voltage and current oscillations, while plume mode is described by large fluctuations that can lead to reduced emitter life. Following the NSTAR criterion, plume mode operation is reached when peak-to-peak neutralizer keeper voltage oscillations exceed $\pm 5 \text{ V}$. This definition includes some margin. As figure 44 illustrates, there is considerable margin at the full-power condition. Transition flow margin has decreased slightly for all beam current conditions over the test duration. The least neutralizer flow margin exists for the low-beam current operating conditions where as little as 0.1 sccm margin is available (for 1 A beam current). Because of this limited margin, the neutralizer keeper gap on the PM thruster design has been increased slightly resulting in double the flow margin at the modest expense of $\sim 1 \text{ V}$ increase in the magnitude of the coupling voltage (ref. 8).

Neutralizer cathode ignition durations have been steady and typically less than 4 min as shown in figure 45. The longest duration occurred following the extended thruster downtime for cryo-pump repairs, 5.5 min. The constant neutralizer ignition durations, even after facility regeneration, offers additional evidence to support the role of the graphite discharge cathode keeper in the observed increasing discharge cathode post-regen ignitions. The neutralizer cathode assembly (NCA) utilizes a metal keeper electrode that will not absorb moisture as effectively as graphite. As a precaution, xenon purge flow of 1.5 sccm through the neutralizer cathode has been maintained during facility regenerations since 7,000 hr.

Ion Optics

Beam power supply and accelerator voltages are set to 1800 and -210 V, respectively, during wear test at the full-power point. Accelerator current and wall-mounted facility ion gauge pressure data are plotted as a function of time in figure 46. As expected, increases in facility pressure cause corresponding increases in accelerator current due to the increased background pressure resulting in higher charge-exchange ion production. Beginning-of-life accelerator currents decreased by 1.5 mA during the first 100 hr of operation, the cause of which is threefold. First, from test start to 49 hr, the vacuum facility had only 9 of the 10 cryo-pumps operating resulting in a higher background pressure accounting for 0.5 mA of the 1.5 mA decrease observed. Second, at 94 hr, the neutralizer flow was reduced to increase propellant utilization efficiency causing a lower background pressure accounting for 0.3 mA of the 1.5 mA decrease observed. Finally, the thruster ion optics experienced a “burn in” period that is typical to ion thrusters. In NEXT ion thrusters operating at full-power, this burn in is primarily due to enlargement of accelerator grid apertures at large ion optics’ radii where the ion current density is low enough to cause direct crossover impingement.

Shifts in the facility pressure and accelerator current runtime data after beginning-of-test are due to changes in facility configuration, i.e., the number of operational cryo-pumps, or the effectiveness of cryo-pump(s). Failure of cryo-pumps during extended wear testing is expected over the course of the test based upon the NSTAR ELT experience and repair of failed cryo-pumps requires removal from the facility (ref. 33). The increase of 0.7 mA at ~ 1900 hr was due to change in facility operation from 10 to 9 pumps, with the nonoperating pump located near the thruster. 9-pump operation was maintained until $\sim 6,300$ hr when 10-pump operation resumed and the accelerator current and pressure dropped accordingly. At $\sim 6,600$ hr a cryo-pump at the far end of the facility was turned off, but there was no observed change in accelerator current. From 6,900 hr until the 9,500 hr, 8 cryo-pumps were operated, with the two nonoperating cryo-pumps located at the far end of the tank from the thruster. The accelerator currents for 8-pump operation with both non-operating cryos located far from the thruster and 9-pump operation with the nonoperating cryo located near the thruster are within 0.2 mA. The indicated dip in facility pressure from 4,000 to 4,500 hr was a measurement error due to contamination of the ion gauge filament that was not degassed periodically, as recommended by the manufacturer, until after this time. Since 9,500 hr, 9 cryo-pumps have been operated with the nonoperating pump located at the far end of the vacuum tank from the thruster. “Spikes” in accelerator current are due to engine shutdowns, restarts, recycles, and facility pressure spikes as the need for a facility regeneration is approached (refs. 25 and 40).

Figure 47 shows the accelerator currents for each of the performance operating conditions as a function of time. The data at alternate operating conditions are consistent with the full-power data. An initial decrease in accelerator current is observed at the beginning of the test due to increased pumping speed when transitioning from 9 to 10 cryo-pumps, the neutralizer flowrate decrease, and the outer radii accelerator grid aperture enlargement. After the beginning of the test, the accelerator current trends have been dominated by changes in the facility background pressure. In contrast, NSTAR ion thruster accelerator currents in wear tests have generally started higher than nominal requiring up to 1,500 hr to decrease to nominal values resulting in greater variability in performance and erosion (refs. 16 and 38). The accelerator current for NSTAR on DS1 was ~ 25 percent less in space than the NSTAR data obtained during pre-flight measurements in a test facility operating with a background pressure level of 3.5×10^{-6} Torr (refs. 3, 33, and 38). Because the NEXT LDT is operating at comparable operating background pressures, it is expected that the NEXT in space accelerator current would be ~ 25 percent less than those measured in this test facility.

Electron backstreaming limits, perveance limits, and screen grid ion transparencies throughout the wear test are plotted for each operating condition in figures 48 to 50, respectively. Electron backstreaming limit is the highest accelerator voltage, lowest in magnitude, which will prevent beam plasma electrons from backstreaming through the ion optics. Electron backstreaming limits are determined by lowering the magnitude of the accelerator grid voltage until the indicated beam current increased by 1 mA due to backstreaming electrons. Impingement-limited total voltage is a measure of the ion optics’ current

extraction capability, and therefore a measure of its perveance. Perveance limits are determined from plots of accelerator current as a function of total voltage where the slope is -0.02 mA/V, the NSTAR criterion. The total voltage is defined as the sum of the beam power supply voltage and the absolute value of the accelerator grid voltage. Screen grid ion transparencies are calculated as described in reference 41.

Electron backstreaming limits and perveance limits have been nominal, within measurement error, over the duration of the test indicating no substantial change in accelerator aperture cusps or grid-gap. A modest decrease, up to 2 percentage points, in screen grid ion transparency is observed over the duration of the test likely due to the increasing sheath thickness on the screen grid caused by the increasing discharge voltage. Changes in electron backstreaming limit, perveance limit, and screen grid ion transparencies are not significant enough to degrade the ion optics' performance and are less than or equal to those exhibited by the NSTAR ion optics during the 8,200 hr wear test and NSTAR ELT (refs. 16 and 18). Electron backstreaming margin and perveance margin are equal to their pre-test values, which for the full-power test condition are ~ 37 and 950 V, respectively. With such large electron backstreaming margin, there is an option to reduce the full-power accelerator voltage magnitude, which was based upon EM ion optics' performance, in order to reduce accelerator erosion.

A recycle event is a series of power supply commands that follows a high-current arc between ion optic grids to prevent instigation of subsequent arcs during which the following occur sequentially the grid potentials are commanded to zero, the discharge current is reduced, the accelerator grid potential is reapplied, the screen grid potential is reapplied, and finally the discharge current is increased back to the nominal value. The LDT recycle rate, averaged over 15 hr, and the total number thruster recycles as a function of time are plotted in figure 51. The recycle data has been corrected by removing recycles occurring during performance testing where numerous recycles can be induced, such as during perveance measurements. The recycle rate has been 1 to 3 recycles per hour over the majority of the test duration. One noticeable exception occurs at $\sim 6,300$ hr where as many as 9 recycles per hour were observed for a short duration. This peak occurred following the extended thruster downtime for facility repairs. Approximately $19\ \mu\text{m}$ of back-sputtered carbon had accumulated on the thruster optics at this time and during the downtime, multiple facility regenerations were executed. Absorption of even small amounts of water by the thin carbon films can lead to spalling, which is visible from the neutralizer enclosure and front mask and was likely the cause of the increased recycle rate observed (refs. 42 to 44). For comparison, the total number of recycles for the NEXT LDT and two NSTAR wear tests are plotted as a function of propellant throughput in figure 7 (refs. 16 and 18). NEXT is expected to demonstrate a slightly higher recycle rate due to its larger beam extraction area and inter-grid electric field at full-power that are 1.6X and 1.5X that of the NSTAR thruster at full-power, respectively. Figure 7 shows that when plotted against propellant throughput, removing the beam extraction area dependence, the recycle rate of the NEXT LDT is similar to the NSTAR wear tests. This is a little unexpected since the total recycles as a function of propellant throughput plot does not account for the increased NEXT inter-grid electric field. Furthermore, the NSTAR ELT was throttled to lower power operating conditions during the test, which resulted in inter-grid electric fields lower than the NSTAR full-power condition (refs. 18 and 33).

Beam Diagnostics and In-Situ Erosion Results

The plume diagnostics and erosion measurement results of the NEXT LDT are presented in this section. A number of high risk or life-limiting items were identified from the DS1 flight spare ELT. The results of the NSTAR ELT were considered and design changes made in the NEXT design to reduce or mitigate thruster failure modes and high risk items listed in table 2.

Radial Beam Current Density Profiles

Radial beam current density profiles are measured as close as 20 mm downstream from the geometric center of the ion optics. No attempt was made to repel charge-exchange ions from the probe or to account for secondary electron emission due to ion bombardment. Sample radial beam current density profiles at

20 mm axial position are plotted in figure 8 for various beam currents demonstrating the shape functions of the ion current density profiles are similar at all beam currents evaluated. The radial profiles are also non-symmetric near centerline, which is an artifact of the engine discharge chamber plasma (refs. 41, 45, and 46).

Assuming azimuthal symmetry, integration of the radial beam current density profile at 20 mm axial location yielded beam currents that are higher than the measured beam current by 8 to 30 percent. Possible sources of error are discussed in references 41 and 45. This error is higher than the 8 percent error observed during the NEXT 2,000 hr wear test (ref. 26). Over the course of the LDT, the peak beam current density at full-power steadily increased as shown in figure 52, most notably for the closest probe. The peak current decreased after multiple sweeps at ~3,700 hr during the performance characterization at multiple operating conditions. This behavior was investigated during the full-power characterization at 8,000 hr. Six sweeps were obtained within 8 hr with various wait times to remove any thermal effects. The results, shown in figure 53, demonstrate a gradual decrease asymptotically approaching a final beam profile. It is theorized that back-sputtered material accumulation on the probe and insulator between the probe and grounded guard has increased the probe collection area. Approximately 30 μm of carbon has been back-sputtered towards the thruster to date. Multiple sweeps through the high-energy beam removes this deposited material. The maximum beam current density for the multiple sweeps at 8,000 and 9,000 hr asymptote to the pre-test maximum value, illustrated in figure 52. Comparison of the beam profiles at axial positions of 20 and 173 mm are shown in figure 54 at 0 hr and the final sweeps at 8000 and 9000 hr illustrating negligible change during the wear test. There is a noticeable change in the probe profile at 238 mm axial position likely due to buildup of back-sputtered material that is not easily removed by the ion beam because of the large distance from the thruster and limited duration in the beam during a typical probe sweep.

At full-power, the beam flatness parameter, defined as the ratio of average-to-peak ion current densities, has ranged from 0.82 to 0.86, over the duration of the test when analyzing only the “clean” probe sweeps. The average ion current density is calculated from the thruster beam current and the active beam extraction area at the thruster. The NEXT beam flatness is a considerable improvement over the NSTAR thruster that had a beam flatness of ~0.5 at full power (refs. 16 and 47). The beam profiles in figure 8 illustrate that the flatness parameter increases with increasing thruster beam current. Ion beam radial current density profiles at the various downstream locations from the ion optics are used to calculate beam divergence half-angles (ref. 41). Beam divergence, defined as the half-angle containing 95 percent of the probe integrated current, has changed from 24.5° to 25.3° at full-power becoming slightly more divergent over the course of 9,000 hr of operation due to enlargement of the accelerator apertures at large radii.

Doubles-to-Singles Ratio

The ExB probe measures the doubles-to-singles signature in the far field over the duration of the test. The doubles-to-singles signatures as a function of elapsed time for various operating conditions are plotted figure 55 showing an increase at full-power from 0.08 to 0.10 over 9,000 hr. For reference the NSTAR ELT doubles-to-singles current ratio varied from 0.12 to 0.20 over the duration of the test (refs. 18 and 33). The increase in the doubles-to-singles ratio for the LDT is caused by the increase in the discharge voltage observed during the wear test. The change in doubles-to-singles by 0.02 for an ~1 V increase in discharge voltage is consistent with the sensitivity observed during the NSTAR ELT (ref. 33). This trend will be discussed in the additional analyses section.

Erosion Results

The discharge cathode assembly (DCA), neutralizer cathode assembly (NCA), accelerator grid center-radius aperture (CRA), mid-radius aperture (MRA) – located 16 cm radially from center, outer-radius apertures (ORA), and ion optics’ cold grid-gap are imaged with their respective CCD cameras. The

locations imaged by the CCD cameras are shown in figure 9 along with the additional test hardware in the picture. The results of the erosion measurements are discussed in the following sections. Erosion images were taken every 200 hr for the first 5,000 hr after which they are obtained at least every 800 hr.

Discharge cathode assembly

Figure 10 shows the discharge cathode at 0 and 9,500 hr and comparison photographs of the NSTAR discharge cathode assembly from the ELT of the flight spare. As time elapses the NEXT discharge cathode faceplate has become slightly textured, but the discharge cathode orifice and keeper orifice diameters, normalized by the beginning-of-test values, have not changed, shown in figure 56. During the NSTAR ELT, the keeper electrode was completely eroded away, exposing the discharge cathode and heater (ref. 48). The NSTAR ELT keeper erosion was characterized by a widening of the keeper orifice compared to erosion observed during the NEXT 2,000 hr and NSTAR 8,200 hr wear tests, in which the most severe erosion was focused towards the mid-radius of the keeper faceplate (refs. 16, 26, and 28). The primary function of the discharge cathode keeper is to protect the discharge cathode from excessive sputter erosion. The EM3 keeper material was changed to graphite, which has a sputter yield an order of magnitude lower than that of the molybdenum at 50 eV (ref. 19). While the LDT results confirm that no enlargement of the keeper orifice has occurred, the erosion of the downstream face of the keeper is not measured in-situ. Discharge cathode keeper downstream surfaces from the NSTAR 8,200 hr and NEXT 2,000 hr wear tests were qualitatively similar with the deepest erosion at a radii of 55 to 60 percent and 40 percent of the total keeper radius, respectively (refs. 16, 26, and 28). Scaling the NEXT 2,000 hr wear test molybdenum discharge keeper erosion rate (depth of 17 percent of the keeper thickness after test) with the decrease in sputter yield of graphite compared to molybdenum gives a conservative estimate of wear through of the keeper after >87 kh at full-power (1800 kg) (refs. 19, 26, and 49).

Neutralizer cathode assembly

Figure 11 shows the NCA pretest and latest image taken after 9,500 hr of thruster operation. Texturing of the neutralizer cathode faceplate is observed and a darkening of the keeper is seen due to back-sputtered carbon deposition from the facility. The NCA is located in the 12 o'clock position of the thruster so any erosion due to placement of the NCA in the high-energy beam would be seen in the bottom of the images taken, which appears pristine. Normalized measurements from the erosion images, shown in figure 57, confirm no observed erosion of the NCA keeper orifice diameter or cathode orifice diameter.

Accelerator grid apertures

Images of the CRA, 16 cm or MRA, and ORA are obtained periodically throughout the wear test. Erosion of the CRA showed formation of the pit and groove pattern within the first few hundred hours of operation. Figure 12 illustrates the pre-test and latest CRA images. The hexagonal pit and groove pattern is clearly evident as well as the buildup of back-sputtered carbon around the inside of the pit and groove pattern. The CRA erosion measurements, normalized to the beginning-of-test value, are shown in figure 58. Within the uncertainty of the measurement, no variation of CRA cusp has been detected. There has been a 10 percent increase in the CRA downstream diameter after 9,000 hr of operation, which appears to be approaching a linear slope for this operating condition.

The lack of CRA cusp enlargement in the NEXT LDT is in sharp contrast to the observed trend during the NSTAR ELT. Figure 13 shows the normalized CRA cusp measurements for the NEXT LDT and NSTAR ELT illustrating the difference in CRA cusp erosion (ref. 33). The NEXT PM ion optics beginning-of-test erosion is described by aperture downstream chamfering compared to the NSTAR ion optics aperture cusp and downstream erosion. The difference is due to the more highly focused beamlets

at the NEXT full-power condition. Aperture cusp enlargement leads to increased neutral losses, more variable performance parameters, and possible electron back-streaming.

Images of the MRA illustrate net-deposition around this aperture located 16 cm from the grid centerline (i.e., 2 cm from outer apertures). The pre-test and most recent MRA images are shown in figure 14. Pit and groove erosion is trumped by the back-sputtered carbon deposition. No noticeable change in the MRA cusp diameter has been observed, but the downstream diameter has increased and the erosion rate begun to approach a final value after 6,000 hr of operation. Over the duration of the test, the MRA downstream diameter has increased by 8 percent. There has been no indication of cross-over impingement at the mid-radius location.

Images are taken of the 4 outer-radius apertures, 3 of them have similar configurations with 3-of-6 active neighboring apertures. The fourth ORA has 4 active neighboring apertures with 2 closed. The geometries of the ORA apertures are shown in figure 15. The formation of “ear” erosion in the first few hundred hours of operation has been closely monitored to ensure that this erosion does not lead to significant structural degradation of accelerator grid. The “ear” formation results from over-focusing at the outer radius apertures due to the low discharge plasma densities at the large radii of the optics and the shape of the distorted sheath due to lack of neighboring apertures (ref. 50).

Erosion images of the ORA’s are shown in figures 16 and 17. Only one of the 3-neighbor outer-radius apertures is shown due to similarity as all exhibit the same aperture geometry. The 4-neighbor ORA exhibits a slightly different “ear” erosion pattern. In both cases, the low-density and non-symmetric aperture sheath causes ions originating near missing aperture locations to crossover resulting in directly impingement (refs. 50 and 51). Erosion measurements of the ORA’s are shown in figure 60 illustrating negligible cusp erosion within the measurement uncertainty. The “ear” dimensions, defined in the schematic in figure 15, increased rapidly during the first 1,000 hr of operation and reached a constant growth rate after ~2,000 hr. The 3-neighbor ORA’s and 4-neighbor ORA “ear” dimensions have increased by ~40 and 25 percent, respectively.

Accelerator pit and groove erosion

Erosion of the accelerator grid downstream surface by charge-exchange ions results in pit and groove formation and leads to the eventual structural failure of the accelerator grid. During the NSTAR ELT, the accelerator pit depth was measured during the first 10,000 hr of operation with laser profilometer. Problems with the diagnostic precluded accurate measurement of pit depth afterwards. The post-test inspection revealed pit erosion through the thickness of the accelerator grid (ref. 33). NEXT 2,000 hr wear test post-test profilometer measurements indicated the deepest pit depth around the center radius aperture was 12 percent of the accelerator grid thickness. The center radius and deepest groove depths were 5 and 6 percent, respectively, with the maximum groove depth at the 1/4-radius location. To measure the NEXT LDT groove depth, a diagnostic technique was developed using the existing accelerator grid CCD cameras to determine the pit and groove depth at the center-radius location. The diagnostic technique is described in the additional analyses section.

There are no discernable pits at the center-radius location in the erosion images such as figure 12, however, grooves are clearly visible. The groove depth measurement has been obtained since 7,600 hr of operation ~ every 700 to 800 hr. The preliminary results are presented in figure 18 for 3 groove depth measurements utilizing 2 different CCD cameras and 3 different references. The groove depth is measured by: the distance between groove and accelerator grid downstream surface using the CRA camera, the distance between the groove and the screen downstream surface using the grid-gap camera, and the distance between the groove and the accelerator grid upstream surface using the grid-gap camera. The measured increase in groove depth is consistent with the results from the NEXT 2,000 hr wear test. The groove erosion is expected to be linear with time, assuming operation at a fixed operating condition. Extrapolation of the groove depth trend indicates wear-through at a throughput of 700 to 800 kg. The wear-through of the accelerator grid does not in itself represent a failure of the thruster. Application of a model developed to predict the effect of backsputtering on grid erosion, with a 3 $\mu\text{m}/\text{hr}$ back-sputter rate

and center-radius aperture ion impingement current density, estimates a maximum of 10 percent reduction in erosion near the beam center, where pit and groove erosion rate is highest, during the test due to back-sputtered carbon (ref. 52). From NSTAR data on DS1, the impingement accelerator current in space was ~25 percent less than the NSTAR data obtained during pre-flight measurements at 1.50 A beam current and 25.56 sccm total flow rate in a test facility operating with a background pressure level of 3.5×10^{-6} Torr (refs. 3, 33, and 38). Because the NEXT LDT is operating at comparable operating background pressures, it is expected that the NEXT in space accelerator current would be ~25 percent less than those measured and therefore grid erosion measurements obtained in this facility are conservative.

Ion optics cold grid-gap

The gap between the screen and accelerator grids decreased over the course of both the NSTAR ELT and 8,200 hr wear test (refs. 16 and 18). Pre and post-test grid-gap measurements from the NSTAR ELT and 8,200 hr wear test indicated 30 and 12 percent reduction in the cold grid-gap, respectively (ref. 33). It is not known how the grid gap varied over the course of the tests because no data is available. Analysis of electron backstreaming data indicates the grid-gap change probably occurred slowly over the duration of the test (ref. 33). Decreases in grid-gap increase the electric field between the grids resulting in reduced electron backstreaming margin and increased arcing between the grids. A CCD camera images the centerline cold grid-gap periodically throughout the LDT. Within the measurement uncertainty, there has been no observed change in the NEXT LDT cold grid-gap, shown in figure 19.

Additional Analyses

Accelerator Groove Depth Measurement Technique

Measurement of the CRA groove depth was accomplished with a technique utilizing the in-situ CRA and grid-gap CCD cameras. The cameras are mounted on a single-axis translation stage allowing motion in the radial direction. The diagnostic technique developed utilizes the CCD camera field-of-depth to estimate the groove depth by moving the camera in the radial direction until various features are in/out of focus. From the domed accelerator geometry axial distances between components of the screen and accelerator grids can be determined, as shown in figure 20. The technique was checked against the known cold grid-gap and accelerator cusp-to-downstream thickness. The technique demonstrated repeatable and representative results.

The groove depth is measured at the center-radius location, which has 5 percent lower beam current density than the off-center peak that is ~2 in. from centerline. While not measuring groove erosion at the peak beam current density, the erosion on centerline is expected to be representative of the peak erosion underestimating erosion depth by roughly 5 percent. The measurement images accelerator grid apertures within 2 in. of centerline for this diagnostic technique. Sample images used in the technique illustrating the various features in and out of focus are shown in figure 21. The preliminary groove depth data agree well with the results of the NEXT 2,000 hr wear test and life assessment predictions. As groove data is collected over a longer duration, higher confidence can be gained in this technique.

Doubles-to-Singles Sensitivity

The production of multiply charged ions is primarily a function of discharge propellant efficiency and discharge voltage. An investigation by Sovey confirmed the latter over a range of discharge voltages for various propellants, shown in figure 23 (ref. 36). The NEXT LDT doubles-to-singles ratio was measured during the pretest performance characterization as a function of discharge propellant efficiency for fixed discharge voltages at 3.52 A beam current, shown in figure 22. The sensitivity of the doubles-to-singles ratio observed during the NSTAR ELT indicated at full power that an increase in discharge voltage of ~1 V can lead to an increase in the doubles-to-singles ratio of 4 percentage points (refs. 33). The less than

2 percentage point increase in the NEXT LDT doubles-to-singles ratio at full-input-power, illustrated in figure 55, results from the increase in discharge voltage from 23.1 to 23.9 V over the first 10,100 hr of thruster operation. This increase is consistent with the observed NSTAR ELT doubles-to-singles current ratio sensitivity to discharge voltage.

At the full-power condition, there is a clear correlation between the doubles-to-singles ratio and discharge voltage as illustrated in figure 24. The linear relationship between the doubles-to-singles ratio and discharge voltage is demonstrated for fixed discharge propellant efficiency over the range of discharge voltage variation from 23 to 24 V.

Discharge Current and Discharge Losses Predictions

Increasing discharge currents during wear tests are required to maintain discharge chamber ion production, and hence beam current, as neutrals are increasingly lost due to erosion of accelerator grid apertures. A model was developed to predict the increase in discharge current based upon the observed and predicted trends in aperture erosion. Erosion of the majority of the accelerator apertures, the exception being the ORA's, has been confined to changes in the downstream surface, with little if any increase in cusp diameters. The CRA, MRA, and ORA erosion measurements are used to estimate the increase in neutral transmission factor during the test to determine the required increase in discharge current to maintain the beam current.

The results from an investigation by Reynolds and Richley of the diffusion through slots and tubes in the free-molecular flow regime are used in this calculation shown in figure 25 (ref. 53). The aperture erosion geometries are modeled as cones, shown in figure 25, to permit use of Reynolds' results. The cone angles, determined from erosion images, are multiplied by a factor less than one to account for the actual erosion geometry, which is better represented as a chamfering of the downstream surface. The change in neutral total transmission factor is calculated from the measured change in accelerator CRA, MRA, and ORA downstream dimensions.

The NEXT PM accelerator apertures have a non-dimensional length of 1.1, which is equal to the accelerator grid thickness divided by the aperture radius. The discharge current required to maintain the beam current is then scaled according to the change in total transmission factor for the CRA, MRA, and ORA images. The contributions of each of the aforementioned scaling parameters are weighted by the estimated number of apertures over the entire accelerator grid that would be more closely represented by that aperture and the accuracy of the model compared to measured data. For the CRA and MRA total transmission factors, the wall half-angle is multiplied by 0.3 to account for the differences in erosion geometry and the resulting value used in figure 25. The measured ORA downstream areas are converted to an effective downstream diameter then multiplied by 0.45 prior to use in figure 25. The bulk discharge neutral density is inversely proportional to total neutral transmission. In the model an inversely proportional increase in primary electron number density, hence discharge current, is assumed from the decrease in neutral density to maintain a given beam current. Discharge current can then be calculated according to equation (1) where numeric factors are the weighting factors.

$$\text{Discharge Current} = \text{Discharge Current}_{t=0} \left(\frac{0.3P_t^{\text{CRA}} + 0.4P_t^{\text{MRA}} + 0.3P_t^{\text{ORA}}}{0.3P_t^{\text{CRA}}|_{t=0} + 0.4P_t^{\text{MRA}}|_{t=0} + 0.3P_t^{\text{ORA}}|_{t=0}} \right) \quad (1)$$

The discharge current calculated utilizing the above model with the measured aperture geometries as inputs is plotted as a function of time with the actual discharge current runtime data in figure 26. The model accurately predicts the discharge current based upon the eroded geometries. During the first few hundred hours, the increasing discharge current is dominated by the ear formation in the outer-radius apertures seen as a sharp increase. Also shown in figure 26 is a predicted trend for discharge current based upon curve fitting to the observed erosion of the apertures and extrapolation of the accelerator aperture erosion trends. The predicted discharge current after 450 kg is expected to increase by ~1.2 A

using this technique. Discharge current during the NSTAR ELT increased by ~1 A, an equivalent of ~7 percent, over the test duration though the exact value is unknown due to the lack of a pretest characterization at full-power prior to the wear test initiation at an intermediate power level (ref. 33).

Discharge losses can also be estimated based upon the discharge current model results and the observed discharge voltage trends. Discharge losses are calculated according to equation (2) with discharge current calculated based upon accelerator erosion geometries. The calculated discharge losses based upon this method agree well with the actual measured discharge losses, as can be seen in figure 27. Extrapolating the observed discharge voltage trend and incorporating the predicted discharge current model, it is possible to predict future discharge losses. The future prediction for discharge losses at the full-power point assumes that there is no a significant change in observed trends in the future. Throttling of the thruster to alternate power levels will alter erosion patterns and rates.

$$\text{Discharge Loss} = \text{Discharge Loss}_{t=0} \times \left(\frac{V_{DC}}{V_{DC}|_{t=0}} \right) \left(\frac{J_{DC}}{J_{DC}|_{t=0}} \right) \quad (2)$$

Substituting curve fits and extrapolation for the CRA, MRA, ORA, and discharge voltages, into equations (1) and (2), the predicted discharge loss curve, shown in figure 27, is produced. The estimate for discharge losses, if operated at full-power, are expected to reach 136 W/A after 450 kg of xenon processed, an increase of 14 W/A or ~10 percent of the beginning of test value. The behavior of the discharge voltage over long test durations is uncertain, based upon NSTAR wear test results (ref. 33). During the NSTAR ELT, discharge voltage actually decreased by ~1 V at the same time as the keeper-to-common short appeared (ref. 18). Thus, the NSTAR ELT discharge voltage data is not useful in predicting NEXT trends. After 450 kg the NEXT LDT full-power discharge losses are predicted to increase by 14 W/A compared to the NSTAR ELT increase of 22 W/A after 210 kg propellant processed.

Conclusion

The results of the NEXT Long-Duration Test (LDT) as of June 21, 2007, are presented. At the full-input-power of 6.9 kW, the NEXT EM3 thruster has accumulated 10,100 hr of operation, processed 207 kg of xenon, and demonstrated a total impulse of 8.5×10^6 N-s. *The NEXT LDT total impulse is the highest ever demonstrated by an ion thruster in the history of space propulsion.* The thruster performance is characterized periodically over the throttling range of 0.5 to 6.9 kW, with calculated thrust of 26 to 237 mN, respectively. At low and high input power, the thruster specific impulses and thrust efficiencies are 1360 s, 0.319, and 4170 s, 0.706, respectively. Overall thruster performance, which includes thrust, input power, specific impulse, and thrust efficiency, has remained constant with no signs of degradation. Impingement-limited total voltages, electron backstreaming limits, and screen grid ion transparencies have been steady to date. Discharge losses have increased by 8 W/A due to increasing discharge voltage and current, with the latter due to increased neutral loss resulting from accelerator aperture erosion. Neutralizer keeper and coupling voltages have been steady, with no indication of performance degradation.

Beam current density profiles have not changed significantly during the wear test. Beam divergence, the half-angle containing 95 percent of the beam current, increased from 24.5° to 25.3° over the duration of the test due to accelerator aperture erosion at large radii. Images of the discharge and neutralizer cathode assemblies indicate, within the accuracy of the measurement, that the discharge and neutralizer cathode keeper and cathode orifice diameters have not changed. Erosion of center-radius and mid-radius accelerator apertures is restricted to chamfering of the downstream diameter with negligible increase to the cusp diameters. Erosion of the outer-radius apertures demonstrate “ear” formation due to over-focusing at these low plasma density regions resulting in impingement on the accelerator grid that reached a constant rate after ~2,000 hr of operation. There has been no change in the ion optics’ cold grid-gap over the duration of the test. Groove formation was evident during the first few hundred hours of

operation at the center radius location. The estimate for the groove depth at the center radius aperture is 20 to 30 percent of the accelerator grid thickness after 207 kg of xenon throughput. Extrapolation of groove depth at the center radius at full-power to complete wear through is consistent with the NEXT service life model value of 700 to 800 kg.

All thruster performance and erosion trends indicate that the NEXT thruster will achieve the qualification propellant throughput of 450 kg. Many of the high-risk NSTAR wear test erosion mechanisms and potential failure modes have been reduced or mitigated in the NEXT thruster, confirmed by in-situ measurements.

References

1. Rayman, M.D., "The Successful Conclusion of the Deep Space 1 Mission: Important Results without a Flashy Title," *World Space Congress/53rd International Astronautical Congress*, Houston, TX, October 10–19, 2002.
2. Lee, M., Weidner, R.J. and Soderblom, L.A., "Deep Space 1 Mission and Observation of Comet Borrelly," *45th IEEE International Midwest Symposium on Circuits and Systems*, Tulsa, OK, August 4, 2002.
3. Polk, J.E., et al., "Performance of the NSTAR Ion Propulsion System on the Deep Space One Mission," AIAA–2001–0965, *39th AIAA Aerospace Sciences Meeting and Exhibit Joint Propulsion Conference*, Reno, NV, January 8–11, 2001.
4. Oleson, S., Gefert, L., Patterson, M.J., Schreiber, J. and Benson, S., "Outer Planet Exploration With Advanced Radioisotope Electric Propulsion," Pasadena, CA, October 15–19, 2001.
5. Oleson, S., et al., "Mission Advantages of NEXT: NASA's Evolutionary Xenon Thruster," AIAA–2002–3969, *38th AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Indianapolis, IN, July 7–10, 2002.
6. Cupples, M., Coverstone, V. and Woo, B., "Application of Solar Electric Propulsion to a Comet Surface Sample Return Mission," AIAA–2004–3804, *40th AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Fort Lauderdale, FL, July 11–14, 2004.
7. Oh, D., Benson, S., Witzberger, K. and Cupples, M., "Deep Space Mission Applications for NEXT: NASA's Evolutionary Xenon Thruster," AIAA–2004–3806, *40th AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Fort Lauderdale, FL, July 11–14, 2004.
8. Herman, D.A., Soulas, G.C. and Patterson, M.J., "Performance Evaluation of the Prototype-Model NEXT Ion Thruster," AIAA–2007–5212, *43rd AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Cincinnati, OH, July 8–11, 2007.
9. Pinero, L.R., Todd, P. and Hopson, M., "Integration and Qualification of the NEXT Power Processing Unit," AIAA–2007–5214, *43rd AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Cincinnati, OH, July 8–11, 2007.
10. Patterson, M.J. and Benson, S.W., "NEXT Ion Propulsion System Development Status and Capabilities," *Proceedings, 2007 NASA Science Technology Conference*, College Park, MD, June 19–21, 2007.
11. Snyder, J.S., et al., "Vibration Test of a Breadboard Gimbal for the NEXT Ion Engine," AIAA–2006–4665, *42nd AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Sacramento, CA, July 9–12, 2006.
12. Aadland, R.S., Monheiser, J.A., D.E., Wilson, F.C., and Benson, S.W., "Development Status of the NEXT Propellant Management System," AIAA–2004–3974, *40th AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Fort Lauderdale, FL, July 11–14, 2004.
13. Hoskins, W.A., et al., "Development of a Prototype Model Ion Thruster for the NEXT System," AIAA–2004–4111, *40th AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Fort Lauderdale, FL, July 11–14, 2004.

14. Monheiser, J., Aadland, R.S. and Wilson, F., "Development of a Ground Based Digital Control Interface Unit (DCIU) for the NEXT Propulsion System," AIAA-2004-4112, *40th AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Fort Lauderdale, FL, July 11-14, 2004.
15. Van Noord, J.L. and Williams, G.J., "Lifetime Assessment of the NEXT Ion Thruster," AIAA-2007-5274, *43rd AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Cincinnati, OH, July 8-11, 2007.
16. Polk, J.E., et al., "An Overview of the Results from an 8200 Hour Wear Test of the NSTAR Ion Thruster," AIAA-1999-2446, *35th AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Los Angeles, CA, June 20-24, 1999.
17. Polk, J.E., et al., "A 1000 Hour Wear Test of the NASA NSTAR Ion Thruster," AIAA-1996-2784, *32nd AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Lake Buena Vista, FL, July 1-3, 1996.
18. Sengupta, A., et al., "An Overview of the Results from the 30,000 Hr Life Test of Deep Space 1 Flight Spare Engine," AIAA-2004-3608, *40th AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Fort Lauderdale, FL, July 11-14, 2004.
19. Doerner, R.P., Whyte, D.G. and Goebel, D.M., "Sputtering Yield Measurements During Low Energy Xenon Plasma Bombardment," *Journal of Applied Physics*, vol. 93, no. 9, pp. 5816-5823, May 1.
20. Sovey, J., Dever, J.A. and Power, J.L., "Retention of Sputtered Molybdenum on Ion Engine Discharge Chamber Surfaces," IEPC Paper 2001-086, *27th International Electric Propulsion Conference*, Pasadena, CA, October 15-19, 2001.
21. Patterson, M.J. and Benson, S.W., "NEXT Ion Propulsion System Development Status and Performance," AIAA-2007-5199, *43rd AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Cincinnati, OH, July 8-11, 2007.
22. Soulas, G.C., Patterson, M.J., Van Noord, J.L. and Herman, D.A., "NEXT Ion Thruster Performance Dispersion Analyses," AIAA-2007-5213, *43rd AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Cincinnati, OH, July 8-11, 2007.
23. Soulas, G.C., Domonkos, M.T. and Patterson, M.J., "Performance Evaluation of the NEXT Ion Engine," AIAA-2003-5278, *39th AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Huntsville, AL, July 20-23, 2003.
24. Patterson, M.J., Foster, J.E., Haag, T.W., Rawlin, V.K. and Soulas, G.C., "NEXT: NASA's Evolutionary Xenon Thruster," AIAA-2002-3832, *38th AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Indianapolis, IN, July 7-10, 2002.
25. Frandina, M.M., Arrington, L.A., Soulas, G.C., Hickman, T.A. and Patterson, M., "Status of the NEXT Ion Thruster Long Duration Test," AIAA-2005-4065, *41st AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Tucson, AZ, July 10-13, 2005.
26. Kamhawi, H., Soulas, G.C. and Patterson, M., "NEXT Ion Engine 2000 Hour Wear Test Plume and Erosion Results," AIAA-2004-3792, *40th AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Fort Lauderdale, FL, July 11-14, 2004.
27. Hickman, T.A., Arrington, L.A., Frandina, M.M. and Soulas, G.C., "Overview of the Diagnostics for the NEXT Long Duration Test," AIAA-2005-4064, *41st AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Tucson, AZ, July 10-13, 2005.
28. Soulas, G.C., Kamhawi, H., Patterson, M.J., Britton, M.A. and Frandina, M.M., "NEXT Ion Engine 2000 Hour Wear Test Results," AIAA-2004-3791, *40th AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Fort Lauderdale, FL, July 11-14, 2004.
29. Pinero, L.R., Patterson, M.J. and Satterwhite, V.E., "Power Console Development for NASA's Electric Propulsion Outreach Program," IEPC Paper 93-250, *23rd International Electric Propulsion Conference*, Seattle, WA, September 13-16, 1993.
30. Snyder, J.S., Kamhawi, H., Patterson, M.J. and Britton, M., "Single-String Integration Test Measurements of the NEXT Ion Engine Plume," AIAA-2004-3790 and NASA/TM-2005-213196, *40th AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Fort Lauderdale, FL, July 11-14, 2004.

31. Brophy, J.R., et al., "The Ion Propulsion System for Dawn," AIAA-2003-4542, *39th AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Huntsville, AL, July 20-23, 2003.
32. Brophy, J.R., et al., "Status of the Dawn Ion Propulsion System," AIAA-2004-3433, *40th AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Fort Lauderdale, FL, July 11-14, 2004.
33. Sengupta, A., et al., "The 30,000-Hour Extended-Life Test of the Deep Space 1 Flight Spare Ion Thruster," NASA/TP-2004-213391, The Jet Propulsion Laboratory and NASA Glenn Research Center, Pasadena, March, 2005.
34. Myers, R.M., "Proceedings of the Nuclear Electric Propulsion Workshop, Volume 1: Introductory Material and Thruster Concepts, Section: "MPD Thruster Technology"," JPL D-9512 vol. 1, June 19-22, 1990.
35. Patterson, M.J., Haag, T.W. and Hovan, S.A., "Performance of the NASA 30 cm Ion thruster," IEPC Paper 93-108, *23rd International Electric Propulsion Conference*, Seattle, WA, Sept. 13-16, 1993.
36. Sovey, J.S., "Improved Ion Containment Using a Ring-Cusp Ion Thruster," *Journal of Spacecraft and Rockets*, vol. 21, no. 5, pp. 488-495, September - October 1984.
37. Stueber, T. and Soulas, G.C., "Electrostatic Ion Thruster Diagnostic Uncertainty Analysis," NASA TP-2007-214665, (to be published).
38. Anderson, J.R., et al., "Results of an On-going Long Duration Ground Test of the DS1 Flight Spare Ion Engine," AIAA-1999-2857, *35th AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Los Angeles, CA, June 20-24, 1999.
39. Anderson, J.R., Goodfellow, K., Polk, J.E., Rawlin, V.K. and Sovey, J.S., "Performance Characteristics of the NSTAR Ion Thruster During an On-Going Long Duration Ground Test," 2000 IEEE Aerospace Conference Proceedings, vol. 4, pp. 123-148, March 2000.
40. Soulas, G.C., Domonkos, M.T., Kamhawi, H. and Gardner, M.M., "Status of the NEXT Ion Engine Wear Test," AIAA-2003-4863, *39th AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Huntsville, AL, July 20-23, 2003.
41. Soulas, G.C., Foster, J.E. and Patterson, M.J., "Performance of Titanium Optics on a NASA 30 cm Ion Thruster," AIAA-2000-3814, *36th AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Huntsville, AL, July 16-19, 2000.
42. Hanna, J., et al., "Carbon Film Deposition and Flaking Studies in Ion Thruster Environments," AIAA-2005-3524, *41st AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Tucson, AZ, July 10-13, 2005.
43. Shigeki, T., Yasuyuki, S., Reizo, K., Shigeki, H., and Shigeru, U., "Humidity Dependence of Microwear Characteristics of Amorphous Carbon Films on Silicon Substrates," *International Journal on the Science and Technology of Friction, Lubrication, and Wear*, vol. 254, pp. 1042-1049, 2003.
44. Klages, C.P., et al., "Deposition and Properties of Carbon-Based Amorphous Protective Coatings," *Surface and Coating Technology*, vol. 80, no. 1-2, pp. 121-128, March 1, 1996.
45. Soulas, G.C., "Performance Evaluation of Titanium Ion Optics for the NSA 30 cm Ion Thruster," IEPC-01-092, *27th International Electric Propulsion Conference*, Pasadena, CA, October 15-19, 2001.
46. Soulas, G.C., Haag, T.W. and Patterson, M.J., "Performance Evaluation of 40cm Ion Optics for the NEXT Ion Engine," AIAA-2002-3834, *38th AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Indianapolis, IN, July 7-10, 2002.
47. Christensen, J.A., et al., "Design and Fabrication of a Flight Model 2.3 kW Ion Thruster for the Deep Space 1 Mission," AIAA-1998-3327, *34th AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, 1998.
48. Sengupta, A., Brophy, J.R. and Goodfellow, K.D., "Status of the Extended Life Test of the Deep Space 1 Flight Spare Ion Engine after 30,352 Hours of Operation," AIAA-2003-4558, *39th AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Huntsville, AL, July 20-23, 2003.

49. Doerner, R.P. and Goebel, D.M., "Sputtering Yields of Ion Thruster Grid and Cathode Materials During Very Low Xenon Plasma Bombardment," AIAA-2003-4561, *39th Joint Propulsion Conference*, Huntsville, AL, July 20-23 2003.
50. Malone, S.P., "Investigation of NEXT Ion Optics Erosion Processes using Computational Modeling," 2005-0356DV, NASA Glenn Research Center, Cleveland, OH, October 20, 2005.
51. Malone, S.P. and Soulas, G.C., "Computational Ion Optics Design Evaluations," AIAA-2004-3784, *40th AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Fort Lauderdale, FL, July 11-14, 2004.
52. Polk, J.E., Duchemin, O.B., Ho, C. and Koel, B.E., "The Effect of Carbon Deposition on Accelerator Grid Wear Rates in Ion Engine Ground Testing," AIAA-2000-3662, *36th AIAA/ASME/SAE/ASEE Joint Propulsion Conference and Exhibit*, Huntsville, AL, July 17-19, 2000.
53. Reynolds, T.W. and Richley, E., "Free-Molecule Flow and Surface Diffusion Through Slots and Tubes—A Summary," NASA TR-R-255, April, 1967.

TABLE 1.—NEXT LDT FULL-INPUT-POWER OPERATING CONDITION

P_{IN} , kW [†]	J_B , A	V_B , V	V_A , V	m_M , sccm	m_C , sccm	m_N , sccm	J_{NK} , A
6.83	3.52	1800	-210	49.6	4.87	4.01	3.00

[†]Nominal value

TABLE 2.—NSTAR ELT THRUSTER HIGH RISK ITEMS, EROSION PHENOMENA,
AND NEXT THRUSTER MITIGATION STRATEGIES

NSTAR ELT issue	NEXT mitigation strategy	NEXT LDT observations
Discharge cathode short (after 5850 hr, 56 kg) and keeper faceplate completely eroded away exposing cathode.	Cathode keeper material changed to graphite to reduce sputter yield and erosion rate.	No short or keeper orifice diameter erosion observed. Substantial margin on keeper orifice plate erosion from 2,000 hr wear test (ref. 28).
Pit and groove erosion through the accelerator grid evident in post-test analysis.	Thicker accelerator grid to achieve longer life.	Groove erosion measurements indicate 700-800 kg propellant throughput before eroded through.
Ion optics cold grid-gap 30 percent decrease over test duration.	Redesigned ion optics mounting ring to reduce thermal stresses.	In-situ cold grid-gap measurement constant over test duration to date.
Electron backstreaming limit reached throttle value for full-power operation after ~10, 900 hr (92 kg).	Thick accelerator grid optics utilized, highly focused beamlets at full-power reduce cusp erosion, and grid-gap decrease mitigated.	Electron backstreaming limit has been constant for all operating conditions over test duration. Negligible cusp erosion to date.
Evidence of arc-tracking in low-voltage propellant isolator.	Low-voltage isolator eliminated. Isolation maintained by the high-voltage propellant isolator.	High impedance measured from anode to discharge cathode to date.

TABLE 3.—NEXT ION THRUSTER THROTTLE TABLE WITH LDT PERFORMANCE OPERATING CONDITIONS
SUBSET SHADED. FULL-POWER WEAR TEST CONDITION IN BOLD

P_{IN} , kW ^a	J_B , A	V_S , V	V_A , V	m_M , sccm	m_C , sccm	m_N , sccm	J_{NK} , A
6.83	3.52	1800	-210	49.6	4.87	4.01	3.00
6.03	3.52	1570	-210	49.6	4.87	4.01	3.00
5.43	3.52	1400	-210	49.6	4.87	4.01	3.00
4.68	3.52	1180	-200	49.6	4.87	4.01	3.00
6.03	3.10	1800	-210	43.5	4.54	4.01	3.00
5.32	3.10	1570	-210	43.5	4.54	4.01	3.00
4.80	3.10	1400	-210	43.5	4.54	4.01	3.00
4.14	3.10	1180	-200	43.5	4.54	4.01	3.00
5.27	2.70	1800	-210	37.6	4.26	3.50	3.00
4.65	2.70	1570	-210	37.6	4.26	3.50	3.00
4.19	2.70	1400	-210	37.6	4.26	3.50	3.00
3.61	2.70	1180	-200	37.6	4.26	3.50	3.00
3.20	2.70	1020	-175	37.6	4.26	3.50	3.00
4.60	2.35	1800	-210	32.4	4.05	3.50	3.00
4.06	2.35	1570	-210	32.4	4.05	3.50	3.00
3.66	2.35	1400	-210	32.4	4.05	3.50	3.00
3.16	2.35	1180	-200	32.4	4.05	3.50	3.00
2.80	2.35	1020	-175	32.4	4.05	3.50	3.00
4.00	2.00	1800	-210	25.8	3.87	2.50	3.00
3.54	2.00	1570	-210	25.8	3.87	2.50	3.00
3.20	2.00	1400	-210	25.8	3.87	2.50	3.00
2.77	2.00	1180	-200	25.8	3.87	2.50	3.00
2.46	2.00	1020	-175	25.8	3.87	2.50	3.00
3.24	1.60	1800	-210	20.0	3.70	2.75	3.00
2.87	1.60	1570	-210	20.0	3.70	2.75	3.00
2.60	1.60	1400	-210	20.0	3.70	2.75	3.00
2.26	1.60	1180	-200	20.0	3.70	2.75	3.00
2.01	1.60	1020	-175	20.0	3.70	2.75	3.00
2.43	1.20	1800	-210	14.2	3.57	3.00	3.00
2.15	1.20	1570	-210	14.2	3.57	3.00	3.00
1.95	1.20	1400	-210	14.2	3.57	3.00	3.00
1.70	1.20	1180	-200	14.2	3.57	3.00	3.00
1.51	1.20	1020	-175	14.2	3.57	3.00	3.00
1.41	1.20	936	-150	14.2	3.57	3.00	3.00
1.31	1.20	850	-125	14.2	3.57	3.00	3.00
1.11	1.20	679	-115	14.2	3.57	3.00	3.00
1.08	1.20	650	-144	14.2	3.57	3.00	3.00
0.777	1.20	400	-394	14.2	3.57	3.00	3.00
0.656	1.20	300	-525	14.2	3.57	3.00	3.00
0.529	1.00	275	-500	12.3	3.52	3.00	3.00

^aNominal values

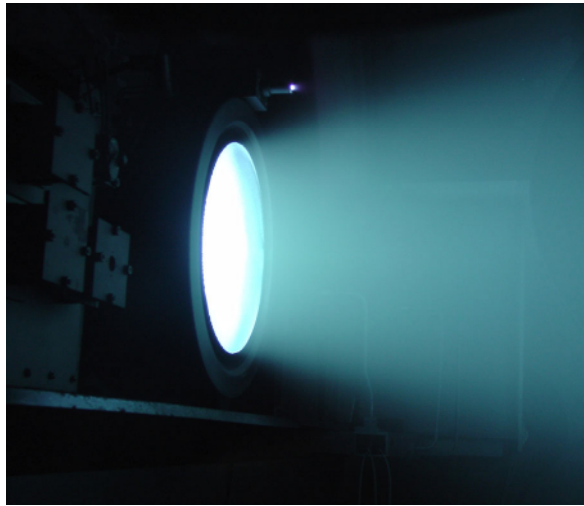


Figure 1.—NEXT EM3 operating at full-power during the long-duration test.

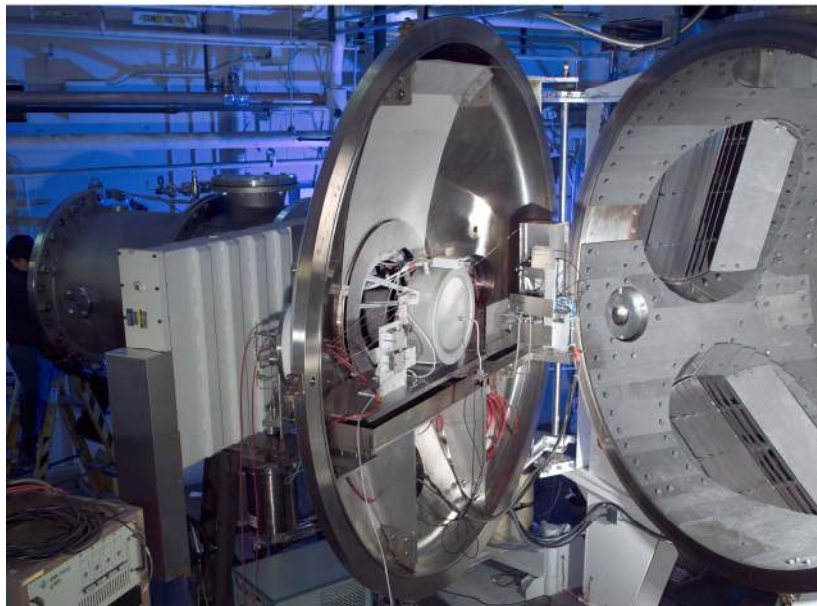


Figure 2.—VF-16 at NASA GRC—end-cap opened. EM3 extends out the emergency bell jar in testing configuration.

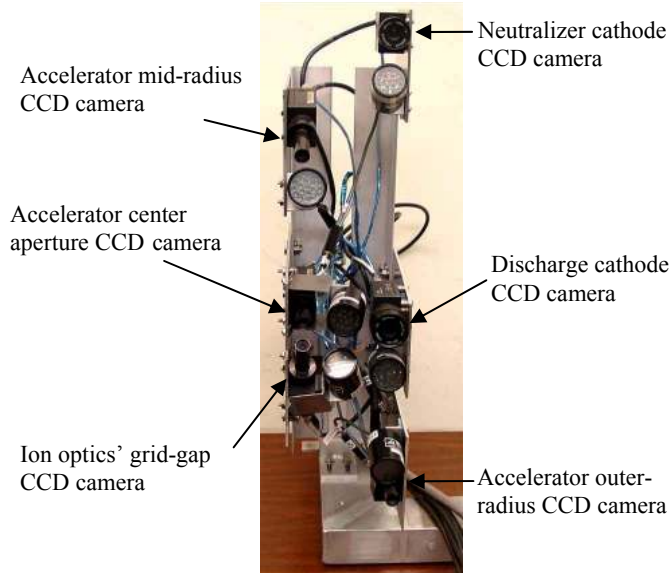


Figure 3.—Erosion cameras mounted to a vertical mast.

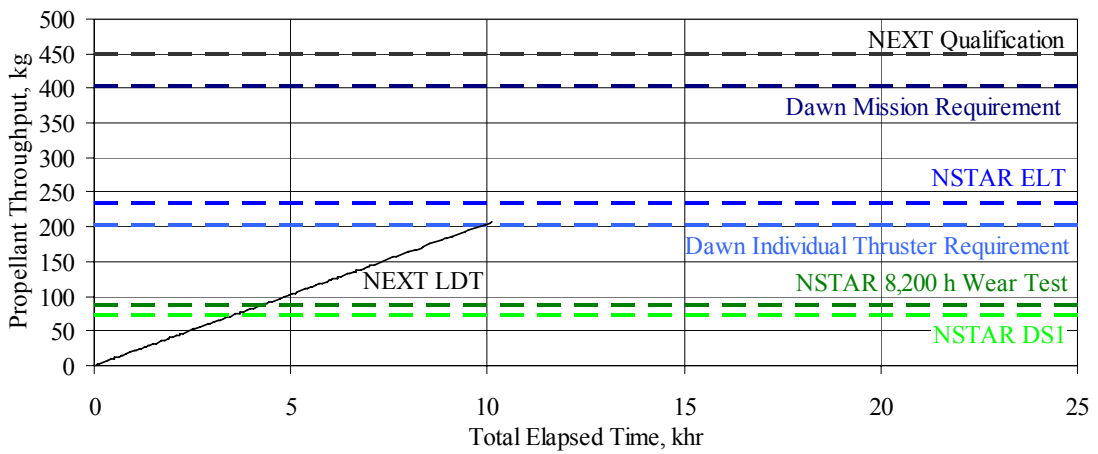


Figure 4.—NEXT LDT propellant throughput data as a function of time with reference milestones.

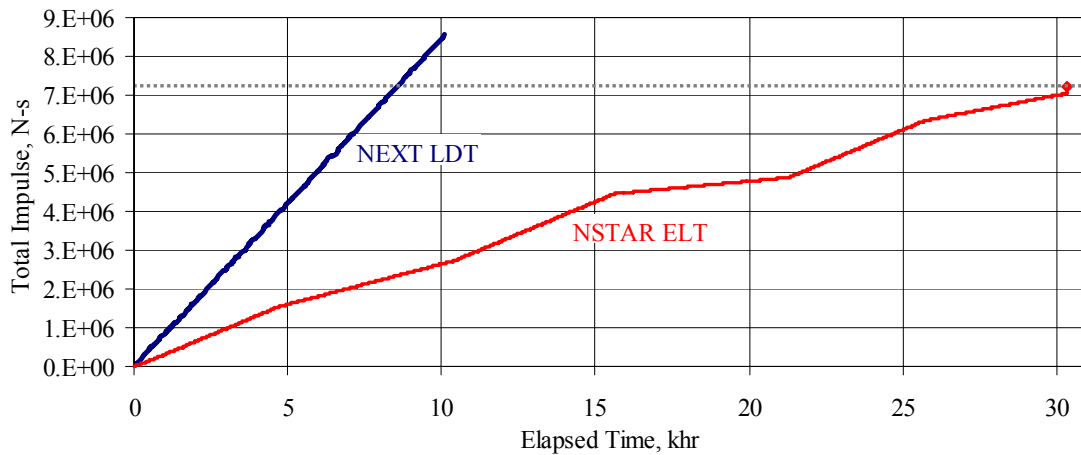


Figure 5.—NEXT LDT and NSTAR ELT total impulse data as a function of time.

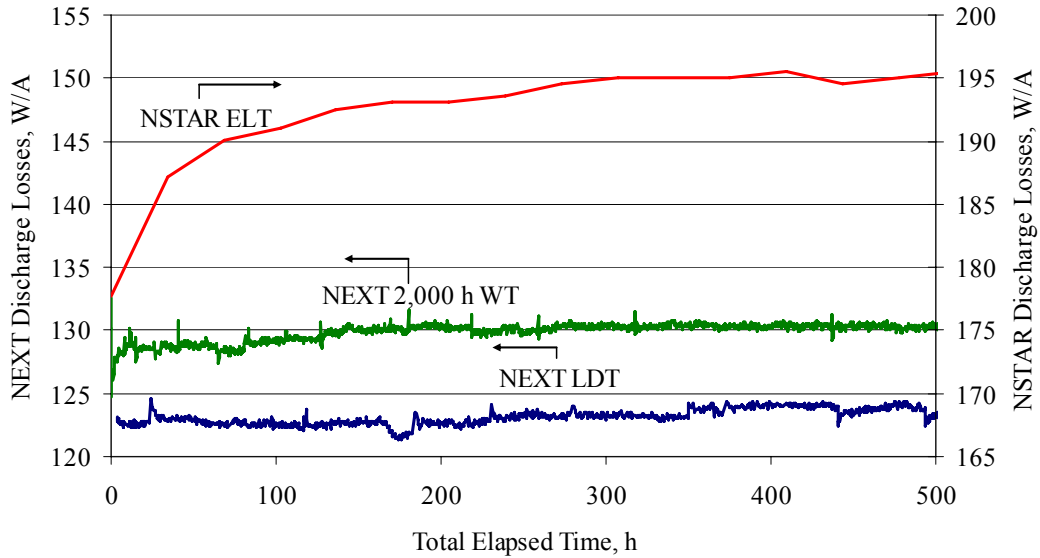


Figure 6.—Discharge losses for the NSTAR ELT, NEXT 2,000 hr wear test, and NEXT LDT (refs. 28, 38, and 39).

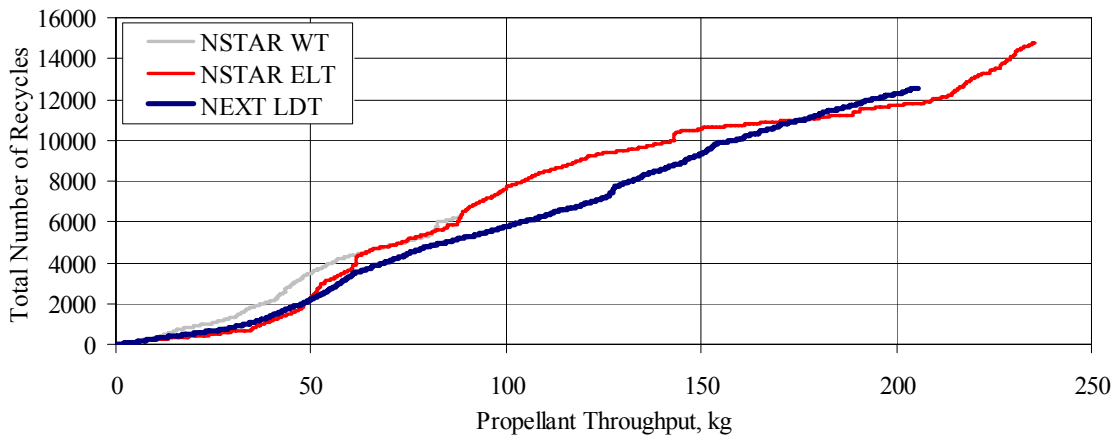


Figure 7.—Total number of recycles as a function of propellant throughput for the NEXT LDT, NSTAR ELT, and NSTAR 8,200 hr wear test (refs. 16 and 33).

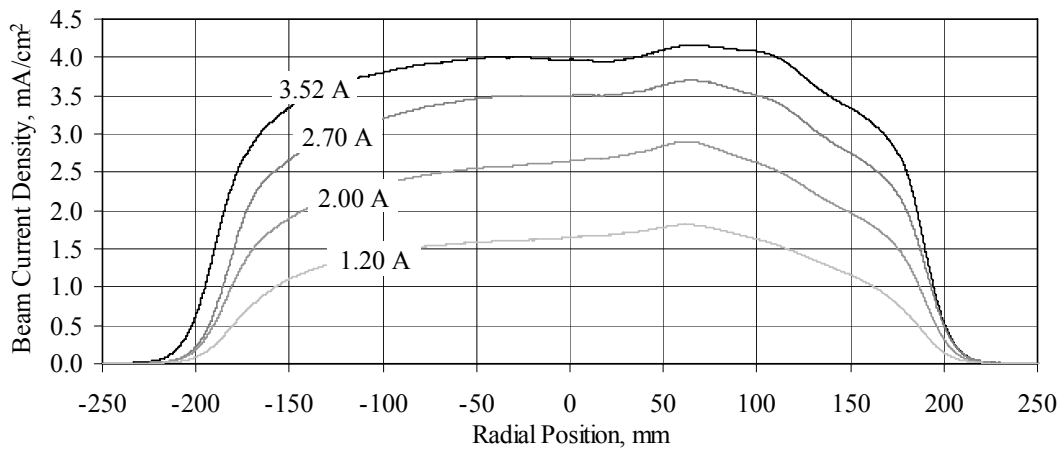


Figure 8.—Sample beam profiles for various beam currents at an axial position of 20 mm for $V_B = 1800$ V.

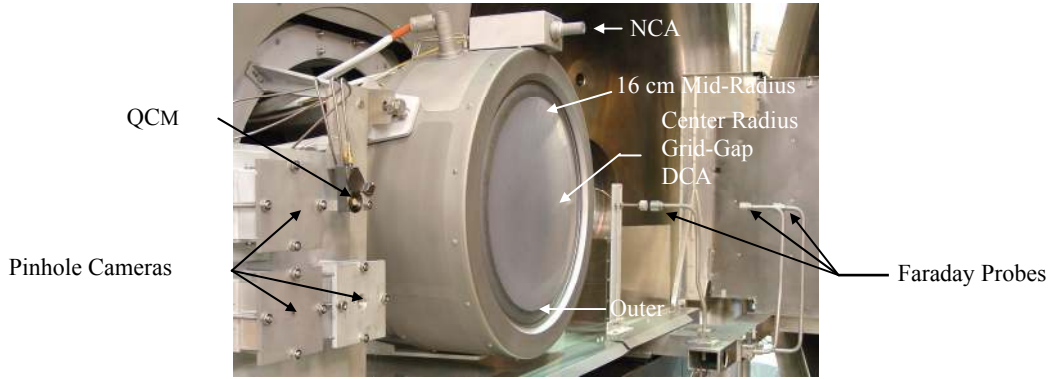


Figure 9.—Pretest setup photograph illustrating QCM, pinhole cameras, CCD image locations, and staggered faraday probes.

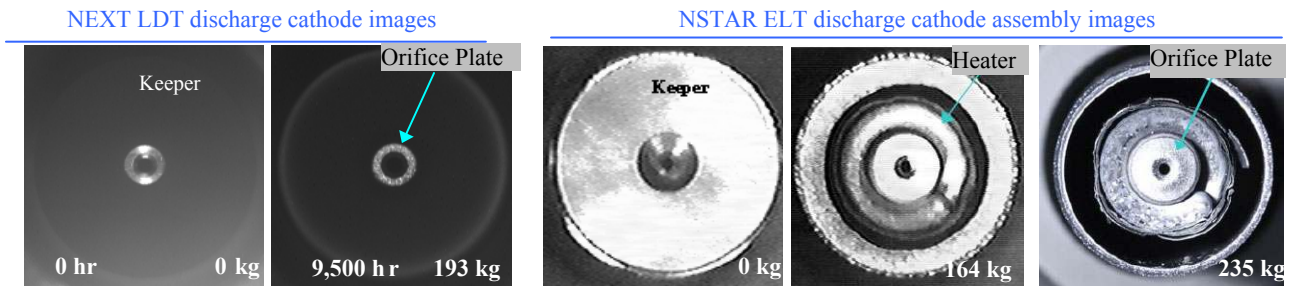


Figure 10.—Discharge cathode assembly images for NEXT LDT (left) and NSTAR ELT (right).

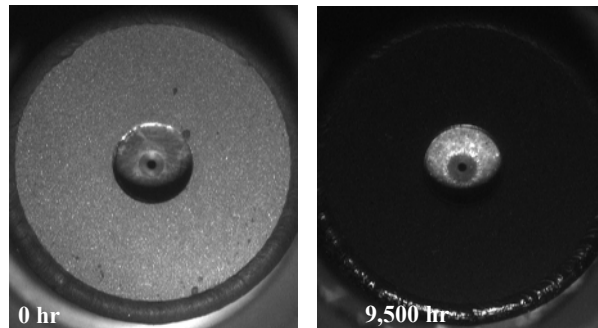


Figure 11.—Neutralizer assembly erosion images.

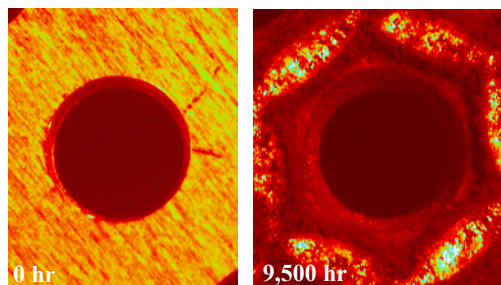


Figure 12.—Accelerator grid center-radius aperture images with an applied red palette.

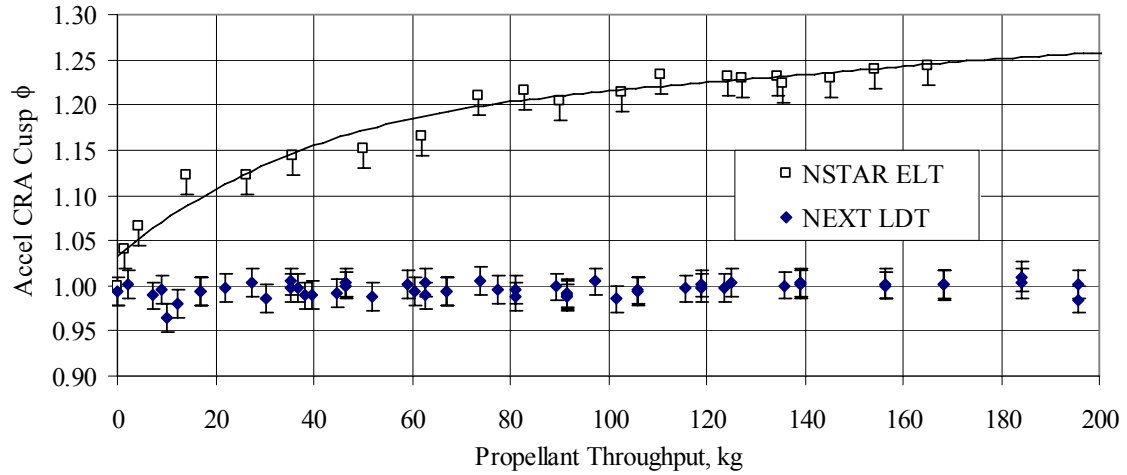


Figure 13.—Accelerator grid center-radius aperture cusp diameter for the NEXT LDT and NSTAR ELT (ref. 33).

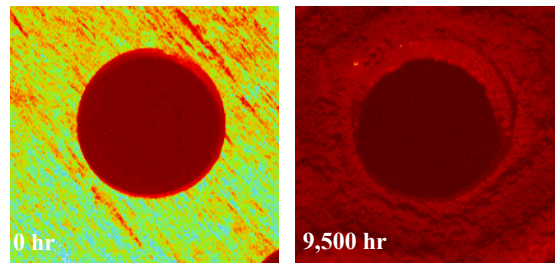


Figure 14.—Accelerator grid mid-radius aperture erosion images with an applied red palette.

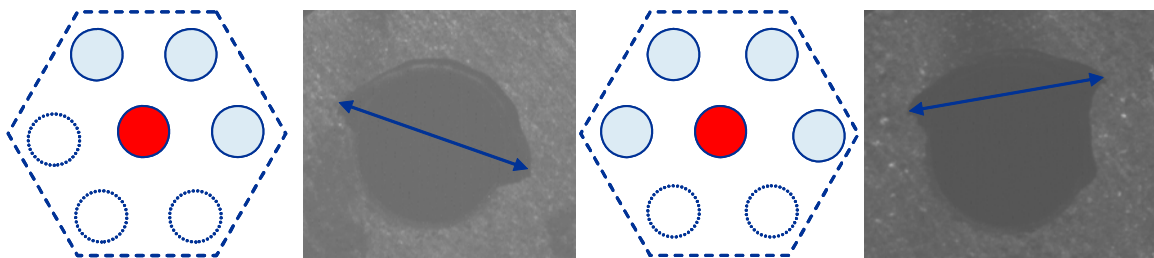


Figure 15.—Accelerator grid outer-radius aperture configurations and “ear” dimensions. The imaged aperture is centered, shaded neighboring apertures are active and open apertures are closed.

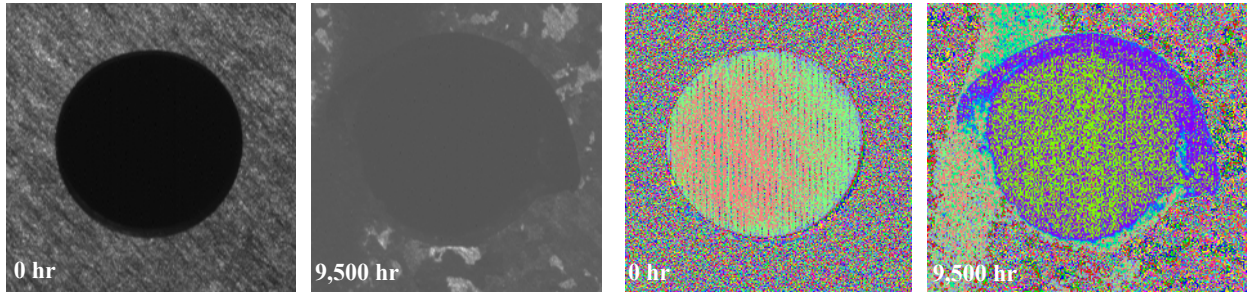


Figure 16.—Accelerator grid 3-neighbor outer-radius aperture images (left) and with applied palette (right).

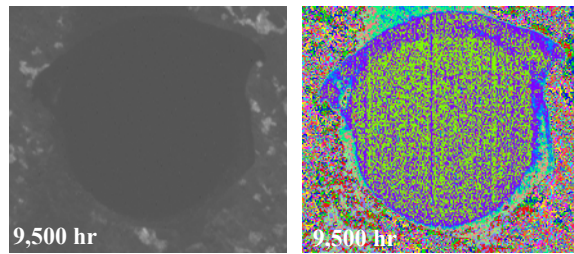


Figure 17.—Accelerator grid 4-neighbor outer-radius aperture image (left) with palette (right).

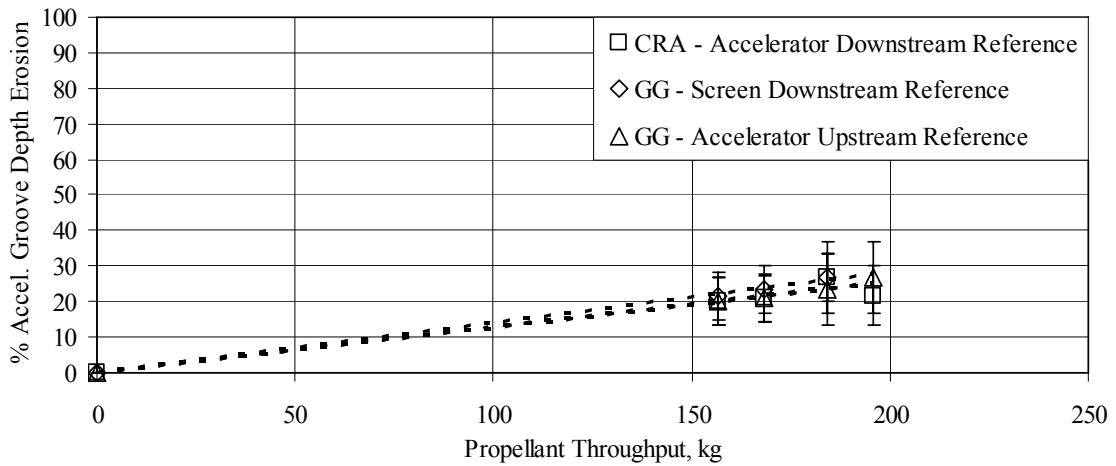


Figure 18.—Accelerator grid center-radius groove erosion depth, percentage of total accelerator grid thickness, as a function of propellant throughput using two CCD's and three different references.

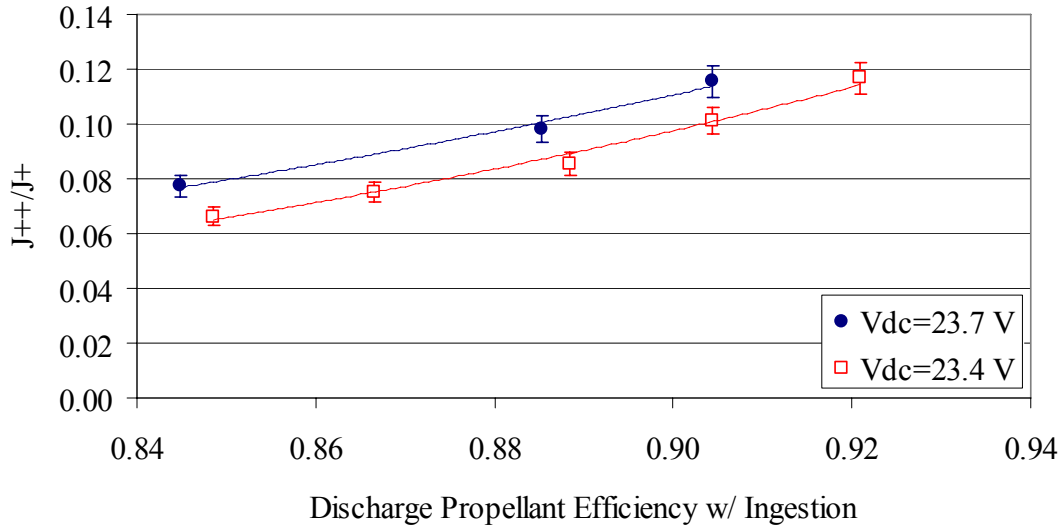


Figure 22.—NEXT LDT doubles-to-singles ratio as a function of discharge propellant efficiency including ingested flow for fixed discharge voltages at 3.52 A beam current.

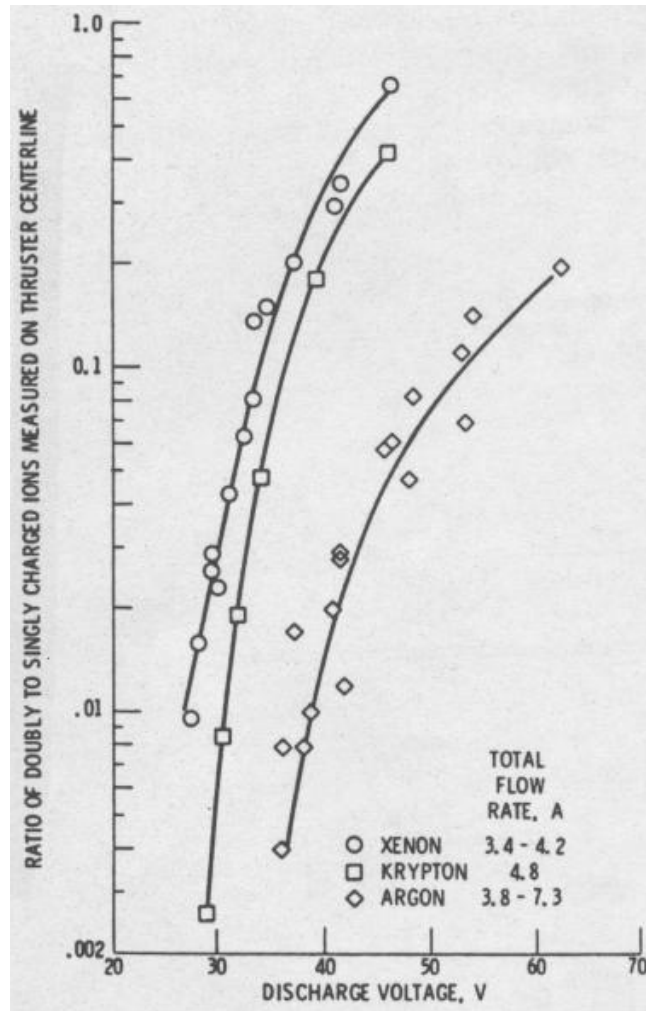


Figure 23.—Doubles-to-singles dependency on discharge voltage (ref. 36).

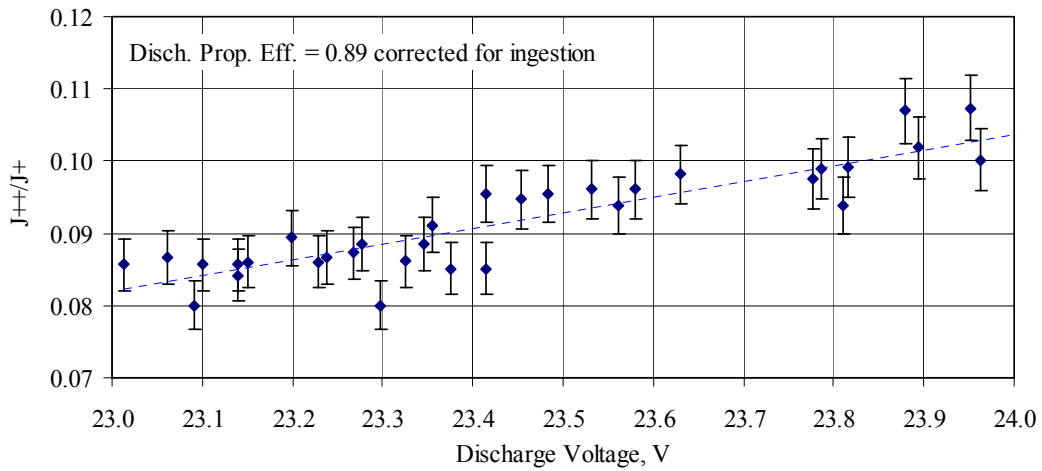


Figure 24.—Sensitivity of doubles-to-singles ratio on discharge voltage at the full-input-power condition.

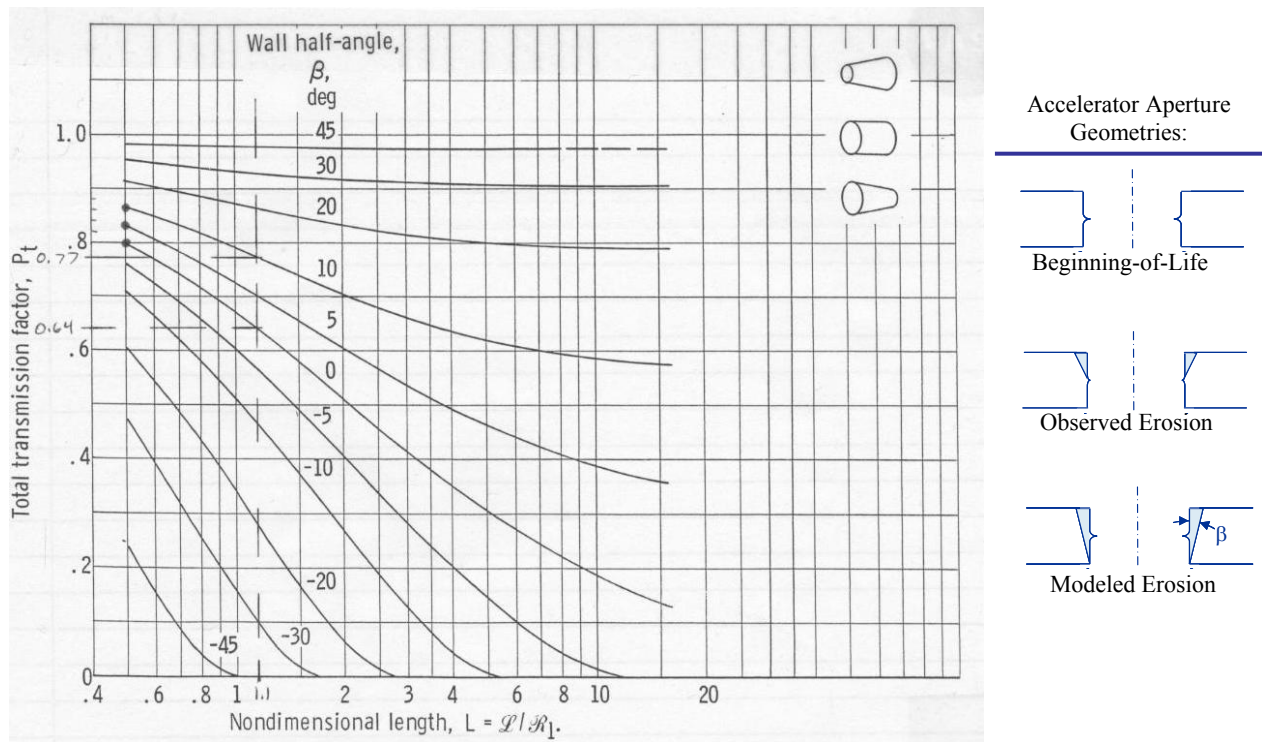


Figure 25.—Neutral transmission factors for various wall half-angles as a function of cone length to radius (left) with the beginning-of-life, observed erosion to date, and modeled aperture geometries (right).

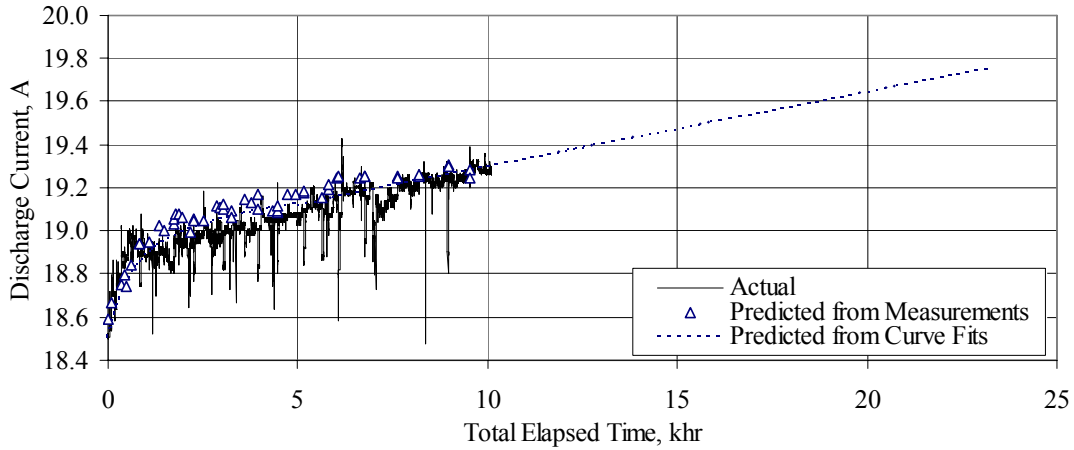


Figure 26.—Calculated discharge current from actual accelerator aperture images, predicted discharge current from aperture erosion curve-fit equations and extrapolations, and actual discharge currents to date.

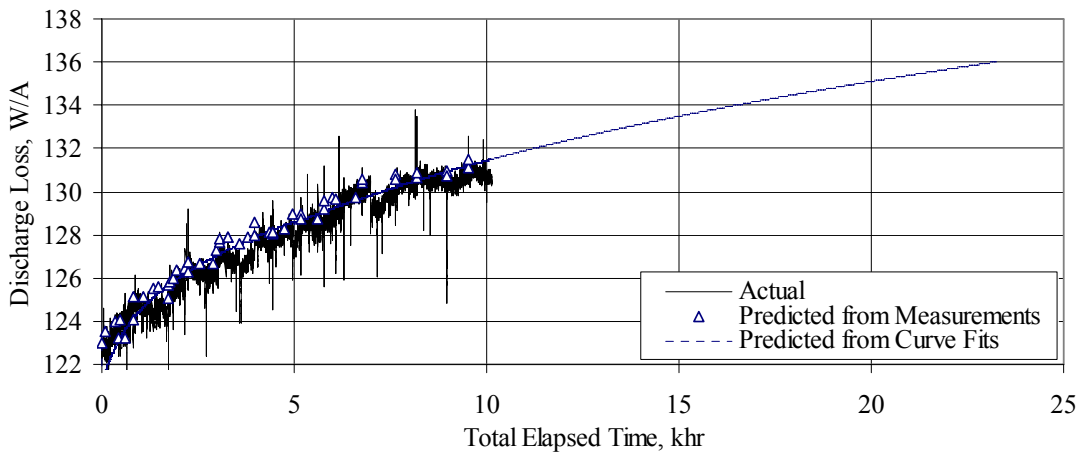


Figure 27.—Calculated discharge losses from actual accelerator aperture images and discharge voltage data, predicted discharge losses from aperture and discharge voltage curve-fit equations and extrapolations, and measured discharge loss data as a function of time to date.

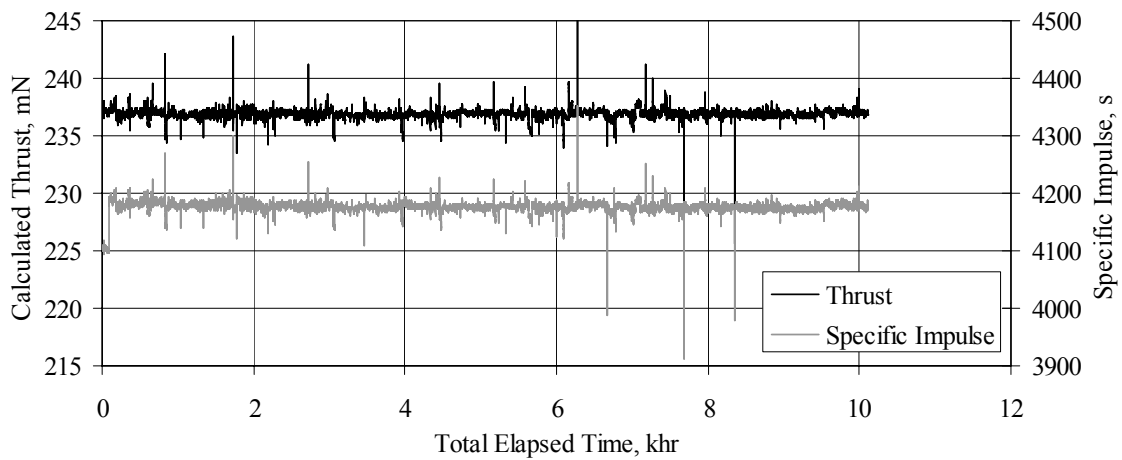


Figure 28.—Full-power calculated thrust and specific impulse runtime data as a function of time.

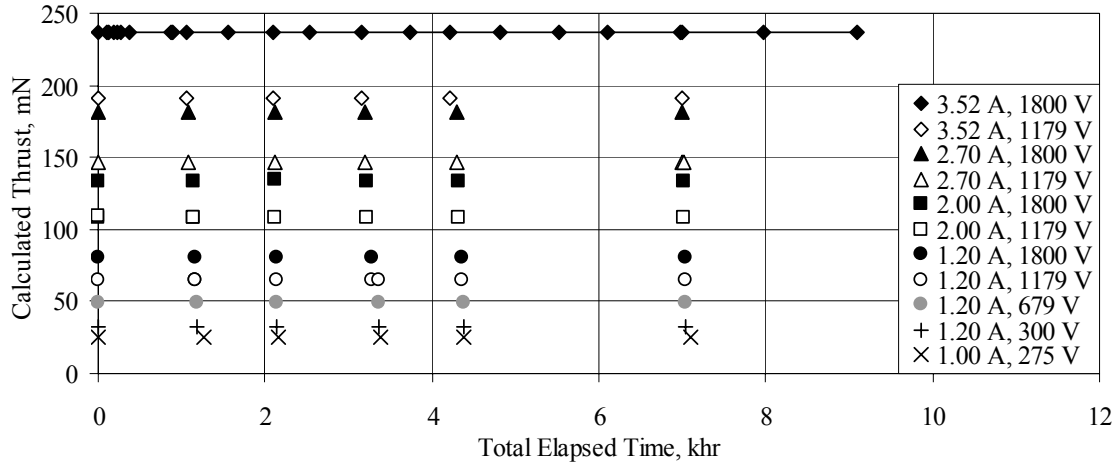


Figure 29.—Performance characterization calculated thrust data as a function of time.

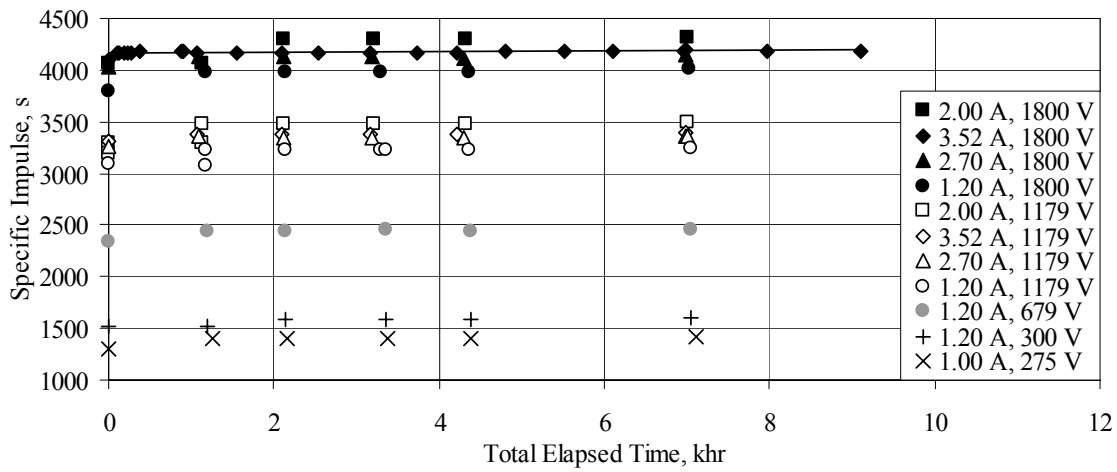


Figure 30.—Performance characterization specific impulse data as a function of time.

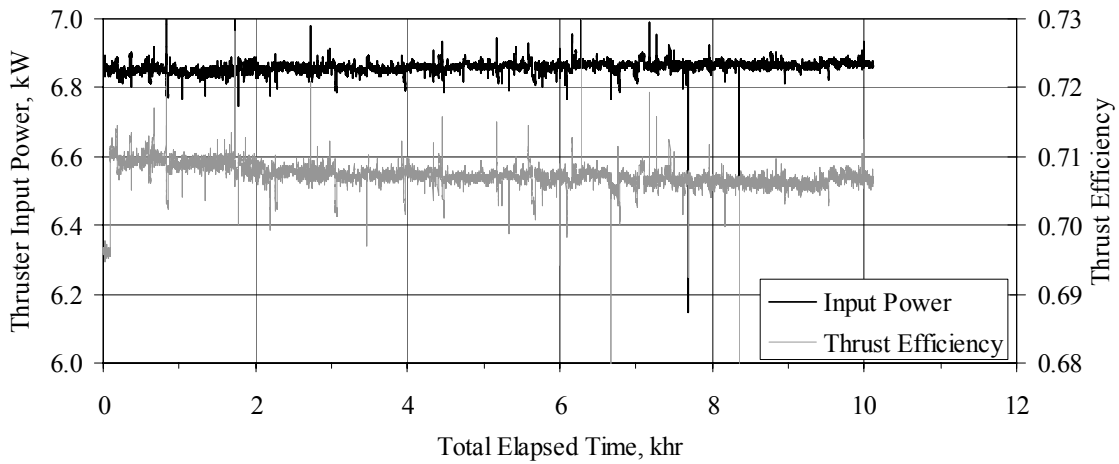


Figure 31.—Full-power thruster input power and thrust efficiency runtime data as a function of time.

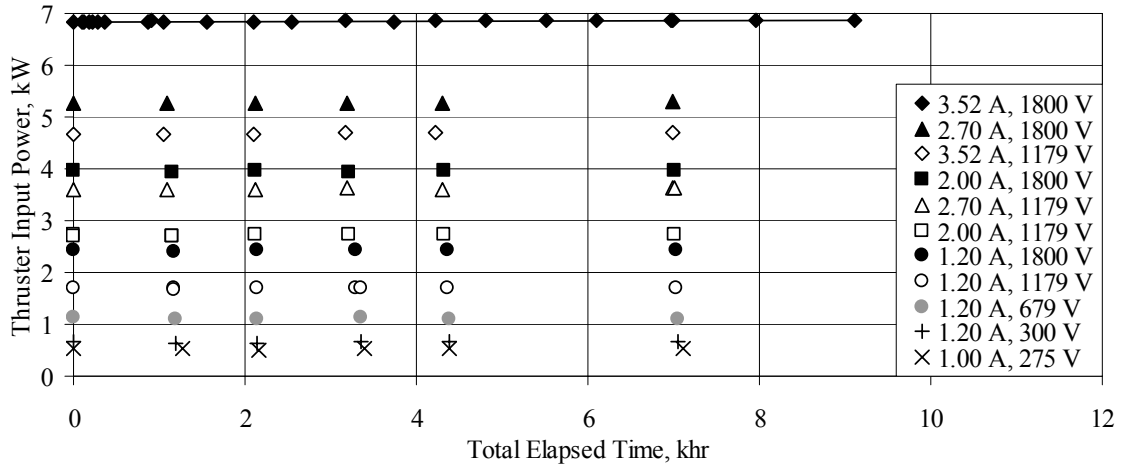


Figure 32.—Performance characterization thruster input power data as a function of time.

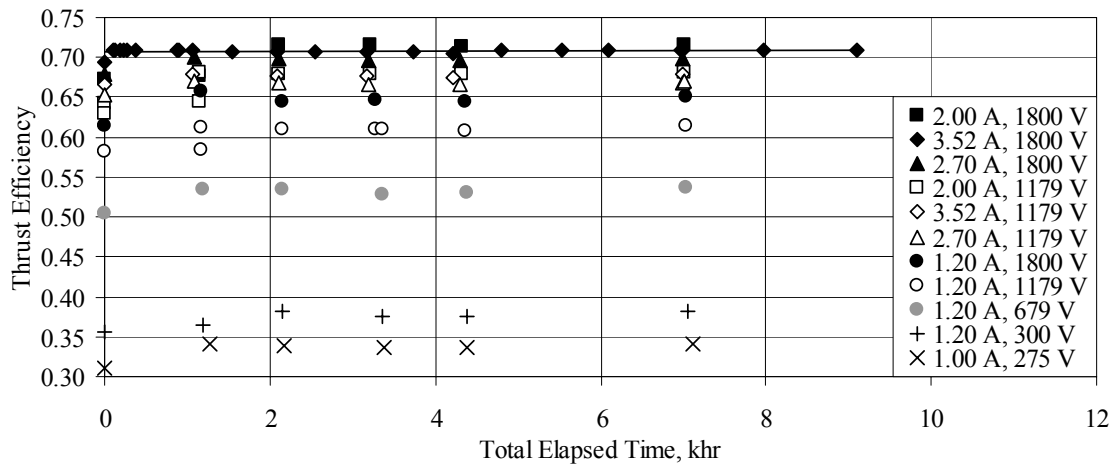


Figure 33.—Performance characterization thrust efficiency data as a function of time.

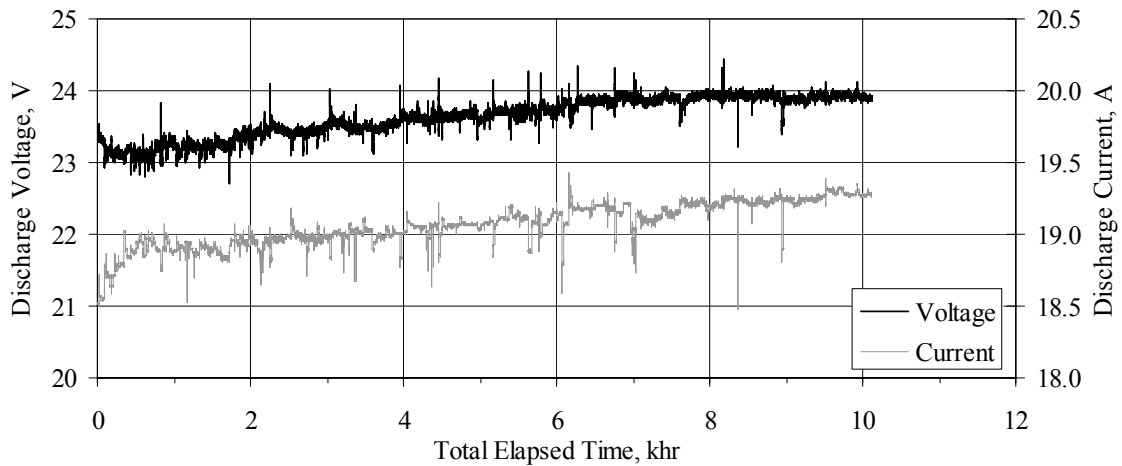


Figure 34.—Full-power discharge voltage and current runtime data as a function of time.

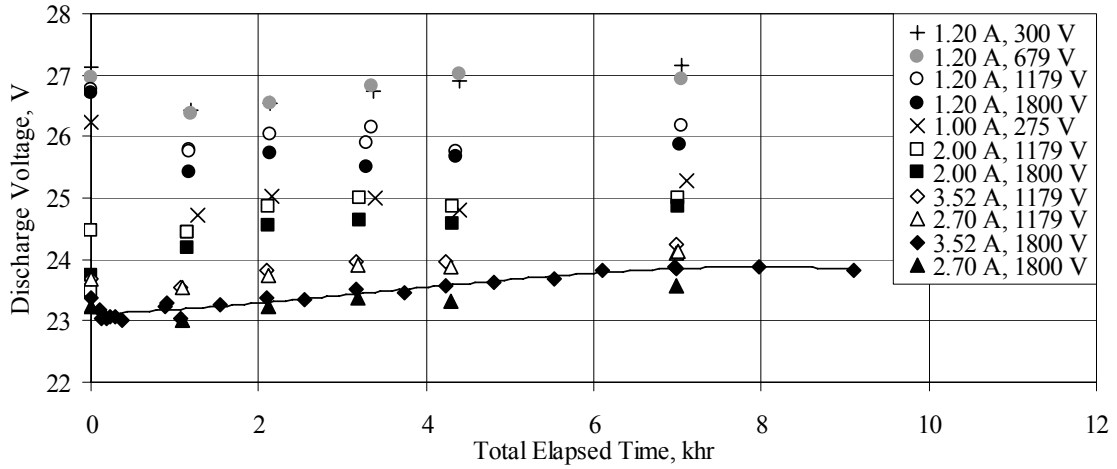


Figure 35.—Performance characterization discharge voltage data as a function of time.

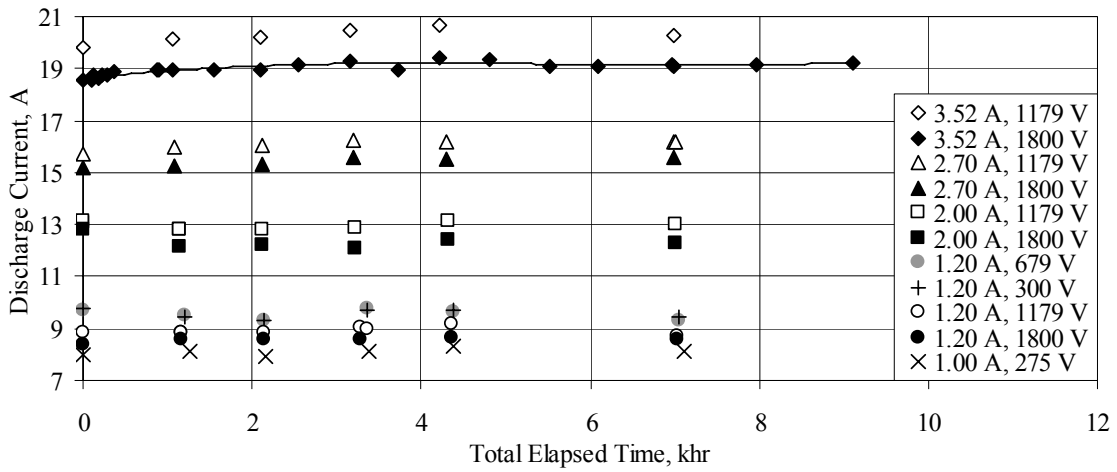


Figure 36.—Performance characterization discharge current data as a function of time.

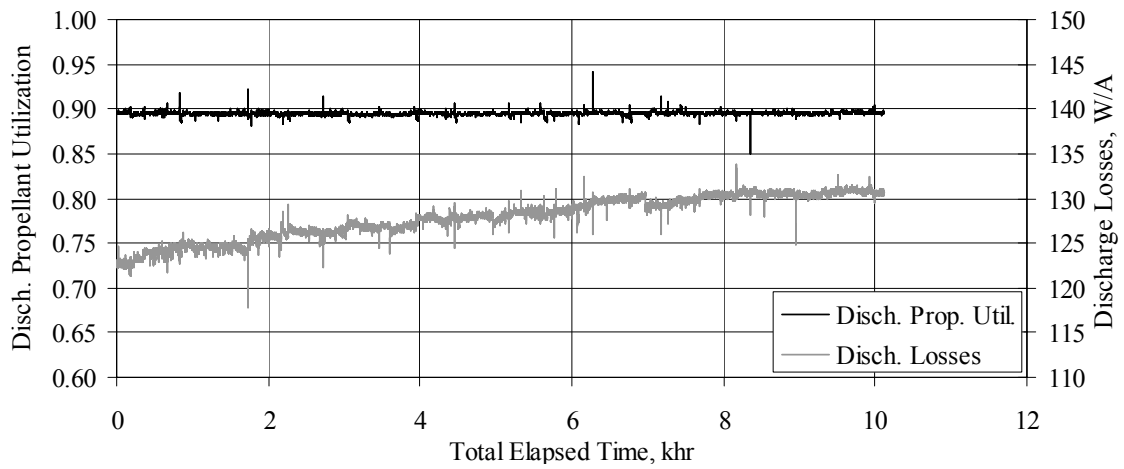


Figure 37.—Full-power thruster discharge propellant utilization efficiency (including mass ingestion, but no doubly-charged ions) and discharge loss runtime data as a function of time.

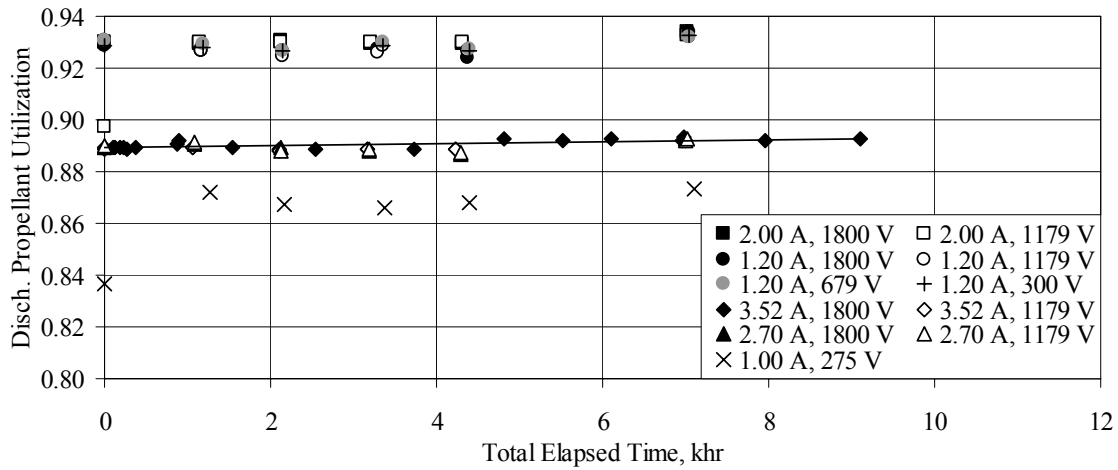


Figure 38.—Performance characterization discharge propellant utilization efficiency (including mass ingestion, but no doubly-charged ions) as a function of time.

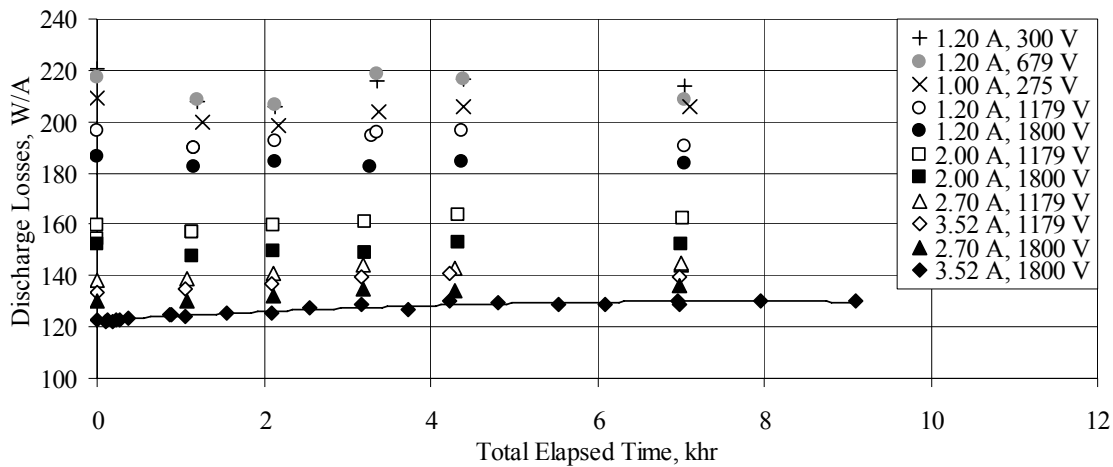


Figure 39.—Performance characterization discharge loss data as a function of time.

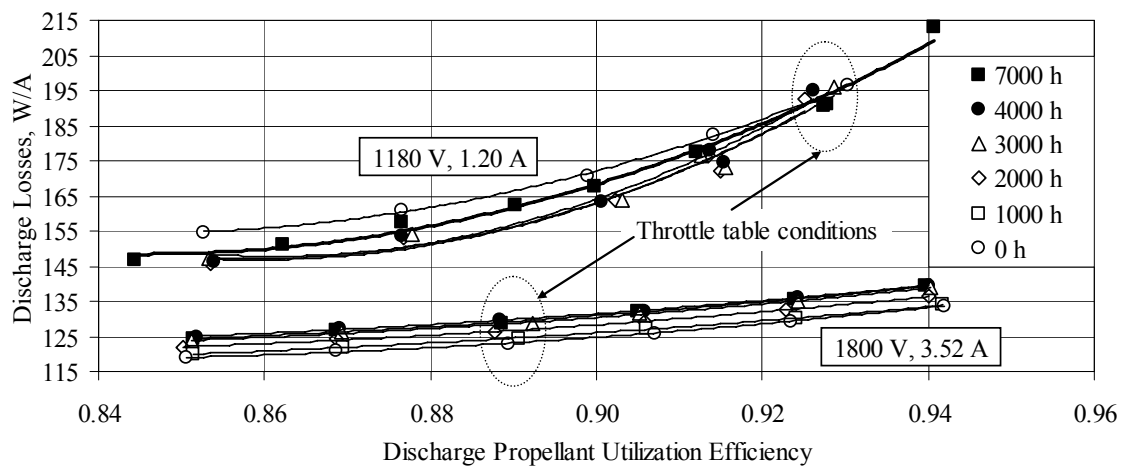


Figure 40.—Full-power discharge losses plotted against discharge propellant utilization efficiency (including mass ingestion) after various test durations for fixed discharge voltage of $23.5\text{ V} \pm 0.5\text{ V}$.

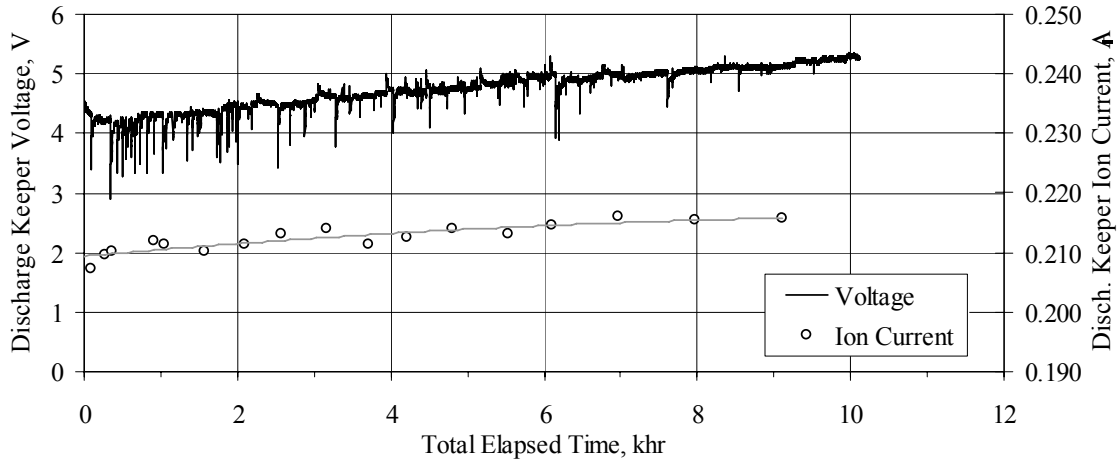


Figure 41.—Full-power discharge keeper voltage and ion saturation current runtime data as a function of time.

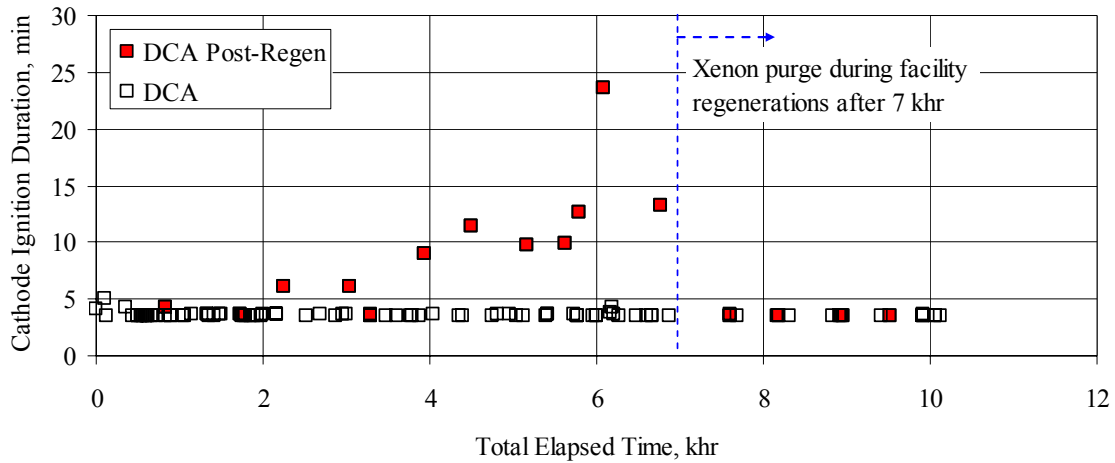


Figure 42.—Thruster discharge cathode ignition data as a function of time.

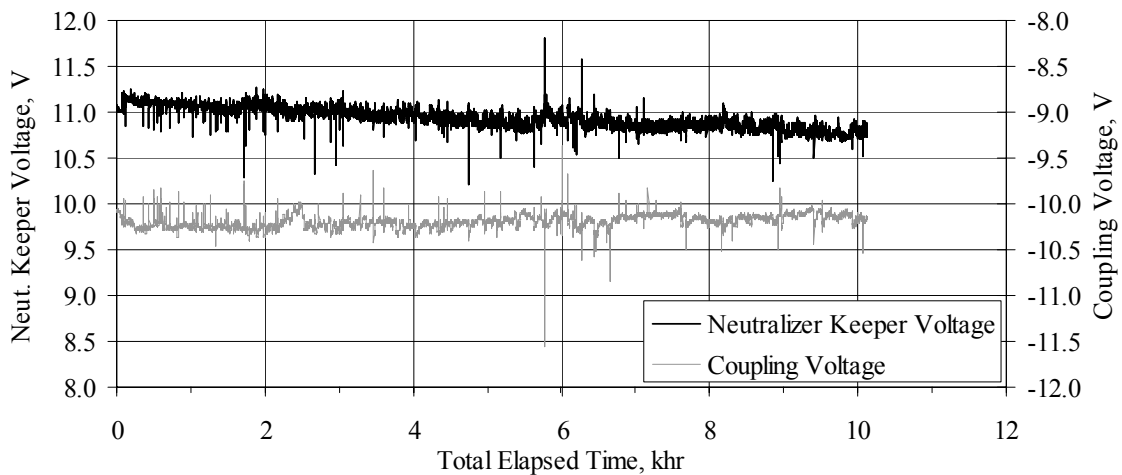


Figure 43.—Full-power neutralizer keeper voltage and coupling voltage runtime data as a function of time.

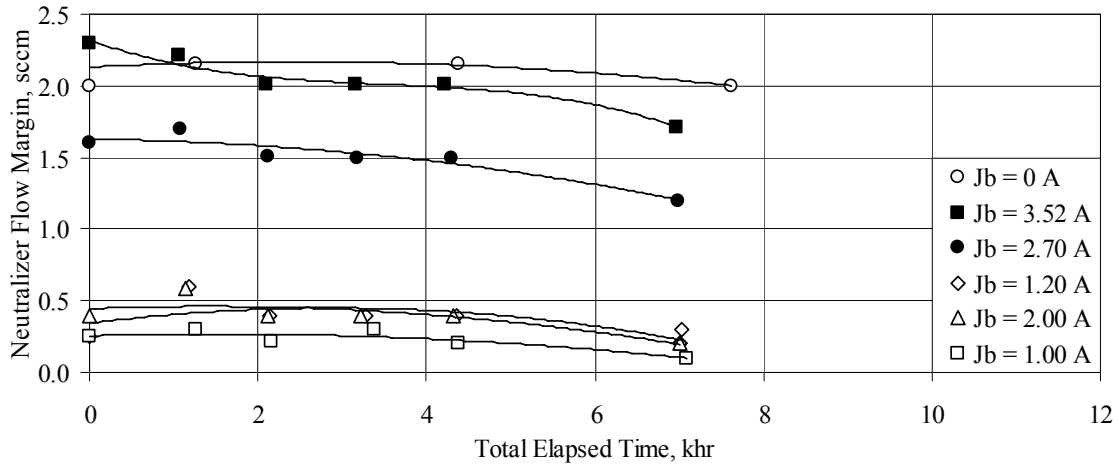


Figure 44.—Neutralizer flow margin, between set flow and transition flow, as a function of time for various beam currents.

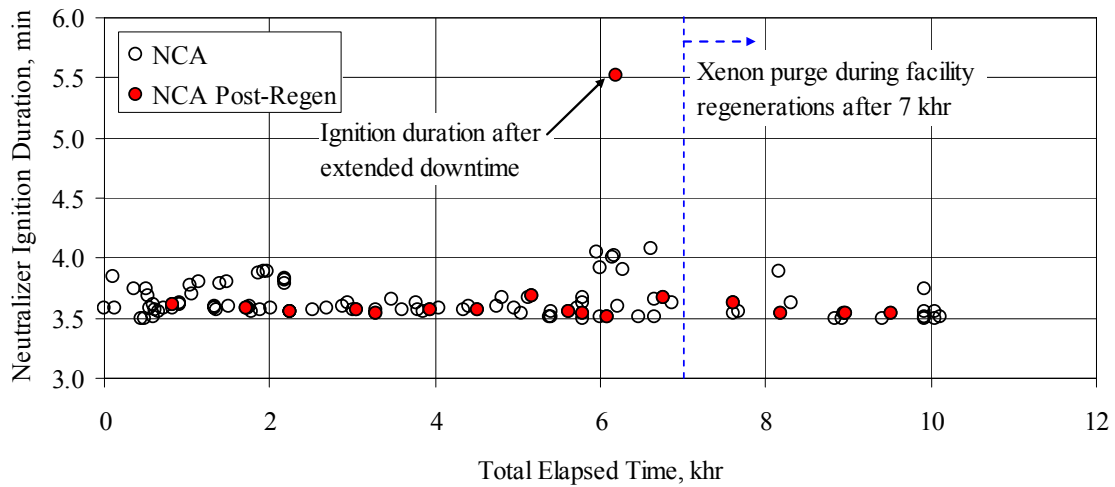


Figure 45.—Thruster neutralizer cathode ignition data as a function of time.

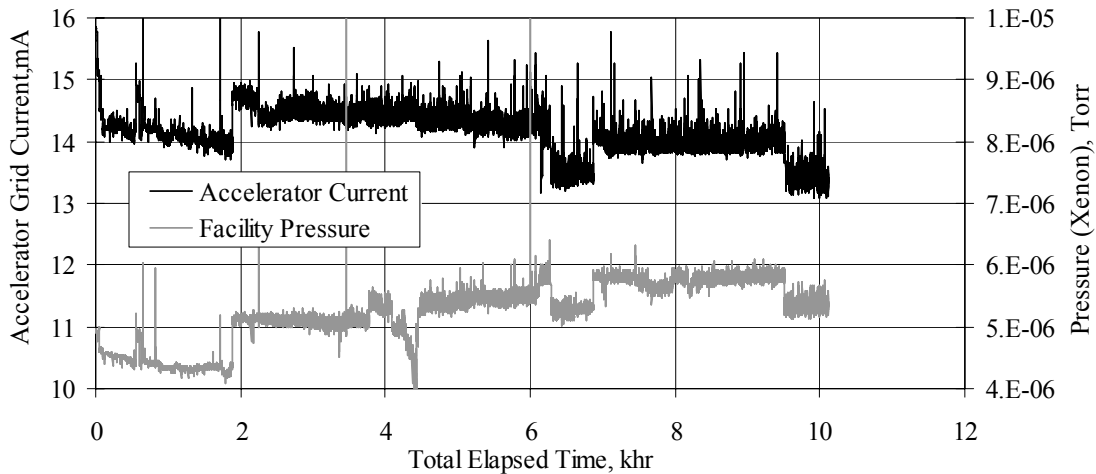


Figure 46.—Full-power accelerator grid current and background facility pressure (wall ion gauge) data as a function of time.

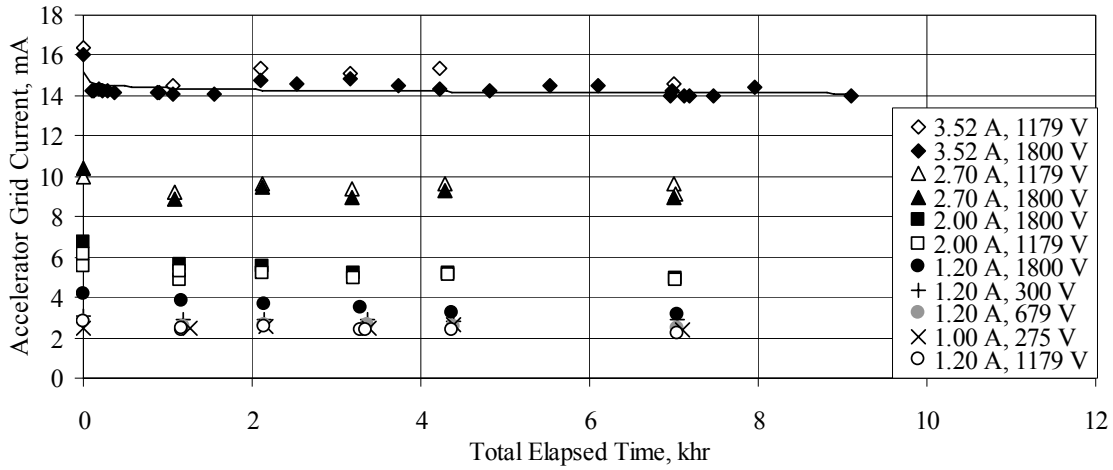


Figure 47.—Performance characterization accelerator grid current data as a function of time.

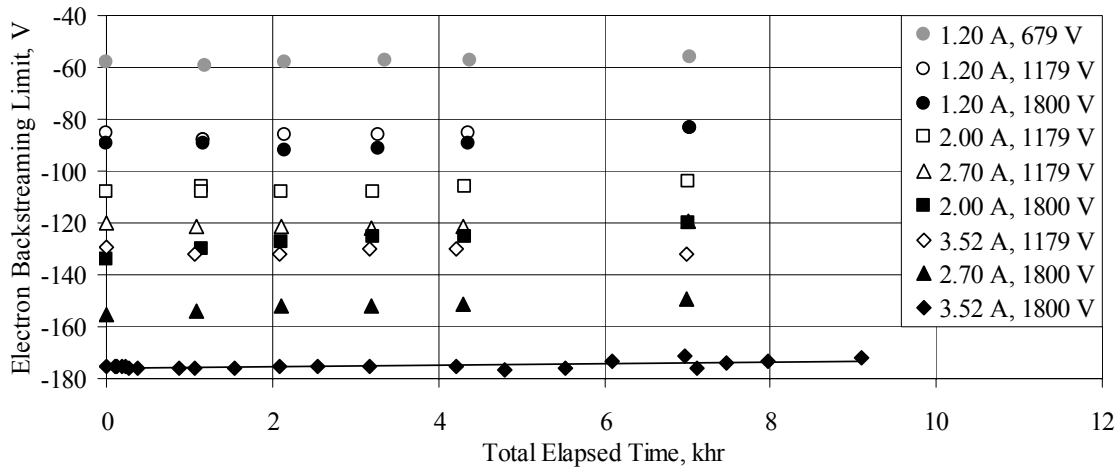


Figure 48.—Performance characterization electron backstreaming limit data as a function of time.

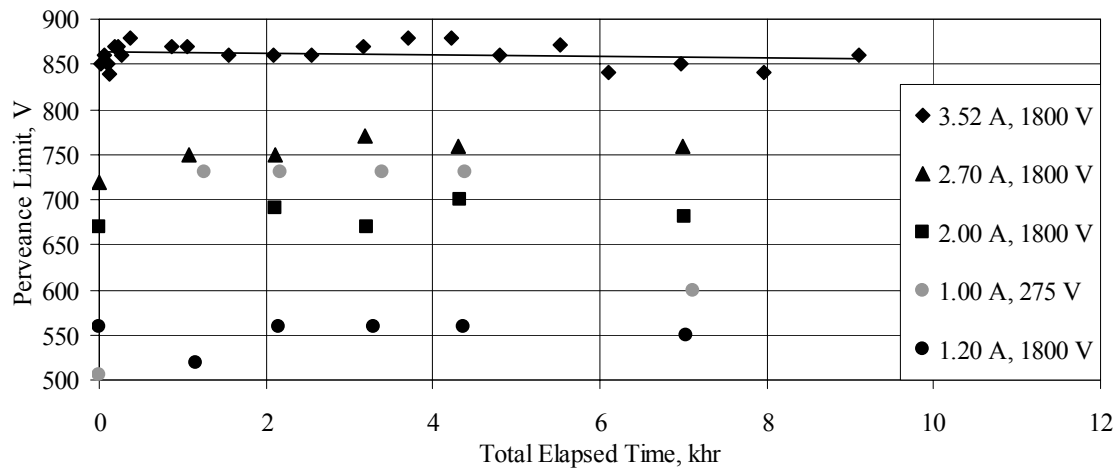


Figure 49.—Performance characterization impingement-limited total voltage data as a function of time.

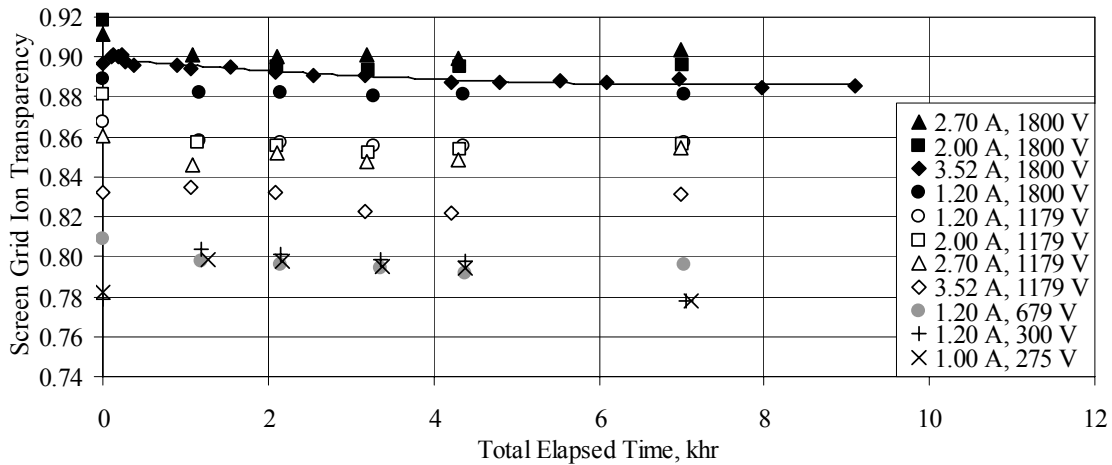


Figure 50.—Performance characterization screen grid ion transparency data as a function of time.

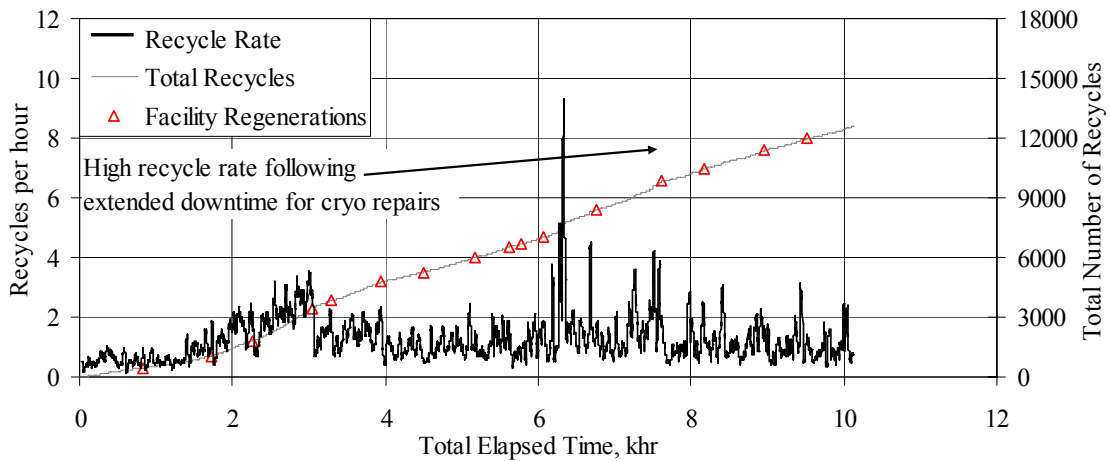


Figure 51.—Full-power thruster recycle rate and total number of recycles data as a function of time.

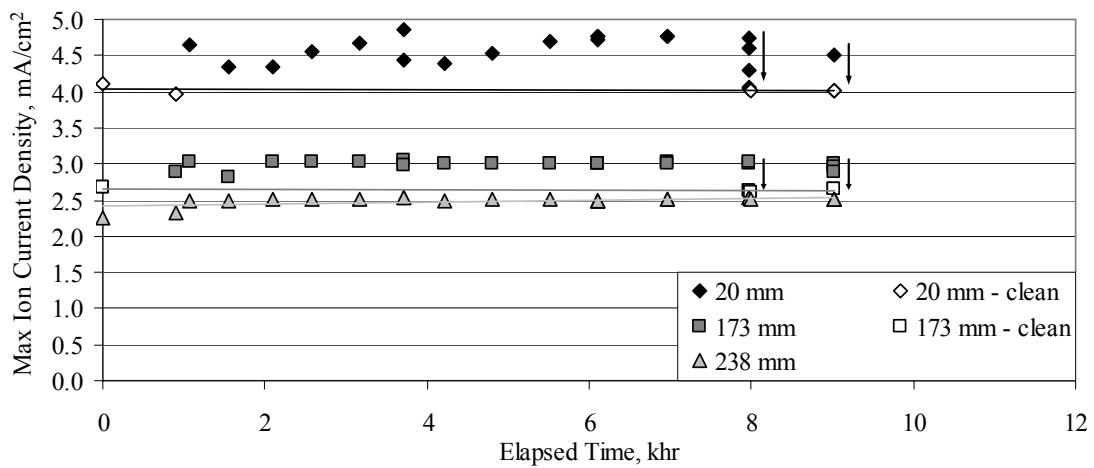


Figure 52.—Full-power peak beam current density as a function of time for each of the three staggered Faraday probes at axial positions of 20, 173, and 238 mm.

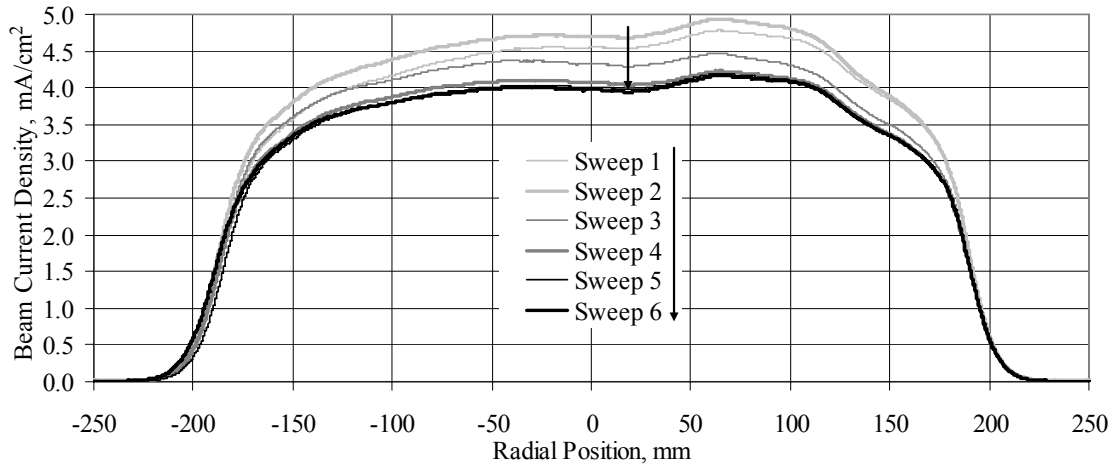


Figure 53.—Full-power radial beam current density profiles for the 20 mm probe for multiple sweeps.

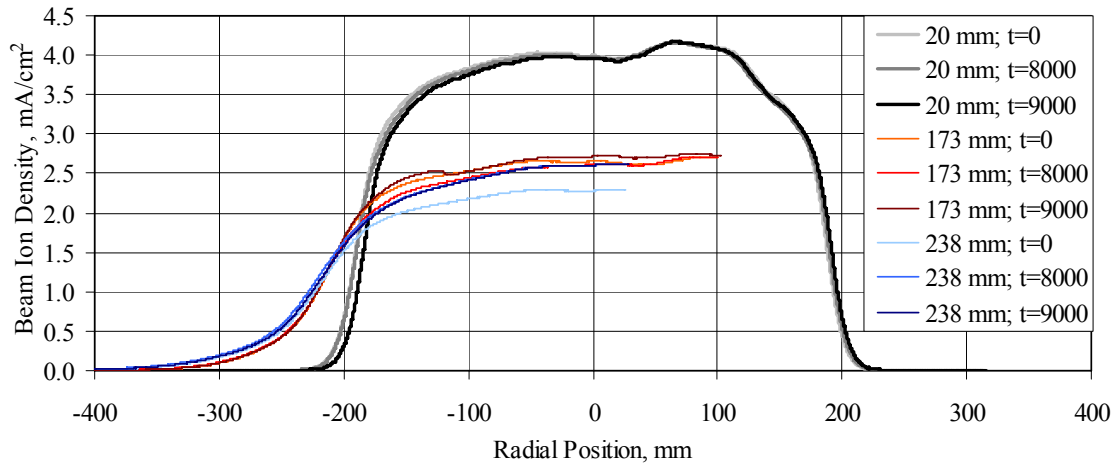


Figure 54.—Full-power thruster pretest and most recent radial beam profiles at axial positions of 20, 173, and 238 mm.

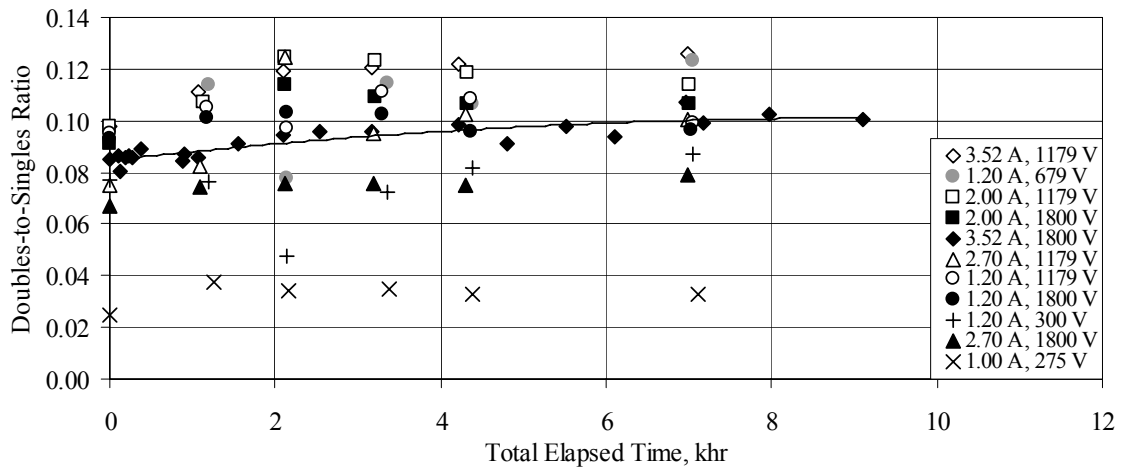


Figure 55.—Performance characterization doubles-to-singles signatures as a function of time.

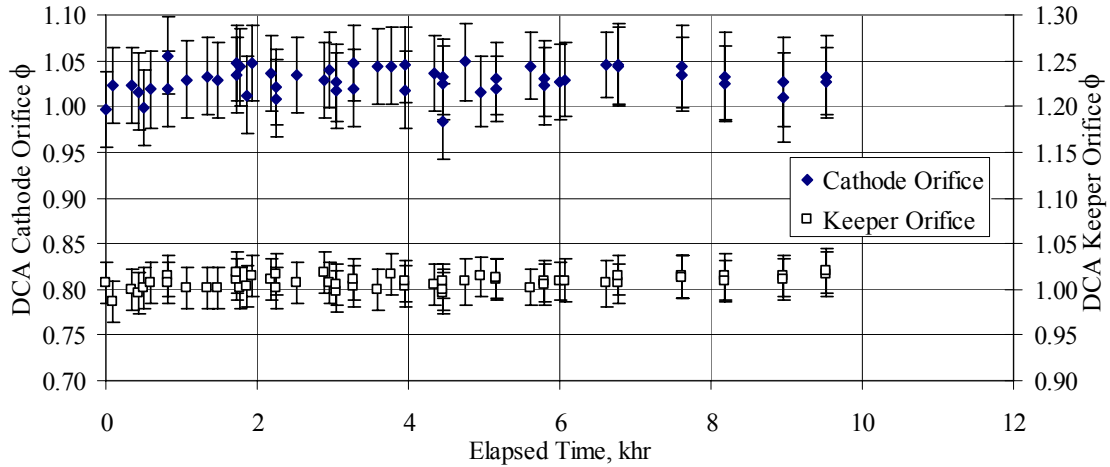


Figure 56.—Discharge cathode orifice diameter and keeper orifice diameter normalized to pretest values as a function of time.

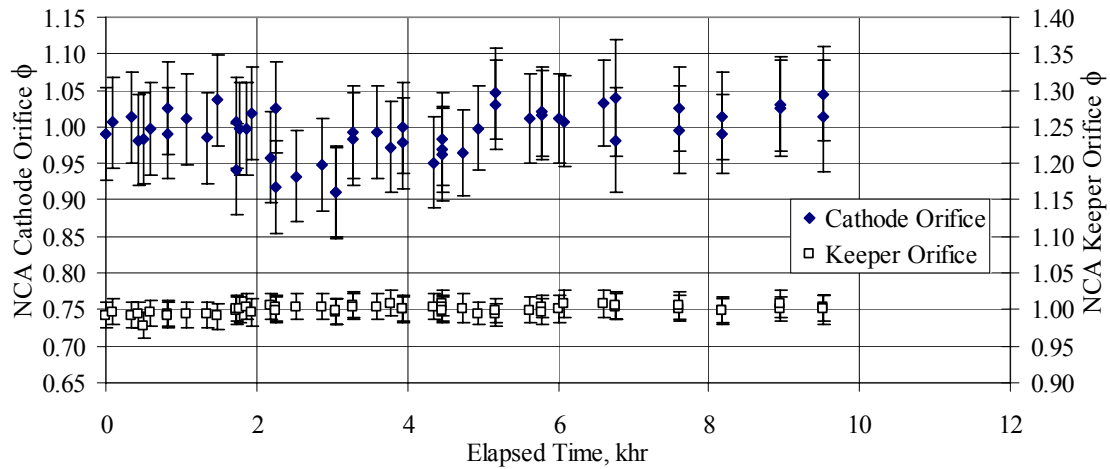


Figure 57.—Neutralizer cathode orifice diameter and keeper orifice diameter normalized to pretest values as a function of time.

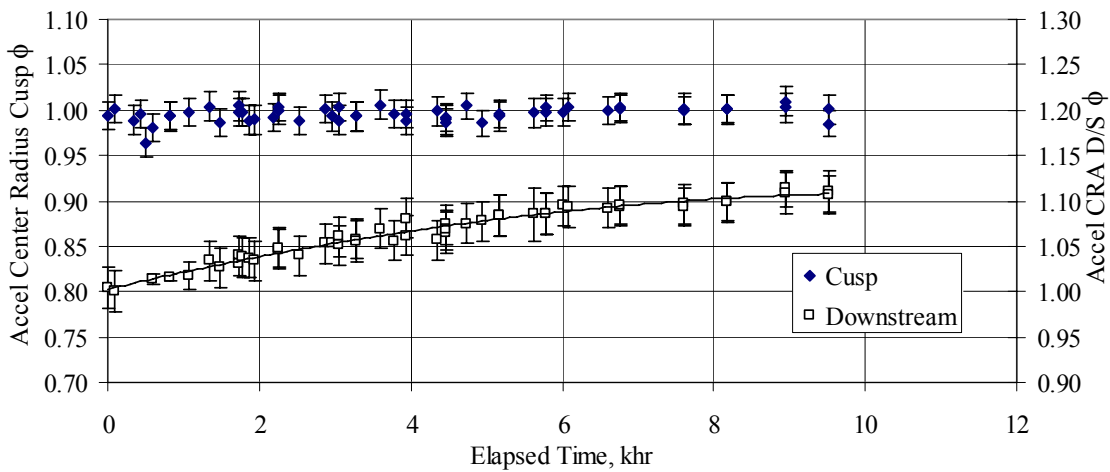


Figure 58.—Accelerator grid center-radius cusp diameter and downstream diameter normalized to pretest values as a function of time.

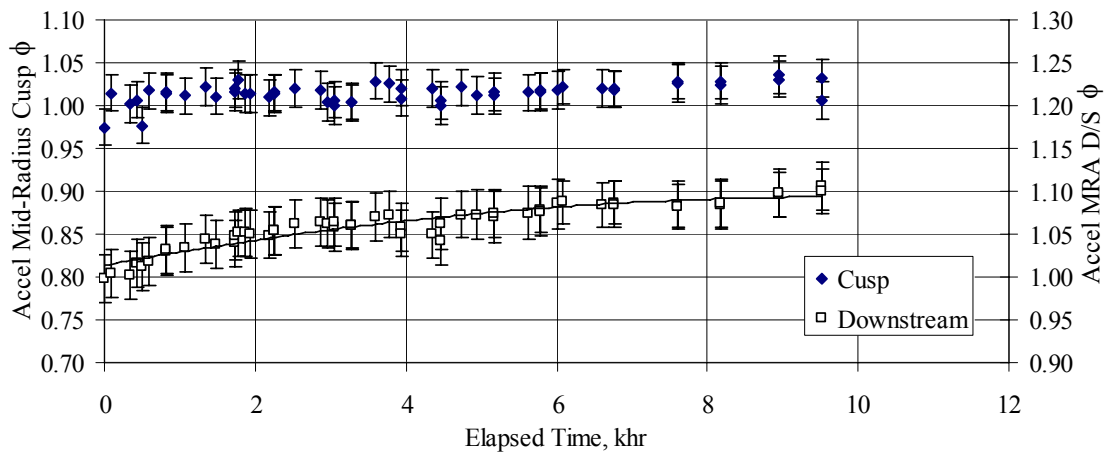


Figure 59.—Accelerator grid mid-radius cusp diameter and downstream diameter normalized to pretest values as a function of time.

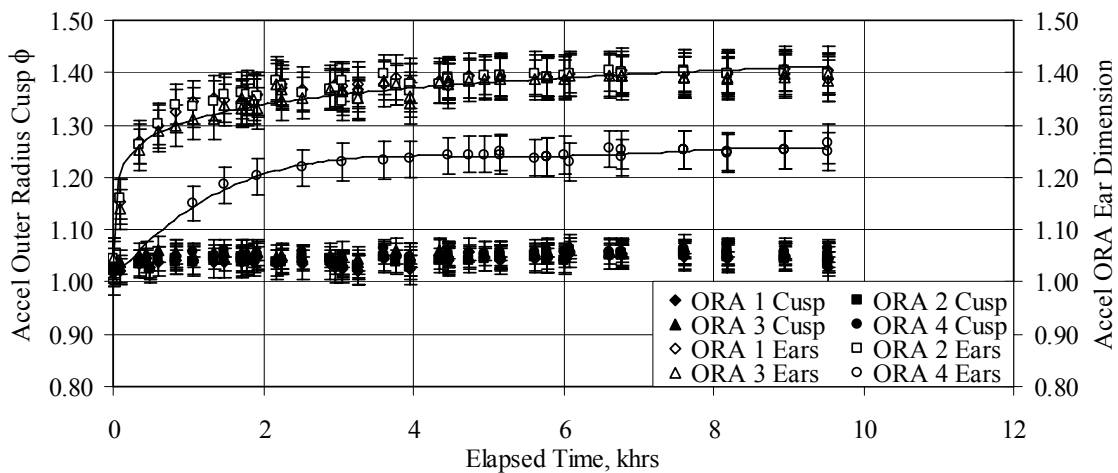


Figure 60.—Accelerator grid outer-radius cusp diameters and downstream “ear” dimensions normalized to pretest values as a function of time. ORA’s 1-3 are 3-neighbor geometry and ORA 4 is 4-neighbor geometry.

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14. ABSTRACT The NASA's Evolutionary Xenon Thruster (NEXT) program is developing the next-generation ion propulsion system with significant enhancements beyond the state-of-the-art in ion propulsion to provide future NASA science missions with enhanced mission capabilities at a low total development cost. As part of a comprehensive thruster service life assessment utilizing both testing and analyses, a Long-Duration Test (LDT) was initiated to validate and qualify the NEXT propellant throughput capability to a qualification-level of 450 kg, 1.5 times the mission-derived throughput requirement of 300 kg. This wear test is being conducted with a modified, flight-representative NEXT engineering model ion thruster, designated EM3. As of June 21, 2007, the thruster has accumulated 10,100 hr of operation at the thruster full-input-power of 6.9 kW with 3.52 A beam current and 1800 V beam power supply voltage. The thruster has processed 207 kg of xenon and demonstrated a total impulse of 8.5'106 N-s; the highest total impulse ever demonstrated by an ion thruster in the history of space propulsion. Thruster performance tests are conducted periodically over the entire NEXT throttle table with input power ranging 0.5 to 6.9 kW. Overall ion thruster performance parameters including thrust, input power, specific impulse, and thruster efficiency have been nominal with little variation to date. Lifetime-limiting component erosion rates have been consistent with the NEXT service life assessment, which predicts the earliest failure sometime after 750 kg of xenon propellant throughput; well beyond the mission-derived lifetime requirement. The NEXT wear test data confirm that the erosion of the discharge keeper orifice, enlarging of nominal-current-density accelerator grid aperture cusps, and the decrease in cold grid-gap observed during the NSTAR Extended Life Test have been mitigated. This paper presents the status of the NEXT LDT to date.					
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